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## Research Article

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# Heat conduction and Thermal Stress Analysis of laminated composites by a variable kinematic MITC9 shell element

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**Abstract:** The present paper considers the linear static thermal stress analysis of composite structures by means of a shell finite element with variable through-the-thickness kinematic. The temperature profile along the thickness direction is calculated by solving the Fourier heat conduction equation. The refined models considered are both Equivalent Single Layer (ESL) and Layer Wise (LW) and are grouped in the Unified Formulation by Carrera (CUF). These permit the distribution of displacements, stresses along the thickness of the multilayered shell to be accurately described. The shell element has nine nodes, and the Mixed Interpolation of Tensorial Components (MITC) method is used to contrast the membrane and shear locking phenomenon. The governing equations are derived from the Principle of Virtual Displacement (PVD). Cross-ply plate, cylindrical and spherical shells with simply-supported edges and subjected to bi-sinusoidal thermal load are analyzed. Various thickness ratios and curvature ratios are considered. The results, obtained with different theories contained in the CUF, are compared with both the elasticity solutions given in the literature and the analytical solutions obtained using the CUF and the Navier's method. Finally, plates and shells with different lamination and boundary conditions are analyzed using high-order theories in order to provide FEM benchmark solutions.

**Keywords:** Fourier heat conduction equation, Calculated Thermal Profile, Thermal stress analysis, Finite Element Method, Mixed Interpolated Tensorial Components, Carrera's Unified Formulation, Shell

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## 1 Introduction

In the typical aeronautical structures composite materials have found an increasing amount of applications. Advanced composite materials combine a number of properties, including high specific strength and stiffness, and nearly zero coefficient of thermal expansion in the fiber orientation. These relevant properties result in a growing use of composite materials in structures subjected to severe thermal environment, such as high temperatures, high gradients and cycling changes of temperature. Consequently the thermal deformations and stresses which are induced by non-uniform temperature in composite structures become important parameters in structural design. Use of higher-order theories will make it possible to determine these parameters precisely in composite structures. In each developed computational model, the stress analysis should be preceded by an accurate thermal analysis, which provides the temperature input data required for the thermal external load. A satisfactory thermal stress analysis is only possible if advanced and refined computational models are developed to correctly approximate the stiffness matrix, and if a correct thermal load is recognized. Sometimes the evaluation of a correct thermal load could be mandatory with respect to any further evaluation for the computational models.

Studies involving the thermo-elastic behaviour using classical or first-order theories are described by Kant and Khare [1] and Khdeir and Reddy [2]. In recent years, several higher-order two-dimensional models have been developed for such problems, which consider only an assumed temperature profile through the thickness. Among these, of particular interest is the higher-order model by Whu and Chen [3]. The same temperature profile is used by Khare et alii [4] to obtain a closed-form solution for the thermomechanical analysis of laminated and sandwich shells. Khdeir [5] and Khdeir et alii [6] assume a linear or constant temperature profile through the thickness. Barut et alii [7] analyze the non-linear thermoelastic behavior of shells by means of the Finite Element Method, but the assigned temperature profile is linear. In the framework

of the arbitrary distribution of temperature through the thickness, Miller et alii [8] and Dumir et alii [9] are noteworthy, in the first a classical shell theory for composite shells is given, the second remarks the importance of the zig-zag form of displacements in the thermal analysis of composite shells. In the case of shells, further investigations were made by Hsu et alii [10] for both closed form and Finite Element method, and by Ding [11] for a weak formulation for the case of state equations including the boundary conditions.

