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A Shape Sensing Methodology for Beams with Generic Cross-Sections: Application to Airfoil Beams

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Abstract

This work presents a shape sensing method capable of handling some geometrical complexities commonly observed in aerospace structures. The method presented is based on the one-dimensional inverse Finite Element Method (1D-iFEM), which is capable of accurately reconstructing structural displacements of beam structures using surface strain measurements. The effects of cross-sectional variation in shear strains due to transverse or torsional loads for any general beam profile is accounted for in this 1D-iFEM formulation. The introduction of these effects allows the use of iFEM for the shape sensing of solid or thin-walled prismatic beams with any general beam profile. The performance of the new method is demonstrated through some example

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problems of prismatic beams under various static loading scenarios.

Keywords: Timoshenko Beam, Shape Sensing, Structural Health Monitoring

1. Introduction

2 For a mechanical structure, obtaining the strain field given a continuous
displacement field is a well-defined problem. Theoretically, it is governed by
4 the strain-displacement relationships, and one example of its numerical appli-
cation is observed when using the Finite Element Method (FEM). However,
6 the inverse problem of obtaining the displacement field using continuous or
discrete strain data is ill-defined and suffers from the issues of existence and
8 uniqueness of a solution. However, it is a problem of considerable significance
for various practical aerospace applications.

10 One field of application is for Structural Health Monitoring (SHM), where
the mechanical quantity measured by sensors embedded in the structure is
12 often strain or acceleration. In this context, a full-scale reconstruction of the
displacement and, subsequently, stress fields can be used to calculate damage
14 indexes based on stress concentration or displacement curvature [1, 2, 3].
Such damage indexes can be used to identify the presence of damage in the
16 structure. Another application is for morphing structures where knowledge

of the structural shape is critical, but often, the displacement field cannot
18 be measured directly. Real-time displacement reconstruction using strain
sensor data can be used as a feedback component when designing a control
20 system for the active control of the shape of the structure [4, 5]. In all these
cases, techniques that can reconstruct the displacement field, using discrete
22 or continuous strain data measured on the surface of the structure, are called
shape sensing techniques [6, 7, 8, 9, 10].

24 The inverse Finite Element Method (iFEM) is one such shape sensing
technique, developed separately for 2D plate and shell structures[11, 12] and
26 1D beam structures [13]. For the shape sensing of any general 3D beam or
frame structure, the 1D iFEM methodology is used. It is based on minimizing
28 a least-square functional defined as the difference between the theoretical and
experimental sectional strain measures. The sectional strain measures are
30 defined based on the Timoshenko beam theory and correspond to the axial,
bending, transverse shear, and torsional strains experienced by any beam
32 section. The results of the iFEM reconstruction are computationally efficient
and robust in the presence of measurement noise. It can handle structures
34 subjected to both static and dynamic loading [13] and in the geometrically
linear and non-linear deformation regimes as well [14]. As the methodology is

36 independent of the structure's material properties and loading conditions, it
is suitable for various SHM problems. Application of iFEM for SHM has been
38 demonstrated for simple 1D structures using Fibre-Optic sensors [15] and
simple plate structures with embedded cracks, using strain sensor grids [16,
40 3]. As 1D iFEM accuracy is dependent on the location and orientation of the
strain sensors on the structure, various optimal sensor placement techniques
42 have also been developed. Multi-parameter optimization problems solved
using the Particle Swarm Optimization (PSO) algorithm [17, 18] can be used
44 to identify optimal sensor positions that minimize the effect of experimental
noise and maximize the accuracy of the iFEM reconstruction.

46 The 1D-iFEM was initially proposed for solving problems involving pris-
matic beams with simple circular or rectangular profiles [13, 19]. While
48 efforts have focused on expanding the method for more complex structures,
such as using an iso-geometric analysis for handling non-prismatic beams[20],
50 the 1D-iFEM formulation still produces inaccurate results when handling
beams with complex cross-sections, such as an airfoil. An initial effort to
52 generalize the 1D formulation for any general beam cross-section was made
by capturing the dependence of transverse shear strain on the shape of the
54 beam cross-section [21]. However, the applicability of the method is con-

strained to those structures satisfying the assumptions of the formulation.

56 Expanding the methodology to more practical aerospace structures, such as
an aircraft wing-box, requires further development of the formulation. The
58 method should be able to handle effects such as torsional deformation, cross-
sectional warping, cases of non-prismatic and tapered beams, and the effects
60 of additional stiffening members such as ribs and spars. This paper presents
a gradual step in this direction by introducing some of these features into
62 the existing 1D iFEM formulation.

The paper begins with the theoretical description of the 1D-iFEM for
64 beams. The improvements proposed to account for shear and torsional load-
ing are described in section 2. Section 3 gives a brief review of some numerical
66 techniques which can be used for computing the transverse shear strain vari-
ation due to shear and torsional loading for any general beam profile. Section
68 4 describes an optimal sensor placement technique and presents the results
of some example problems of prismatic beams under various static loading
70 scenarios. Finally, section 5 concludes with the paper's main achievements
and presents potential areas for future work.

72 2. Theoretical Formulation

The theoretical formulation of the 1D and 2D iFEM for beam, plate, and
74 shell structures has been discussed extensively in previous papers by Gherlone
et al.[13, 19] and Tessler et al. [11]. The readers are encouraged to go through
76 these to get a detailed explanation of the technique. This section briefly
explains the iFEM formulation for 1D structures and the efforts to generalize
78 the formulation for prismatic beams with any general cross-sectional profile.

2.1. inverse Finite Element Method for Beams

80 The 1D-iFEM is based on the Timoshenko beam theory, where the dis-
placement field of the beam can be defined using the six kinematic variables,
82 $\{u, v, w, \theta_x, \theta_y, \theta_z\}$ (see Figure 1) as,

$$\begin{aligned}u_x(x, y, z) &= u(x) + z\theta_y(x) - y\theta_z(x) \\u_y(x, y, z) &= v(x) - z\theta_x(x) \\u_z(x, y, z) &= w(x) + y\theta_x(x)\end{aligned}\tag{1}$$

Based on the small-strain hypothesis, the axial and transverse shear strain
84 components on any section of the beam can be described as,

$$\begin{aligned}
\epsilon_x(x, y, z) &= e_1(x) + ze_2(x) + ye_3(x) \\
\gamma_{xz}(x, y, z) &= e_4(x) + ye_6(x) \\
\gamma_{xy}(x, y, z) &= e_5(x) - ze_6(x)
\end{aligned} \tag{2}$$

where $e_i, (i = 1, \dots, 6)$ are the sectional strain measures of the beam. These
86 strain measures represent the axial, bending, transverse shear, and torsional
strains and can be defined in terms of the six kinematic variables as,

$$\begin{aligned}
e_1(x) &= u_{,x}(x) & e_2(x) &= \theta_{y,x}(x) & e_3(x) &= -\theta_{z,x}(x) \\
e_4(x) &= w_{,x} + \theta_y(x) & e_5(x) &= v_{,x} - \theta_z(x) & e_6(x) &= \theta_{x,x}(x)
\end{aligned} \tag{3}$$