In the last few years many contributions have been proposed, which are based on Carrera Unified Formulation, to investigate the thermal effects in composite structures. In [12] a study on the influence of the through-the-thickness temperature profile on the thermo-mechanical response of multilayered anisotropic thick and thin plates has been addressed. The partially coupled stress problem was considered by solving the Fourier's conductivity equation. The importance of mixed theories for a correct prediction of transverse shear/normal stresses due to thermal loadings have been remarked in [13, 14]. A fully coupled thermo-mechanical analysis applied to plate structure is employed in [15]. Different type of loads as problems related to uniform, triangular, bi-triangular (tentlike), and localized in-plane distribution of temperature were considered in [16]. The Ritz method, based on the choice of trigonometric trial functions, was used in [17]. Extension to Functionally Graded Materials (FGMs) has been done in [18]. A thermal stability analysis of functionally graded material, isotropic and sandwich plates is studied in [19], the Ritz method is employed and uniform, linear, and non-linear temperature profile is taken into account for different cases. An extension of the thermoelastic formulation to shells has been done in [20] and the Fourier heat conduction equation was employed for shell in [21]. The thermo-mechanical analysis of functionally graded shell is considered in [22]. Analytical closed form solutions are available in very few cases. The solution of the most of the practical problems demand applications of approximated computational methods.

In this paper, the authors desire to demonstrate as the assumption of a priori linear temperature profile in the thickness direction could be wrong for particular shell and plate configurations, and as the use of the Fourier heat conduction equation could result mandatory to obtain a correct thermal load. Therefore, we would like to demonstrate as a wrong thermal load invalidates the static response of plate and shell structures even when advanced computational models are employed. An improved doubly-curved shell finite element for the analysis of composite structures under thermal loads is here presented, it is a natural

extension of the plate finite element presented in [23]. The shell finite element is based on the Carrera's Unified Formulation (CUF), which was developed by Carrera for multi-layered structures [24, 25]. Both Equivalent Single Layer (ESL) and Layer Wise (LW) theories contained in the CUF have been implemented in the shell finite element. The Mixed Interpolation of Tensorial Components (MITC) method [26–29] is used to contrast the membrane and shear locking. The governing equations for the linear static analysis of composite structures are derived from the Principle of Virtual Displacement (PVD), in order to apply the finite element method. The temperature profile is calculated solving the Fourier heat conduction equation and compared with the linear profile through the thickness for each two-dimensional model. Cross-ply plate, cylindrical and spherical shells with simply-supported edges and subjected to bi-sinusoidal thermal loads are analyzed. The results obtained with the different models contained in the CUF, are compared with the exact solution given in the literature and the analytical Navier's solution type. Finally, plates and shells with different lamination and boundary conditions are also analyzed using high-order theories in order to provide FEM benchmark solutions

## 2 Geometrical and constitutive relations for shell

Shells are bi-dimensional structures in which one dimension (in general the thickness in  $z$  direction) is negligible with respect to the other two in-plane dimensions. Geometry and the reference system are indicated in Fig. 1. By considering multilayered structures, the square of an infinitesimal linear segment in the layer, the associated infinitesimal area and volume are given by:

$$\begin{aligned} ds_k^2 &= H_\alpha^{k^2} d\alpha_k^2 + H_\beta^{k^2} d\beta_k^2 + H_z^{k^2} dz_k^2, \\ d\Omega_k &= H_\alpha^k H_\beta^k d\alpha_k d\beta_k, \\ dV &= H_\alpha^k H_\beta^k H_z^k d\alpha_k d\beta_k dz_k, \end{aligned} \quad (1)$$

where the metric coefficients are:

$$H_\alpha^k = A^k(1 + z_k/R_\alpha^k), \quad H_\beta^k = B^k(1 + z_k/R_\beta^k), \quad H_z^k = 1. \quad (2)$$

$k$  denotes the  $k$ -layer of the multilayered shell;  $R_\alpha^k$  and  $R_\beta^k$  are the principal radii of the midsurface of the layer  $k$ .  $A^k$  and  $B^k$  are the coefficients of the first fundamental form of

$\Omega_k$  ( $\Gamma_k$  is the  $\Omega_k$  boundary). In this paper, the attention has been restricted to shells with constant radii of curvature (cylindrical, spherical, toroidal geometries) for which  $A^k = B^k = 1$ . Details for shells are reported in [30].