88 The displacement reconstruction is performed by discretizing the beam
into a set of inverse elements and interpolating the kinematic variables, and as
90 a consequence the sectional strain measures, using element shape functions.
For each inverse element of the beam, e , and each sectional strain measure,
92 k , the functional ϕ_k^e can be defined as the least square error between the

theoretical and the experimental value of that strain measure,

$$\phi_k^e = \frac{L^e}{n} \sum_{i=1}^n [e_k(x_i) - e_k^{exp,i}]^2 \quad (k = 1, \dots, 6) \quad (4)$$

94 where L^e is the element length, and n is the number of axial locations in
 an element where the experimental strain measures have been calculated.
 96 The experimental sectional strain measures are calculated using linear strain
 measurements made on the outer surface of the beam. The element functional
 98 ϕ^e can be written as the sum of the individual functionals, ϕ_k^e , multiplied by
 suitable weighing coefficients, w_i , ($i = 1, \dots, 6$), which are calculated using the
 100 area, second moments of inertia, and polar moment of inertia of the beam
 cross-section [13].

$$\phi^e = \sum_{k=1}^6 w_k^e \phi_k^e \quad (5)$$

102 Individual contributions from all the elements are assembled as in the
 direct FEM, and the final functional, ϕ , is minimized with respect to the
 104 unknown nodal degrees of freedom to get a set of linear algebraic equations.
 These equations can be solved to obtain the nodal displacements of the beam.

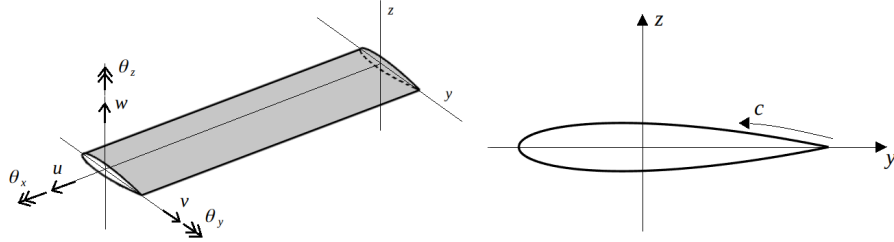


Figure 1: Sign conventions used for the six kinematic variables of the beam (left); NACA 0016 airfoil profile with the parameter c indicating the distance along the perimeter (right)

106 *2.2. Estimating Strain Measures*

To implement the iFEM procedure, the six sectional strain measures of
 108 the beam have to be calculated using experimental linear strain measure-
 ments made on the surface of the beam. This calculation can be done by
 110 formulating a set of equations relating these two quantities. The calcula-
 tion of the strain measures for prismatic beams with simple profiles has been
 112 discussed previously by Gherlone et al. [13, 19]. This formulation was ex-
 panded by Roy et al. [21] to accommodate prismatic beams with any general
 114 cross-sectional profile, with specific examples of symmetric airfoil profiles.
 However, the previous work was not successful in accurately accounting for
 116 the distribution of transverse shear strains due to torsion. The present work
 handles this limitation and provides a more detailed explanation of the entire
 118 procedure.

Assuming a prismatic beam with any arbitrary solid cross-section sub-

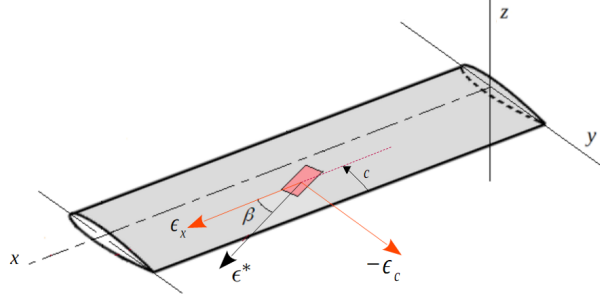


Figure 2: Strain gauge placed on the surface of the beam, oriented at an angle β with respect to the axis of the beam

120 jected to a generalized tip load consisting of axial, transverse and torsional
 loads. The magnitude of strain, ϵ^* , measured by a linear strain gauge placed
 122 on the outer surface of the beam and oriented at an angle, β , with respect to
 the beam-axis (see Figure 2) can be written as a function of the axial strain, ϵ_x ,
 124 tangential strain, ϵ_c , and tangential shear strain, γ_{xc} , on the perimeter of the
 beam section. Using a suitable strain-tensor transformation, it can be defined
 126 as[13],

$$\epsilon^*(x, c, \beta) = \epsilon_x(x, c)\cos^2\beta + \epsilon_c(x, c)\sin^2\beta + \gamma_{xc}(x, c)\cos\beta\sin\beta \quad (6)$$

where c represents the distance along the perimeter from the trailing edge
 128 for either solid or thin-walled beam sections (see Figure 1). Equation 6 can

be further simplified as [13],

$$\epsilon^*(x, c, \beta) = \epsilon_x(x, c)(\cos^2\beta - \nu\sin^2\beta) + \gamma_{xc}(x, c)\cos\beta\sin\beta \quad (7)$$

130 Here the axial strain, $\epsilon_x(x, c)$, and tangential shear strain, $\gamma_{xc}(x, c)$, should
 be represented in terms of the strain measures. Using Equation 2, the dis-
 132 tribution of ϵ_x along the perimeter of the beam can be written as,

$$\epsilon_x(x, c) = e_1(x) + e_2(x)z(c) + e_3(x)y(c) \quad (8)$$

Variation of γ_{xc} along the perimeter can be represented as a superposition
 134 of strains due to transverse and torsional loads. Under a transverse or tor-
 sional load, the shear strain at any section is not constant but is a function
 136 of the parameter c . The tangential shear strain variation due to a transverse
 load along the z -axis can be represented with the help of the shear strain
 138 variation function, $f_1(c)$, and the tangential shear strain maximum, $\gamma_{xc,max}^z$,
 which represents the amplitude of the variation. Similarly, the tangential
 140 shear strain variation due to a transverse load along the y -axis can be rep-
 resented with the help of the shear strain variation function, $f_2(c)$, and the
 142 tangential shear strain maximum, $\gamma_{xc,max}^y$, which represents the amplitude of

the variation. The overall tangential shear strain variation due to the two
 144 transverse loads will be the sum of these two contributions.