Geometrical relations permit the in-plane  $\epsilon_p^k$  and out-plane  $\epsilon_n^k$  strains to be expressed in terms of the displacement  $\mathbf{u}$ . The following relations hold:

$$\begin{aligned} \epsilon_p^k &= [\epsilon_{\alpha\alpha}^k, \epsilon_{\beta\beta}^k, \epsilon_{\alpha\beta}^k]^T = (\mathbf{D}_p^k + \mathbf{A}_p^k) \mathbf{u}^k, \\ \epsilon_n^k &= [\epsilon_{\alpha z}^k, \epsilon_{\beta z}^k, \epsilon_{zz}^k]^T = (\mathbf{D}_{n\Omega}^k + \mathbf{D}_{nz}^k - \mathbf{A}_n^k) \mathbf{u}^k. \end{aligned} \quad (3)$$

The explicit form of the introduced arrays is:

$$\begin{aligned} \mathbf{D}_p^k &= \begin{bmatrix} \frac{\partial_\alpha}{H_\alpha^k} & 0 & 0 \\ 0 & \frac{\partial_\beta}{H_\beta^k} & 0 \\ \frac{\partial_\beta}{H_\beta^k} & \frac{\partial_\alpha}{H_\alpha^k} & 0 \end{bmatrix}, \quad \mathbf{D}_{n\Omega}^k = \begin{bmatrix} 0 & 0 & \frac{\partial_\alpha}{H_\alpha^k} \\ 0 & 0 & \frac{\partial_\beta}{H_\beta^k} \\ 0 & 0 & 0 \end{bmatrix}, \\ \mathbf{D}_{nz}^k &= \begin{bmatrix} \partial_z & 0 & 0 \\ 0 & \partial_z & 0 \\ 0 & 0 & \partial_z \end{bmatrix}, \end{aligned} \quad (4)$$

$$\mathbf{A}_p^k = \begin{bmatrix} 0 & 0 & \frac{1}{H_\alpha^k R_\alpha^k} \\ 0 & 0 & \frac{1}{H_\beta^k R_\beta^k} \\ 0 & 0 & 0 \end{bmatrix}, \quad \mathbf{A}_n^k = \begin{bmatrix} \frac{1}{H_\alpha^k R_\alpha^k} & 0 & 0 \\ 0 & \frac{1}{H_\beta^k R_\beta^k} & 0 \\ 0 & 0 & 0 \end{bmatrix}. \quad (5)$$

The definition of the 3D constitutive equations permits the stresses to be expressed by means of the strains. The generalized Hooke's law is considered, by employing a linear constitutive model for infinitesimal deformations. In a composite material, these equations are obtained in material coordinates (1, 2, 3) for each orthotropic layer  $k$  and then rotated in the general curvilinear reference system  $(\alpha, \beta, z)$ . Therefore, the stress-strain relations after the rotation are:

$$\begin{aligned} \sigma_p^k &= [\sigma_{\alpha\alpha}^k, \sigma_{\beta\beta}^k, \sigma_{\alpha\beta}^k]^T = \sigma_{pd}^k - \sigma_{pT}^k = \mathbf{C}_{pp}^k \epsilon_p^k + \mathbf{C}_{pn}^k \epsilon_n^k - \lambda_p^k \theta^k \\ \sigma_n^k &= [\sigma_{\alpha z}^k, \sigma_{\beta z}^k, \sigma_{zz}^k]^T = \sigma_{nd}^k - \sigma_{nT}^k = \mathbf{C}_{np}^k \epsilon_p^k + \mathbf{C}_{nn}^k \epsilon_n^k - \lambda_n^k \theta^k \end{aligned} \quad (6)$$

where

$$\begin{aligned} \mathbf{C}_{pp}^k &= \begin{bmatrix} C_{11}^k & C_{12}^k & C_{16}^k \\ C_{12}^k & C_{22}^k & C_{26}^k \\ C_{16}^k & C_{26}^k & C_{66}^k \end{bmatrix}, \quad \mathbf{C}_{pn}^k = \begin{bmatrix} 0 & 0 & C_{13}^k \\ 0 & 0 & C_{23}^k \\ 0 & 0 & C_{36}^k \end{bmatrix} \\ \mathbf{C}_{np}^k &= \begin{bmatrix} 0 & 0 & 0 \\ 0 & 0 & 0 \\ C_{13}^k & C_{23}^k & C_{36}^k \end{bmatrix}, \quad \mathbf{C}_{nn}^k = \begin{bmatrix} C_{55}^k & C_{45}^k & 0 \\ C_{45}^k & C_{44}^k & 0 \\ 0 & 0 & C_{33}^k \end{bmatrix} \end{aligned} \quad (7)$$