For the case of a torsional load, it is assumed that the effect of warping is
 146 negligible. Hence, the tangential shear strain variation for a beam due to a
 torsional load at the tip is represented using the torsional strain measure, e_6 ,
 148 and the function $f_3(c)$, which indicates the variation of tangential shear strain
 associated with a unit rate of twist ($e_6 = 1$). Hence the overall variation of
 150 γ_{xc} along the perimeter of the beam can be written as,

$$\gamma_{xc}(x, c) = \gamma_{xc,max}^z(x)f_1(c) + \gamma_{xc,max}^y(x)f_2(c) + e_6(x)f_3(c) \quad (9)$$

The quantities $\{\gamma_{xc,max}^z, \gamma_{xc,max}^y, e_6\}$ are proportional to the magnitudes of
 152 the loads along the z and y -axes and torsional load along the x -axis respec-
 tively. For a 3D beam, the transverse shear strain varies over the cross-section
 and is in contrast to the Timoshenko beam theory that proposes a constant
 154 transverse shear strain for any cross-section. Therefore, for implementing
 the 1D-iFEM, $\{\gamma_{xc,max}^z, \gamma_{xc,max}^y\}$, which are measured experimentally have to
 156 be related to the sectional strain measures, $\{e_4, e_5\}$, which are based on the
 Timoshenko beam theory.
 158

For the case of a cantilevered prismatic beam with a tip load, F_z , applied

160 along the z -axis, this can be done by equating the shear strain energy per
unit length at the mid-beam cross-section (to avoid end-effects) for a 3D
162 beam, with a similar case for a Timoshenko beam. Under the Timoshenko
beam theory, the shear strain energy per unit length will be the same in any
164 beam section. For the present case, the shear strain energy per unit length
for the Timoshenko beam, ϕ_{SE}^{Tim} , can be defined as,

$$\phi_{SE}^{Tim} = \frac{F_z^2}{2AG} \quad (10)$$

166 where A indicates the area of the beam cross-section, and G represents the
shear modulus of the beam material. The two shear strain energies can be
168 equated using a coefficient, k_{tz} , which is the classical shear correction factor,
and is defined as the ratio between the two quantities,

$$k_{tz} = \frac{\phi_{SE}^{Tim}}{\phi_{SE}^{FE}} = \frac{F_z^2}{2AG\phi_{SE}^{FE}} = \frac{F_z/GA}{e_4} \quad (11)$$

170 where ϕ_{SE}^{FE} is the shear strain energy calculated using a high-fidelity 3D FE
analysis or any equivalent highly accurate approach. The coefficient, k_{tz} , can
172 be calculated similarly by calculating ϕ_{SE}^{Tim} and ϕ_{SE}^{FE} for a load applied along

the y -axis.

$$k_{ty} = \frac{\phi_{SE}^{Tim}}{\phi_{SE}^{FE}} = \frac{F_y^2}{2AG\phi_{SE}^{FE}} = \frac{F_y/GA}{e_5} \quad (12)$$

174 Now, $\{e_4, e_5\}$ can be related to $\{\gamma_{xc,max}^z, \gamma_{xc,max}^y\}$ using the coefficients, $\{k_{ey}, k_{ez}\}$, which are defined as the ratio between the two,

$$k_{ez} = \frac{e_4}{\gamma_{xc,max}^z} = \frac{F_z/GA}{k_{tz}\gamma_{xc,max}^z} \quad , \quad k_{ey} = \frac{e_5}{\gamma_{xc,max}^y} = \frac{F_y/GA}{k_{ty}\gamma_{xc,max}^y} \quad (13)$$

176 The coefficients, $\{k_{ty}, k_{ey}, k_{tz}, k_{ez}\}$, are purely functions of the shape of the beam section. The tangential shear strain along the perimeter of the
178 beam can be written in terms of the shear coefficients as,

$$\gamma_{xc}(x, c) = \frac{1}{k_{ez}}e_4(x)f_1(c) + \frac{1}{k_{ey}}e_5(x)f_2(c) + e_6(x)f_3(c) \quad (14)$$

Finally, the experimentally measured surface strains can be written as a
180 function of the sectional strain measures by substituting Equations 8 and 14 into Equation 7,

$$\begin{aligned} \epsilon^*(x, c, \beta) = & \left(e_1(x) + e_2(x)z(c) + e_3(x)y(c) \right) [\cos^2\beta - \nu\sin^2\beta] + \\ & \left(\frac{1}{k_{\epsilon z}} e_4(x)f_1(c) + \frac{1}{k_{\epsilon y}} e_5(x)f_2(c) + e_6(x)f_3(c) \right) \cos\beta\sin\beta \end{aligned} \quad (15)$$

182 As Equation 15 is a linear algebraic equation with six unknowns, at least
 six experimental strain measurements are required at a beam section for
 184 solving the equation.

2.3. Calculating coefficients and functions

186 A more detailed explanation of the procedure used for calculating the
 functions, $\{f_1, f_2, f_3\}$, and coefficients, $\{k_{ty}, k_{ey}, k_{tz}, k_{\epsilon z}\}$, is provided for the
 188 case of a prismatic beam with a solid NACA 0016 airfoil profile. The beam
 considered has a length of 20 m and a chord length of 1m. Results from
 190 a high fidelity 3D FE model of the beam is used for the calculations. The
 beam model is meshed in ABAQUS using solid C3D8R elements, with 8490
 192 elements used per cross-section and 100 elements used along the beam length
 to ensure convergent results (see Table 1). An example of the meshed beam
 194 cross-section is shown in Figure 3.

For calculating the coefficients and functions for a load along the z -axis,

Table 1: Element discretization details of the FE model used in the numerical procedure: for the case of solid beam models

Solid Profile	Elements used		
	Per cross-section	Along beam length	Total
NACA 0016	8490	100	849000
NACA 6516	8710	100	871000

Table 2: Element discretization details of the FE model used in the numerical procedure: for the case of thin-walled beam models (thickness=5mm)

Thin-walled Profile	Elements used		
	Per cross-section	Along beam length	Total
NACA 0016	4828	150	642124
NACA 6516	4495	150	674250

196 the FE beam model, clamped at one end, is subject to a unit tip load along
the z -axis. The FE results are used to calculate the shear strain energy per
198 unit length for any beam cross-section. Undesired contributions due to end
effects caused by beam clamping and loading are avoided by considering the
200 beam's mid-way cross-section.