$$\begin{aligned} \lambda_p^k &= \mathbf{C}_{pp}^k \alpha_p^k + \mathbf{C}_{pn}^k \alpha_n^k \\ \lambda_n^k &= \mathbf{C}_{np}^k \alpha_p^k + \mathbf{C}_{nn}^k \alpha_n^k \end{aligned} \quad (8)$$

$$\alpha_p^k = \begin{bmatrix} \alpha_1^k \\ \alpha_2^k \\ 0 \end{bmatrix}, \quad \alpha_n^k = \begin{bmatrix} 0 \\ 0 \\ \alpha_3^k \end{bmatrix} \quad (9)$$

$$\lambda_p^k = \begin{bmatrix} \lambda_1^k \\ \lambda_2^k \\ \lambda_6^k \end{bmatrix}, \quad \lambda_n^k = \begin{bmatrix} 0 \\ 0 \\ \lambda_3^k \end{bmatrix} \quad (10)$$

The subscripts  $d$  and  $T$  mean mechanical and thermal contributions. The material coefficients  $C_{ij}$  depend on the Young's moduli  $E_1, E_2, E_3$ , the shear moduli  $G_{12}, G_{13}, G_{23}$  and Poisson moduli  $\nu_{12}, \nu_{13}, \nu_{23}, \nu_{21}, \nu_{31}, \nu_{32}$  that characterize the layer material.  $\alpha_{ij}$  are the thermal expansion coefficients,  $\lambda_{ij}$  are the coefficients of thermo-mechanical coupling and  $\theta^k$  is the difference with a reference temperature.

### 3 Carrera Unified Formulation for Shell

The variation of the displacement variables along the thickness direction is a-priori postulated. Several displacement-based theories can be formulated on the basis of the following generic kinematic field. The main feature of the Unified Formulation by Carrera [25, 31, 32] (CUF) is the unified manner in which the displacement variables are handled.

$$\begin{aligned} \mathbf{u}^k(\alpha, \beta, z) &= F_s(z) \mathbf{u}_s^k(\alpha, \beta); \\ \delta \mathbf{u}^k(\alpha, \beta, z) &= F_\tau(z) \delta \mathbf{u}_\tau^k(\alpha, \beta) \quad \tau, s = 0, 1, \dots, N \end{aligned} \quad (11)$$

where  $(\alpha, \beta, z)$  is a curvilinear reference system, in which  $\alpha$  and  $\beta$  are orthogonal and the curvature radii  $R_\alpha$  and  $R_\beta$  are constant in each point of the domain  $\Omega$  (see Fig. 1). The displacement vector  $\mathbf{u} = \{u, v, w\}$  has its components expressed in this system.  $\delta \mathbf{u}$  indicates the virtual displacement associated to the virtual work and  $k$  identifies the layer.  $F_\tau$  and  $F_s$  are the so-called thickness functions depending only on  $z$ .  $\mathbf{u}_s$  are the unknown variables depending on the coordinates  $\alpha$  and  $\beta$ .  $\tau$  and  $s$  are sum indexes and  $N$  is the order of expansion in the thickness direction assumed for the displacements.

#### Classical Theories

The simplest plate/shell theory is based on the Kirchhoff/Love's hypothesis, and it is usually referred to as Classical Lamination Theory (CLT)[33],[34]. Both transverse

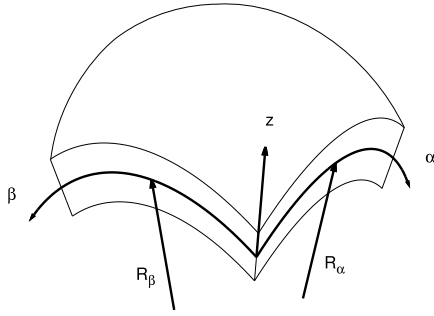


Figure 1: Reference system of the double curvature shell.