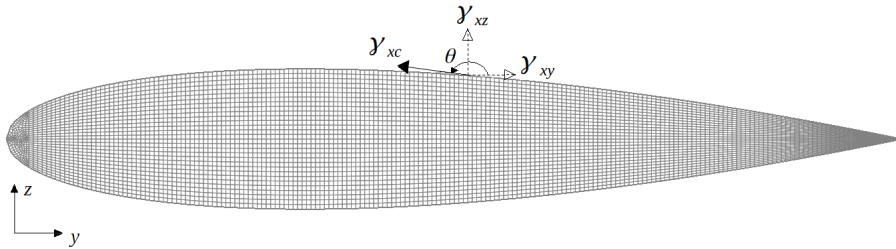


Figure 3: FE mesh of the beam cross-section for the solid beam model with a NACA 0016 airfoil profile

As the beam is meshed using solid elements, it is assumed that the strain
 202 and stress variations within each element is a constant and has a value equal
 to that at the centroid of the element. Now that the stress and strain variation
 204 is independent of the element length, the shear strain energy per unit length
 for each element is calculated as the product of the shear stress and shear
 206 strain integrated over the projected element area in the mid-beam cross-
 section. The total shear strain energy per unit length can be found as the
 208 sum of the shear strain energies of each element of the cross-section,

$$\phi_{SE}^{FE} = \frac{1}{2} \int_A (\tau_{xz}^{FE} \gamma_{xz}^{FE} + \tau_{xy}^{FE} \gamma_{xy}^{FE}) dA = \frac{1}{2} \sum_{i=1}^{N_{elem}} (\tau_{xz} \gamma_{xz} + \tau_{xy} \gamma_{xy})_i A_i^e \quad (16)$$

where, A_i^e indicates the projected area of each element of the cross-section and
 210 the transverse shear strains and stresses, $\{\tau_{xz}, \gamma_{xz}, \tau_{xy}, \gamma_{xy}\}_i$ are calculated at
 the centroid of each element of the cross-section. The variation of tangential
 212 shear strain, γ_{xc} , along the perimeter can be calculated as a combination of
 the two transverse shear strain components along the perimeter. For any
 214 node lying on the perimeter of the beam section, γ_{xc} on that node can be

calculated as (see Figure 3),

$$\gamma_{xc}(c) = \gamma_{xy}(c) \cos(\theta(c)) + \gamma_{xz}(c) \sin(\theta(c)) \quad (17)$$

216 where, angle θ represents the angle between the horizontal, y -axis, and the
tangent to the cross-section at that node. Iterating through all the nodes
218 along the perimeter, the variation of γ_{xc} and the corresponding tangential
shear strain maxima, $\gamma_{xc, max}^z$, can be calculated for a unit tip load along the
220 z -axis. Using the shear strain energy and the tangential shear strain max-
ima obtained, the shear coefficients, $\{k_{tz}, k_{\epsilon z}\}$ can be calculated using Equa-
222 tions 11 and 13. The variation function, $f_1(c)$, is obtained by calculating
the tangential shear strain variation using Equation 17 and representing the
224 variation using a suitable Fourier series or polynomial function.

A similar process can be used to obtain the shear coefficients, $\{k_{ty}, k_{\epsilon y}\}$
226 and variation function, $f_2(c)$, using a FEM beam model with a unit tip load
applied along the y -axis.

228 To obtain the variation function, $f_3(c)$, a beam model with a unit torsional
strain applied at the beam tip is used and Equation 17 is used for calculating
230 the tangential shear strain variation. For the example case of a solid prismatic
beam with a NACA 0016 airfoil profile, the shear coefficients calculated using

Table 3: Shear Coefficients for some common airfoil profiles

Beam Profile	Type	k_{ty}	k_{tz}	k_{ey}	k_{ez}
NACA 0016	Solid	0.91	0.31	0.84	1.24
NACA 0016	Thin-walled (t=5mm)	0.75	0.03	0.85	3.93
NACA 6516	Solid	1.01	0.28	0.81	1.15
NACA 6516	Thin-walled (t=5mm)	0.83	0.03	0.73	3.06

232 the above procedure is provided in Table 3 and the variation functions are
 represented using a suitable Fourier series expansions as,

$$f_1(c) = \cos\left(\frac{3}{2}\pi\frac{c}{P}\right) \quad , \quad f_2(c) = \sin\left(\pi\frac{c}{P}\right) \quad (18)$$

$$f_3(c) = 0.9 \sin(2.3c + 0.4) + 0.24 \sin(12.6c - 1.6)$$

234 where, P , indicates the half perimeter distance of the cross-section. A similar
 procedure was used to calculate the shear coefficients for some alternative
 236 solid and thin-walled, symmetric, and cambered airfoil profiles. The details
 of the FE mesh used for these beam profiles are provided in Tables 1 and 2,
 238 and the coefficients calculated using the numerical procedure are shown in
 Table 3.

240 3. Numerical methods for calculating coefficients and functions

The calculation of the functions, $\{f_1, f_2, f_3\}$, and coefficients, $\{k_{ey}, k_{ez}\}$,
242 for any arbitrary beam profile is essential for shape reconstruction based on
the iFEM methodology described above. As these functions and coefficients
244 are dependent on the beam profile, it is necessary to calculate them before
implementation. A possible procedure was demonstrated in detail in the
246 previous section, where a high fidelity 3D FE model of a beam with the
desired profile under different loading scenarios was used. The drawbacks of
248 using this procedure are that the results are obtained at a high computational
cost due to the high fidelity mesh used and require an iterative exercise to
250 simulate each beam profile under different loading scenarios. In this context,
it would be useful to investigate alternative methods that would provide
252 accurate analytical or numerical solutions at a lower computational cost.

The analytical solution of transverse shear strain for the bending or tor-
254 sion of a cantilevered prismatic beam requires a solution based on the theory
of elasticity. A solution for the torsion problem can be obtained using Saint-
256 Venant's Semi-Inverse Method, where the axial displacement is considered
a function of the warping function, $\psi(y, z)$. This problem can be solved by
258 writing the warping function in terms of a stress function, $\Phi(y, z)$, satisfying

the Poisson's equation,

$$\Phi_{,yy} + \Phi_{,zz} = -2G\theta_{,x,x} \quad (19)$$

260 where G is the shear modulus of the beam material. For solving the bend-
ing problem, the Semi-Inverse Method can be used again by making certain
262 assumptions regarding the stress distribution across the beam. It can also
be solved by representing the shear stresses using a suitable stress function,
264 which satisfies the equilibrium equations, boundary conditions, and com-
patibility conditions. Closed-form solutions for Saint-Venant's bending and
266 torsion problems exist only for a few simple cross-sections like a circle or rect-
angle [22]. The difficulties of finding an analytical solution can be avoided
268 by considering a few simple numerical and semi-analytical methods. Some
of these methods are discussed below.