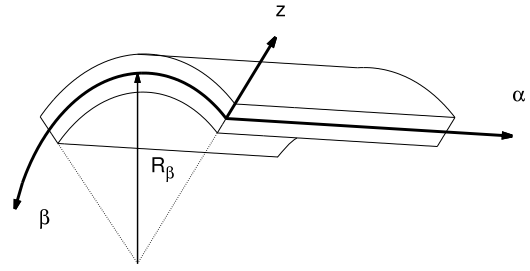


Figure 2: Reference system of the cylindrical shell.

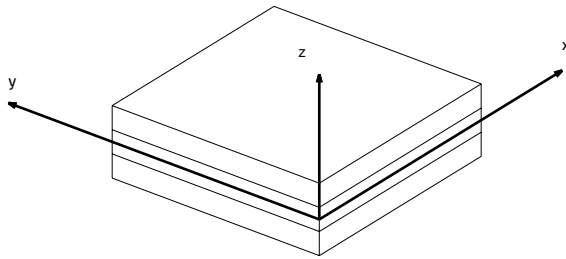


Figure 3: Reference system of the plate.

shear strains and transverse normal strains are discarded, in usual applications being negligible with respect to the in-plane ones,

$$\begin{cases} u(\alpha, \beta, z) = u_0(\alpha, \beta) - z \frac{\partial w_0}{\partial \alpha} \\ v(\alpha, \beta, z) = v_0(\alpha, \beta) - z \frac{\partial w_0}{\partial \beta} \\ w(\alpha, \beta, z) = w_0(\alpha, \beta) \end{cases} \quad (12)$$

The inclusion of transverse shear strains, in the theory mentioned here, leads to Reissner-Mindlin Theory, also known as First-order Shear Deformation Theory (FSDT) [35],

$$\begin{cases} u(\alpha, \beta, z) = u_0(\alpha, \beta) + z u_1(\alpha, \beta) \\ v(\alpha, \beta, z) = v_0(\alpha, \beta) + z v_1(\alpha, \beta) \\ w(\alpha, \beta, z) = w_0(\alpha, \beta) \end{cases} \quad (13)$$

However, these theories, due to their inconsistency in discarding the transverse normal stress in the material constitutive equations, are no longer valid when 3D local effects appear. Among these, stretching effects are neglected in CLT and FSDT models because the transverse

displacement is assumed constant in the thickness direction. Moreover, local effects due to concentrated loads cannot be represented if linear displacement field is considered. To remove the inconsistency completely, higher-order expansion of the unknown with respect to the  $z$  coordinate are needed. For more details, the readers can refer to the article [24].

### Equivalent Single Layer Theories

Many attempts have been made to improve classical plate/shell models. The CUF has the capability to expand each displacement variable in the displacement field at any desired order independently from the others and with respect to the accuracy and the computational cost has been introduced. Such an artifice permits us to treat each variable independently from the others. This becomes extremely useful when multifield problems are investigated such as thermoelastic and piezoelectric applications [13, 20, 36].

In the case of Equivalent Single Layer (ESL) models, a Taylor expansion is employed as thickness functions:

$$\mathbf{u} = F_0 \mathbf{u}_0 + F_1 \mathbf{u}_1 + \dots + F_N \mathbf{u}_N = F_s \mathbf{u}_s, \quad s = 0, 1, \dots, N. \quad (14)$$

$$F_0 = z^0 = 1, \quad F_1 = z^1 = z, \quad \dots, \quad F_N = z^N. \quad (15)$$

Following this approach the displacement field can be written as:

$$\begin{cases} u(\alpha, \beta, z) = u_0(\alpha, \beta) + z u_1(\alpha, \beta) + \dots + z^N u_N(\alpha, \beta) \\ v(\alpha, \beta, z) = v_0(\alpha, \beta) + z v_1(\alpha, \beta) + \dots + z^N v_N(\alpha, \beta) \\ w(\alpha, \beta, z) = w_0(\alpha, \beta) + z w_1(\alpha, \beta) + \dots + z^N w_N(\alpha, \beta) \end{cases} \quad (16)$$