270 Even though analytical solutions for the torsion problem of Equation 19
exists for simple cross-sections, it is not easy to find an exact solution for a
272 general class of airfoil profiles. However, it is possible to obtain solutions for
certain specific airfoil shapes using a specific definition of the stress function,
274 which satisfies Equation 19, like a specific family of airfoils. This approach
is described by Wang [23], where the stress function, is defined using specific

276 terms of a power series as,

$$\Phi = G\theta_{x,x} \left(-\frac{y^2 + z^2}{2} + a_0 + a_2(y^2 - z^2) + a_4(y^4 - 6y^2z^2 + z^4) \right) \quad (20)$$

Here the values of the coefficients a_0 , a_2 , and a_4 are chosen such that the
278 stress function satisfies Equation 19. They also define the boundary profile
where, the stress function should be a constant ($\Phi = const = 0$). Values
280 for the coefficients are chosen such that the boundary profile represents a
family of airfoils. The limitation is that the closed-form torsion solutions
282 are only available for some classes of symmetric airfoils. This method can
only be used for the torsion problem, and hence only the function, f_3 , can
284 be calculated. Nonetheless, it does offer the ease of using a direct analytical
solution for the iFEM procedure.

286 As discussed above, representation in terms of a power series is a powerful
tool for solving such problems. Kosmatka [24] describes another approach
288 for a prismatic beam under any general loading scenario. The overall bend-
ing and torsional warping function, ψ , is defined using a double power series

290 represented in terms of the coordinates of the beam profile,

$$\psi(y, z) = \sum_{m=0}^{\infty} \sum_{n=0}^{\infty} C_{mn} y^m z^n \quad (21)$$

Here the overall warping function can be assumed to be a linear combination of warping contributions due to beam bending and torsion. The coefficients, C_{mn} , are calculated using the principle of minimum potential energy, which is subsequently simplified to get a set of variationally defined linear algebraic equations. This procedure can be implemented numerically by discretizing the beam area using a series of triangular elements and solving the equations for each element. As it is used for solving the bending and torsion problems, all three functions, $\{f_1, f_2, f_3\}$, and coefficients can be calculated using this method. The advantage of this method is that it is capable of handling any general beam cross-section effectively.

As the aerospace domain is primarily concerned with thin-walled beam sections, the assumption that the shell thickness tends to zero can be used for obtaining useful, practical results regarding the transverse shear strain variation. Based on the above assumption, the Jourawski formula [25] offers a way of calculating the transverse shear variation functions, $\{f_1, f_2\}$, for closed thin-walled beam profiles, as a function of the length along the perimeter.

It can only be applied for transverse loading scenarios where the beam un-
308 dergoes torsion-less bending. Hence, the function, f_3 , cannot be computed.

It can be applied to relatively simple sections, such as a thin-walled airfoil,
310 or more complicated profiles such a thin-walled airfoil with a single or mul-
tiple supporting spar structures. The results, however, are constrained by
312 the initial assumptions made. As wall thickness is a crucial assumption for
thin-walled beams, more accurate solutions will be achieved when the wall
314 thickness is low.

Further simplifications can be made for solving the torsional problem
316 in the case of some thin-walled sections. Given the assumption that wall
thickness tends to zero ($t \rightarrow 0$), the shear strain along the thickness can
318 be considered uniform with a value equal to that along the center-line of
the section thickness [25]. In such a scenario, it can be shown that the
320 product of tangential shear strain along the section wall, $e_6(x)f_3(c)$, and
the wall thickness, $t(c)$, is a constant. Hence, for a section with a constant
322 wall thickness, the tangential shear strain variation, $f_3(c)$, is a constant with
respect to the parameter, c .

324 The methods presented here summarize some analytical and numerical
techniques explored by the authors for applying the iFEM for any complex

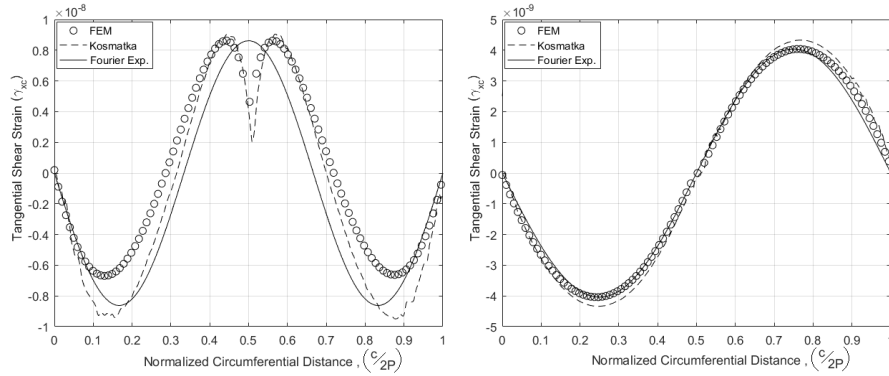


Figure 4: Comparison plot of the shear variation functions for a solid NACA 0016 airfoil profile, calculated based on different numerical methods: $f_1(c)$ due to a transverse load along the z -axis (left), $f_2(c)$ due to a transverse load along the y -axis (right)

326 beam profile. The list is by no means exhaustive, and more suitable method-
 ologies might be available in the literature. For the case of a solid prismatic
 328 beam with a NACA 0016 airfoil profile, the three functions obtained using
 FE results, the approaches of Wang [23] and Kosmatka [24] and Fourier series
 330 approximations of Equation 18 are compared and plotted in Figures 4 and 5.
 Apart from the Fourier series approximations, the other approaches provide
 332 similar results. Similar results are also observed for beams with different
 profiles considered in Table 3. The shear coefficients can also be evaluated
 334 using the different approaches with similar results.

For these reasons, the FE approach is used to calculate the shear coeffi-
 336 cients (Table 3), and these results will be used for the example problems de-
 scribed in Section 4. For calculating the functions $\{f_1, f_2, f_3\}$, a combination

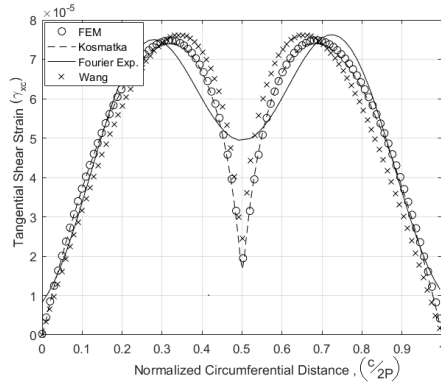


Figure 5: Comparison plot of the shear variation functions for a solid NACA 0016 airfoil profile, calculated based on different numerical methods: $f_3(c)$ due to a torsional load along the x -axis

338 of various methods are used. This is because the FE results do not provide
a direct analytical expression, but a suitable Fourier series approximation
340 has to be used to fit the data, leading to potential errors. Therefore, for all
solid beam problems, functions $\{f_1, f_2\}$, are computed using the method of
342 Kosmatka [24] and the torsion function, f_3 , using the method of Wang [23].
For thin-walled beam problems, the Jourawski formula is used for comput-
344 ing functions, $\{f_1, f_2\}$, and a constant shear strain variation is used for the
function, f_3 , based on FE results. As described above, no one method is used
346 for all problems. Depending on the beam cross-section, the method that of-
fers the greatest ease in application, without loss of accuracy, is used in the
348 iFEM formulation.