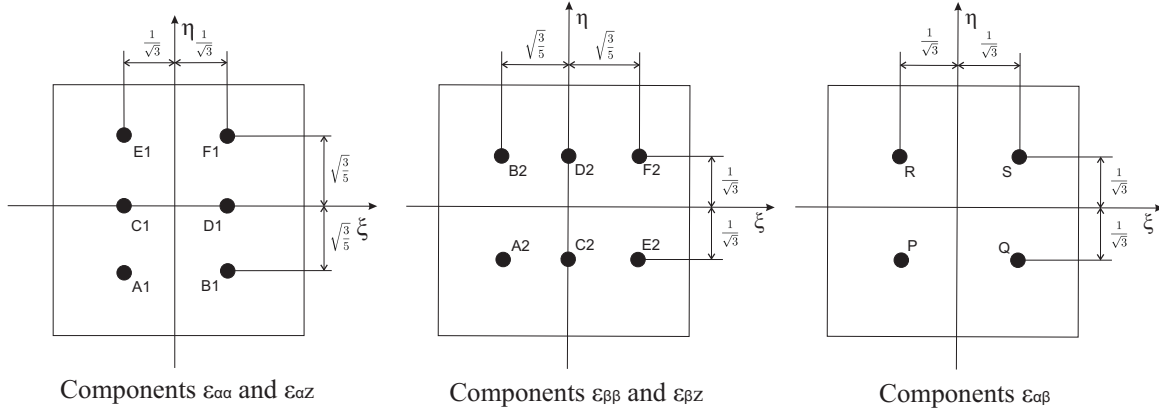


Figure 4: Tying points for the MITC9 shell finite element.

In general:

$$\begin{cases} u(\alpha, \beta, z) = F_0(\alpha, \beta) + F_1 u_1(\alpha, \beta) + \dots + F_N u_N(\alpha, \beta) \\ v(\alpha, \beta, z) = F_0(\alpha, \beta) + F_1 v_1(\alpha, \beta) + \dots + F_N v_N(\alpha, \beta) \\ w(\alpha, \beta, z) = F_0(\alpha, \beta) + F_1 w_1(\alpha, \beta) + \dots + F_N w_N(\alpha, \beta) \end{cases} \quad (17)$$

Classical models, such as those based on the First-order Shear Deformation Theory (FSDT), can be obtained from an ESL theory with  $N = 1$ , by imposing a constant transverse displacement through the thickness via penalty techniques (that is, the degree of freedom given by the linear part of the transverse displacement is penalized by assigning infinite value to the corresponding term in the diagonal of the stiffness matrix). Also a model based on the hypotheses of Classical Lamination Theory (CLT) can be expressed by means of the CUF by applying a penalty technique to the constitutive equations (the penalty is here applied to the shear modulus in the matrix of the material coefficients). This permits to impose null transverse shear strains in the shell.

### Zig-Zag Models and Layer Wise Theories

Due to the intrinsic anisotropy of multilayered structures, the first derivative of the displacement variables in the  $z$ -direction is discontinuous. The Layer-Wise (LW) models, in respect to the ESLs, allow the zig-zag form of the displacement distribution in layered structures to be modelled. It is possible to reproduce the zig-zag effects also in the framework of the ESL description by employing the Murakami theory. According to references [37], a zig-zag term can be introduced into equation(14) as follows:

$$\mathbf{u}^k = F_0 \mathbf{u}_0^k + \dots + F_N \mathbf{u}_N^k + (-1)^k \zeta_k \mathbf{u}_Z^k. \quad (18)$$

Subscript  $Z$  refers to the introduced term. Such theories are called zig-zag ( $Z$ ) theories. Following this approach the displacement field can be written as:

$$\begin{cases} u(\alpha, \beta, z) = F_0(\alpha, \beta) + F_1 u_1(\alpha, \beta) + \dots + F_{N-1} u_{N-1}(\alpha, \beta) + (-1)^k \zeta_k u_Z^k \\ v(\alpha, \beta, z) = F_0(\alpha, \beta) + F_1 v_1(\alpha, \beta) + \dots + F_{N-1} v_{N-1}(\alpha, \beta) + (-1)^k \zeta_k v_Z^k \\ w(\alpha, \beta, z) = F_0(\alpha, \beta) + F_1 w_1(\alpha, \beta) + \dots + F_{N-1} w_{N-1}(\alpha, \beta) + (-1)^k \zeta_k w_Z^k \end{cases} \quad (19)$$

In the case of Layer-Wise (LW) models, the displacement is defined at  $k$ -layer level:

$$\begin{aligned} \mathbf{u}^k &= F_t \mathbf{u}_t^k + F_b \mathbf{u}_b^k + F_r \mathbf{u}_r^k = F_s \mathbf{u}_s^k, \\ s &= t, b, r, \quad r = 2, \dots, N. \end{aligned} \quad (20)$$

$$F_t = \frac{P_0 + P_1}{2}, \quad F_b = \frac{P_0 - P_1}{2}, \quad F_r = P_r - P_{r-2}. \quad (21)$$

in which  $P_j = P_j(\zeta_k)$  is the Legendre polynomial of  $j$ -order defined in the  $\zeta_k$ -domain:  $-1 \leq \zeta_k \leq 1$ .  $P_0 = 1, P_1 = \zeta_k, P_2 = (3\zeta_k^2 - 1)/2, P_3 = (5\zeta_k^3 - 3\zeta_k)/2, P_4 = (35\zeta_k^4 - 30\zeta_k^2 + 3)/8$ . The top ( $t$ ) and bottom ( $b$ ) values of the displacements are used as unknown variables and one can impose the following compatibility conditions:

$$u_t^k = u_b^{k+1}, \quad k = 1, N_l - 1. \quad (22)$$

## 4 Finite Element approximation and MITC9 method

In this section, the derivation of a shell finite element for the analysis of multilayered structures is presented. The element is based on both the ESL and LW theories contained



in the Unified Formulation. After an overview in scientific literature about the methods that permit to withstand the membrane and shear locking, the MITC technique has been adopted for this element. Considering a 9-nodes finite element with doubly-curved geometry, the displacement components are interpolated on the nodes of the element by means of the Lagrangian shape functions  $N_i$  [38]:

$$\mathbf{u}_s = N_j \mathbf{u}_{s_j} \quad \delta \mathbf{u}_\tau = N_i \delta \mathbf{u}_{\tau_i} \quad \text{with } i, j = 1, \dots, 9 \quad (23)$$

where  $\mathbf{u}_{s_j}$  and  $\delta \mathbf{u}_{\tau_i}$  are the nodal displacements and their virtual variations. Substituting in the geometrical relations (3) one has:

$$\begin{aligned} \boldsymbol{\epsilon}_p &= F_\tau (\mathbf{D}_p + \mathbf{A}_p) (N_i \mathbf{I}) \mathbf{u}_{\tau_i} \\ \boldsymbol{\epsilon}_n &= F_\tau (\mathbf{D}_{n\Omega} - \mathbf{A}_n) (N_i \mathbf{I}) \mathbf{u}_{\tau_i} + F_{\tau,z} (N_i \mathbf{I}) \mathbf{u}_{\tau_i} \end{aligned} \quad (24)$$

where  $\mathbf{I}$  is the identity matrix.

Considering the local coordinate system  $(\xi, \eta)$ , the MITC shell elements ([39]-[40]) are formulated by using, instead of the strain components directly computed from the displacements, an interpolation of these within each element using a specific interpolation strategy for each component. The corresponding interpolation points, called *tying points*, are shown in Fig. 4 for a nine-nodes element. Note that the transverse normal strain  $\epsilon_{zz}$  is excluded from this procedure and it is directly calculated from the displacements.