4. Application Problems

350 The performance of the formulation is tested through different example
problems of prismatic beams under various static loading scenarios. The ex-
352 perimental strains required for the iFEM reconstruction are obtained from a
high fidelity FE model of the beam modeled in ABAQUS. The solid beam
354 geometries are meshed using the C3D8R element, an 8-node linear brick
element with reduced integration. The thin-walled beams are modeled as
356 shell structures and are meshed using the S8R element, an 8-node thick shell
element with reduced integration and a quadratic interpolation of displace-
358 ments. The iFEM reconstruction accuracy is assessed by comparing it to the
displacement results of the direct FE model. All beams geometries used have
360 a length of 20 m, and the chord length of the airfoil profiles used is 1 m. The
thin-walled beams have a shell thickness of 5 mm.

362 Two different 1D iFEM elements are used to perform the displacement
reconstruction: 0^{th} and 1^{st} order inverse elements. The elements were devel-
364 oped by Gherlone et al. [13, 19] and have been discussed extensively previ-
ously, but is briefly described in this section. For each 0^{th} order element, the
366 strain measures $\{e_1, e_4, e_5, e_6\}$ are a constant throughout the element while
 $\{e_2, e_3\}$ has a linear interpolation across the element. Hence, the element

368 requires knowledge of the sectional strains at two axial locations for each ele-
 ment. At each axial location being investigated, three sensor positions along
 370 the beam perimeter are used. Furthermore, at each sensor position, two lin-
 ear strain measurements are used, one along the axis to measure the axial
 strain and the second at an angle, β (usually 45^0), with respect to the beam
 372 axis to measure the shear strain, following the requirements of Equations 15.
 374 Therefore, a total of 12 experimental strain measurements are required per
 element. For the 1st order element, strain measures $\{e_1, e_6\}$ are a constant,
 376 $\{e_4, e_5\}$ are linear and $\{e_2, e_3\}$ have parabolic interpolation across the ele-
 ment. So the element requires knowledge of the sectional strains at three
 378 axial locations and 18 experimental strain measurements per element.

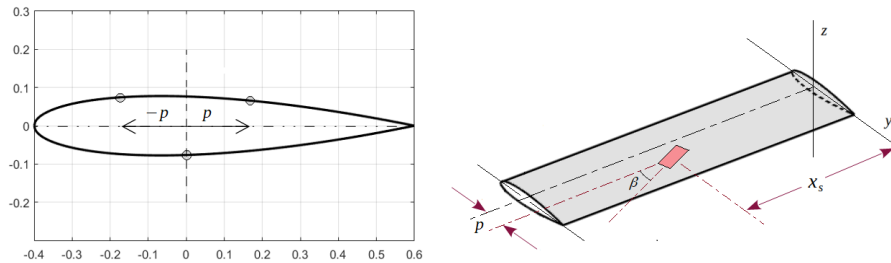


Figure 6: Parametrical representation of the sensor positions as a function of the chord length(left); The parameters, $[p, x_s, \beta]$, used to represent the sensor position and orientation for a strain gauge placed on the upper surface of the beam (right)

A simple convention is proposed to accurately describe the position and
 380 orientation of a sensor at any location on the surface of the beam regardless

of the airfoil profile: $[p^\pm, x_s, \beta]$, where the variable, p , indicates the position
382 of a sensor on the surface of the beam, with respect to the centroid of the
section, and is measured as the distance along the chord line of the airfoil
384 profile. It is parameterized with respect to the airfoil chord length to get a
normalized distance value for any profile. The superscript indicates whether
386 the sensor is positioned on the beam's top or bottom surface, respectively.
The variables x_s and β denote the length along the centroidal axis from the
388 root and the orientation of the sensor with respect to the centroidal axis,
respectively. A simple representation of the parameters is shown in Figure 6.
390 So, for a sensor located 10% of the chord length from the centroid, placed on
the upper surface and is one-third of the beam length, L , from the root and
392 oriented at an angle of 60° with respect to the centroidal axis, the notation
would be $[0.1^+, \frac{L}{3}, 60]$. This particular sensor position and arrangement can
394 be visualized by referring to Figure 7.

4.1. Optimal Strain Sensor Placement

396 Finding the optimal location for placing the strain gauges on the surface
of the beam is an aspect which influences the results of the iFEM solution.
398 When choosing the sensor location, the objective is to maximize the quantity
and quality of strain information available in any section. In this context, it

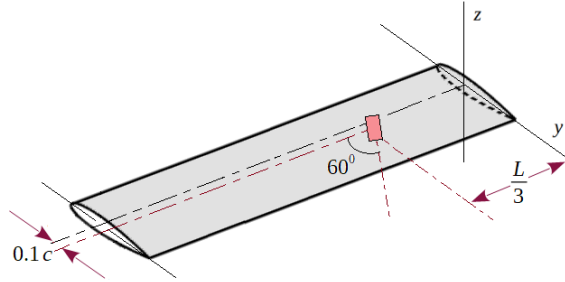


Figure 7: A strain sensor placed on the upper side of the beam, whose position and orientation conforms to the notation $[0.1c^+, \frac{L}{3}, 60]$

400 may seem obvious that two sensors placed too close-by or too far apart may
 not provide the desired strain information. An iterative study is performed
 402 to gain a quantitative measure of the optimal sensor location for a symmetric
 airfoil profile [21].

404 The parameter being iterated is the variable, p , defined previously. Three
 strain sensor positions are required at each axial location of the beam, and
 406 a sensor configuration that is symmetric with respect to the centroid of the
 beam is desired. A solid cantilevered prismatic beam with a NACA 0016
 408 airfoil profile, with unit tip forces applied along the two transverse axes, is
 used as the model for the study, and a high fidelity FE model is used to
 410 obtain the input strains and reference displacements. One 0^{th} -order beam
 element is used for the iFEM reconstruction and the sensor configuration

412 used can be described as: $[(\pm p^+, 0^-), (\frac{L}{3}, \frac{2L}{3}), (0, 45)]$. For each iterated
 sensor configuration, the beam tip displacements are reconstructed, and the
 414 percentage error in the displacements are calculated using Equation 22.

$$\%Error_U = \left(\frac{U_{tip}^{FEM} - U_{tip}^{iFEM}}{U_{tip}^{FEM}} \right) X 100, \quad U = \{u, v, w, \theta_x, \theta_y, \theta_z\} \quad (22)$$

Only the percentage error in the axial displacement, u , and the two transverse
 416 displacements, $\{v, w\}$, are used for the optimal sensor placement effort. The
 results of the rotational displacements are ignored. For a suitable comparison
 418 between the absolute error values for all three displacements, the results are
 further normalized using Equation 23 so that they lie within the range of
 420 $(0, 1)$. The absolute and normalized percentage error values are plotted in
 Figure 8.

$$Normalized \%Error_U = \left(\frac{\%Error_U - \%Error_{U,min}}{\%Error_{U,max} - \%Error_{U,min}} \right) \quad (23)$$

422 The results of the iterative study show that percentage error increases for u
 and v and decreases for w , the further a sensor is positioned from the centroid.

424 The absolute value of percentage errors shows that the magnitude of the

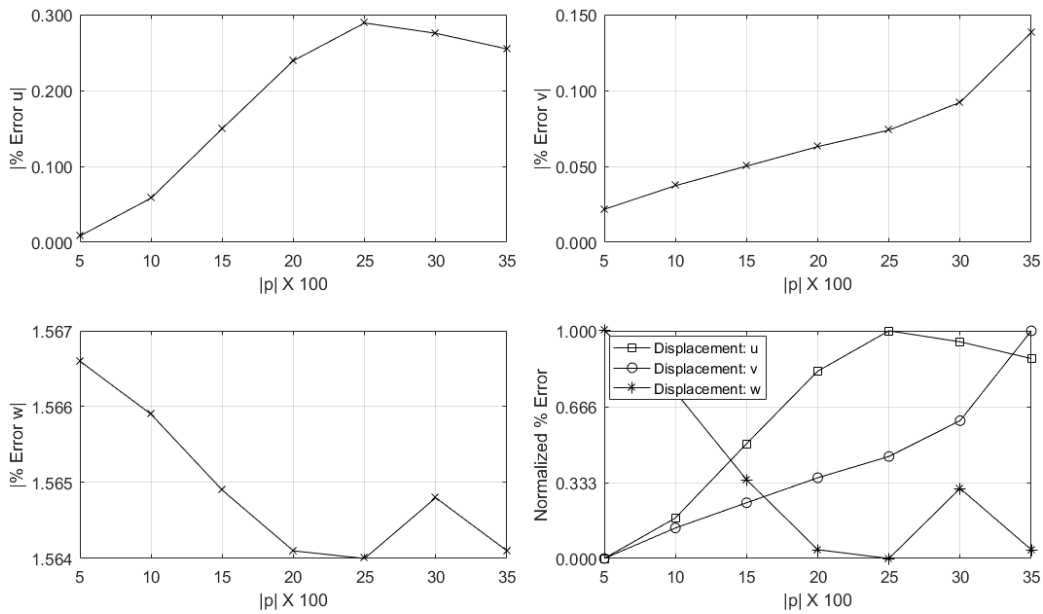


Figure 8: Absolute and normalized percentage error plots of beam tip displacements, u , v and w , plotted as a function of the sensor position p

errors are low, $< 1\%$ for u , v , and $< 2\%$ for w , highlighting the high accuracy
of the results. The absolute values also illustrate the higher sensitivity of
426 displacements u and v with varying sensor positions compared to w . For
428 selecting a suitable sensor position, the plot of the normalized percentage
error is used. A suitable sensor position would be one that presented a
430 minimum in all the three tip displacement errors. Based on the results of
Figure 8, a position approximately 10-20 % of the chord length from the
432 centroid seems suitable. The theoretical formulation of the iFEM does not
influence the choice of axial position for the sensors. It is only influenced by

434 practical considerations such as any unwanted strain fields due to the effect
of the free end or constraints present at the root. Hence, axial locations
436 sufficiently far from the root or the tip and from each other would be ideal.

4.2. Tip Loading Cases

438 The performance of the new iFEM formulation is tested using the example
problem of a solid prismatic beam with a cambered airfoil profile (NACA
440 6516). The beam is subjected to a generalized tip load consisting of axial,
transverse concentrated loads, out-of-plane moments, and torsion applied at
442 the centroid of the beam (see Figure 9). The magnitude of the forces is $1N$
and the moments and torsion are $1Nm$.

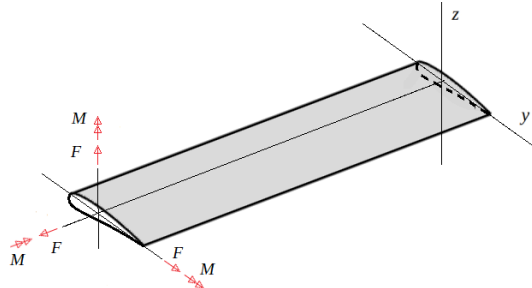


Figure 9: Generalized tip load on a solid prismatic beam with a NACA 6516 airfoil profile
 $[F = 1N, M = 1Nm]$

444 One 0^{th} -order beam element is used for the displacement reconstruction,
and the sensor configuration used is described in Table 4. The accuracy of

Table 4: Sensor position used for the 0^{th} -order beam element and the percentage error in tip displacements for the solid prismatic beam under a generalized tip load

Sensor Positions	u	v	w	θ_x	θ_y	θ_z
$[(\pm 0.1^+, 0^-), (\frac{L}{3}, \frac{2L}{3}), (0, 45)]$	0.93	-5.51	-0.53	-4.66	1.09	-5.59

446 the iFEM is assessed by comparing the reconstructed tip displacements and
rotations to the reference results coming from the direct FE model using
448 Equation 22.

The percentage error in tip displacements and rotations are shown in
450 Table 4. The results present an interesting case where the magnitude of
displacement in the two transverse directions will be different under similar
452 loads due to the unsymmetrical nature of the beam profile. Due to greater
moment of inertia about the z -axis than about the y -axis, ($I_{zz} > I_{yy}$), the
454 displacement along the z -axis will be greater than along y -axis ($w > v$).
This difference in the displacement field is also reflected in the strain data,
456 causing a more accurate reconstruction of w than v . The reconstruction of
the torsional rotation is also seen to be accurate, with an error of -4.66% . In
458 previous works [21], as transverse shear strain due to torsion was erroneously
considered a function of the distance from the shear center, the error in θ_x
460 was found to be around 34% . So the present results provide validation to the
improvements made in the formulation.

462 The current reconstruction results were obtained by using only one in-
verse element for the entire beam. Increasing the number of inverse elements
464 used (and correspondingly, the number of experimental strain measurements)
along the beam axis would improve the reconstruction accuracy of the iFEM.
466 For the current problem, increasing the number of elements used is not es-
sential as the results obtained using one element are accurate. As the main
468 focus of this paper is on extending the 1D-iFEM to beams with complex
cross-sections, increasing the number of elements is not investigated further
470 in this paper. It is left for future works where more complicated beam struc-
tures will be investigated.

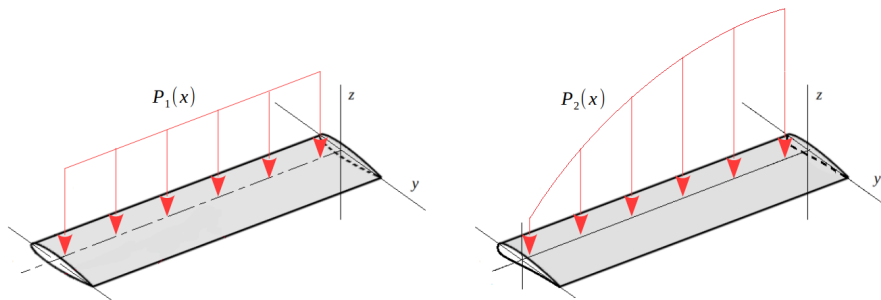


Figure 10: Uniform distributed load on a thin-walled prismatic beam with a NACA 0016 airfoil profile (left); non-uniform distributed load on a thin-walled prismatic beam with a NACA 6516 airfoil profile (right)

472 *4.3. Distributed Loading Cases*

The effect of a distributed load on the reconstruction results is tested
474 with two prismatic, thin-walled (5mm thick), beams: one with a symmetric
profile (NACA 0016) and the other with a cambered profile (NACA 6516).
476 The former is subjected to a uniform distributed load, $P_1(x) = 1N/m$. The
latter is subjected to a parabolic distributed load, made to resemble the
478 aerodynamic loading experienced by an aircraft wing, with a greater load
at the beam root, and it reduces and eventually vanishes at the beam tip
480 (Figure 10). The distributed load is defined as a function of the centroidal
axis and the beam length, L , as,

$$P_2(x) = \frac{1}{L^2}(L^2 - x^2) \quad , 0 < x < L \quad (24)$$

482 For each load case, the displacement reconstruction results obtained using
one 0^{th} -order beam element is compared to the results obtained using one
484 1^{st} -order beam element. According to the requirements of both elements,
two different sensor distributions are also used (Table 5).

486 For the two distributed load cases, the reconstruction results of the dis-
placement component, w , from both 0^{th} and 1^{st} order elements are normalized
488 with respect to the tip displacement of the FE beam model and is plotted

Table 5: Sensor configurations used for the 0^{th} and 1^{st} -order beam elements in the case of a prismatic beam with distributed loading

Element Type	Number of Sensors	Sensor Location
0^{th} order	12	$[(\pm 0.2^+, 0^-), (\frac{L}{3}, \frac{2L}{3}), (0, 45)]$
1^{st} order	18	$[(\pm 0.2^+, 0^-), (\frac{L}{4}, \frac{2L}{4}, \frac{3L}{4}), (0, 45)]$

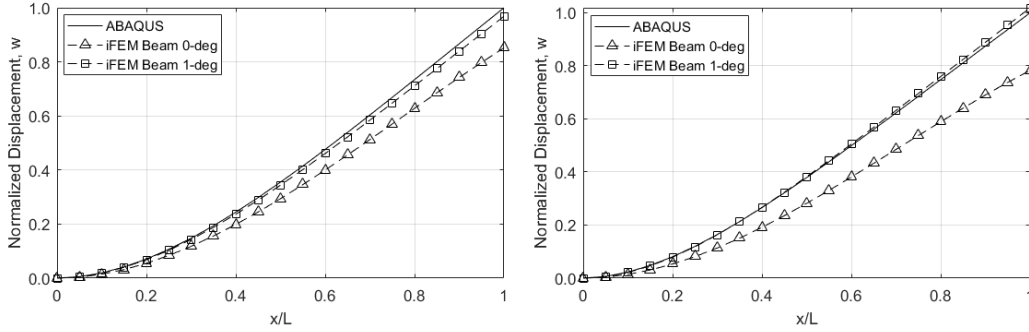


Figure 11: Plot of reconstructed normalized displacements using 0^{th} order and 1^{st} order inverse beam elements: uniform distributed load (left); parabolic distributed load (right)

in Figure 11. The percentage error in tip displacement, calculated using
 490 Equation 22, is shown in Table 6. It can be seen that the displacement
 reconstruction of the 1^{st} order element is less than 4% and improves sig-
 492 nificantly on that of the 0^{th} order element. As explained in the previous
 section, increasing the number of 0^{th} and 1^{st} order elements used for iFEM
 494 will improve the accuracy of the reconstructed displacements. However, the
 accuracy of the results when using the 1^{st} order element indicates that one
 496 element is sufficient, and further refinement is unnecessary. Structures with
 geometrical complexities that warrant further refinement will be addressed

Table 6: Percentage error in tip displacement, w , for the prismatic beams under distributed loading

Beam Profile	Load Condition	0 th -order	1 st -order
NACA 0016	Uniform Loading	-14.83	-3.15
NACA 6516	Parabolic Loading	-21.27	1.85

498 in future works.

In the present case, as the primary load was along the z -axis, the other
500 two directions (x and y -axis) remain largely unloaded. Hence, the magnitude
of the displacements $\{u, v\}$ are significantly smaller (by more than a factor
502 of 10^{-3}). Hence, only the displacements in the loading direction have been
analyzed.

504 5. Conclusion

This paper presented an improved framework for the 1D iFEM for han-
506 dling geometrical complexities commonly encountered in the shape sensing
of 3D aerospace structures. The effects of shear and torsion are important
508 factors in describing the mechanical behavior of structures. This work pre-
sented the efforts in reconciling the 2D effects of shear and torsional strains
510 with the assumptions of simple 1D beam theories. This was achieved with
the help of certain shear coefficients and variation functions used to link the
512 linear strain measured on the beam surface with an equivalent theoretical

transverse shear and torsional strain at any cross-section.

514 The performance of this new formulation was demonstrated through some
example problems of prismatic beams under various static loading scenarios.
516 The results of the iFEM show much greater reconstruction accuracy for tor-
sional displacements compared to results from previous works, underlying the
518 importance of the changes proposed in this work. Transverse displacement
reconstruction of prismatic beams was also seen to be accurate under both
520 uniform and non-uniform distributed loading, with greater accuracy seen to
be obtained when using a higher-order inverse beam element.

522 Although the changes introduced in this paper offer a way of applying
the 1D-iFEM for the shape sensing of a greater number of real-life aerospace
524 structures, it can by no means account for all the geometrical complexities
observed in real structures. Further refinement of the methodology is an
526 essential part of existing research in this area. Future research aims to bridge
this gap by focusing on thin-walled beam structures with additional stiffening
528 elements such as ribs and spars and how they affect overall shape sensing
behavior. As the current work focused primarily on prismatic beams, future
530 work will emphasize beams with a tapered or variable profile along the beam
length. Reconstruction performance under high and low-frequency structural

532 excitations will also be investigated.

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