The interpolating functions are Lagrangian functions and are arranged in the following arrays:

$$\begin{aligned} N_{m1} &= [N_{A1}, N_{B1}, N_{C1}, N_{D1}, N_{E1}, N_{F1}] \\ N_{m2} &= [N_{A2}, N_{B2}, N_{C2}, N_{D2}, N_{E2}, N_{F2}] \\ N_{m3} &= [N_P, N_Q, N_R, N_S] \end{aligned} \quad (25)$$

From this point on, the subscripts  $m1$ ,  $m2$  and  $m3$  indicate quantities calculated in the points  $(A1, B1, C1, D1, E1, F1)$ ,  $(A2, B2, C2, D2, E2, F2)$  and  $(P, Q, R, S)$ , respectively. Therefore, the strain components are interpolated as follows:

$$\begin{aligned} \boldsymbol{\epsilon}_p &= \begin{bmatrix} \epsilon_{\alpha\alpha} \\ \epsilon_{\beta\beta} \\ \epsilon_{\alpha\beta} \end{bmatrix} = \begin{bmatrix} N_{m1} & 0 & 0 \\ 0 & N_{m2} & 0 \\ 0 & 0 & N_{m3} \end{bmatrix} \begin{bmatrix} \epsilon_{\alpha\alpha_{m1}} \\ \epsilon_{\beta\beta_{m2}} \\ \epsilon_{\alpha\beta_{m3}} \end{bmatrix} \\ \boldsymbol{\epsilon}_n &= \begin{bmatrix} \epsilon_{\alpha z} \\ \epsilon_{\beta z} \\ \epsilon_{zz} \end{bmatrix} = \begin{bmatrix} N_{m1} & 0 & 0 \\ 0 & N_{m2} & 0 \\ 0 & 0 & 1 \end{bmatrix} \begin{bmatrix} \epsilon_{\alpha z_{m1}} \\ \epsilon_{\beta z_{m2}} \\ \epsilon_{zz} \end{bmatrix} \end{aligned} \quad (26)$$

where the strains  $\epsilon_{\alpha\alpha_{m1}}$ ,  $\epsilon_{\beta\beta_{m2}}$ ,  $\epsilon_{\alpha\beta_{m3}}$ ,  $\epsilon_{\alpha z_{m1}}$ ,  $\epsilon_{\beta z_{m2}}$  are expressed by means of eq.s (24) in which the shape functions  $N_i$  and their derivatives are evaluated in the tying points. For example, one can consider the strain component  $\epsilon_{\alpha\alpha}$  that is calculated as follows:

$$\begin{aligned} \epsilon_{\alpha\alpha} &= N_{A1} \epsilon_{\alpha\alpha_{A1}} + N_{B1} \epsilon_{\alpha\alpha_{B1}} + N_{C1} \epsilon_{\alpha\alpha_{C1}} + N_{D1} \epsilon_{\alpha\alpha_{D1}} \\ &\quad + N_{E1} \epsilon_{\alpha\alpha_{E1}} + N_{F1} \epsilon_{\alpha\alpha_{F1}} \end{aligned} \quad (27)$$

with:

$$\epsilon_{\alpha\alpha_{A1}} = N_{i,\alpha}^{(A1)} F_\tau \mathbf{u}_{\tau_i} + \frac{1}{H_\alpha R_\alpha} N_i^{(A1)} F_\tau w_{\tau_i} \quad (28)$$

The superscript (A1) indicates that the shape function and its derivative are evaluated in the point of coordinates  $(-\frac{1}{\sqrt{3}}, -\sqrt{\frac{3}{5}})$ . Similar expressions can be written for  $\epsilon_{\alpha\alpha_{B1}}, \epsilon_{\alpha\alpha_{C1}}, \epsilon_{\alpha\alpha_{D1}}, \epsilon_{\alpha\alpha_{E1}}, \epsilon_{\alpha\alpha_{F1}}$ .

## 5 Governing FEM equations

The PVD for a multilayered doubly-curved shell reads:

$$\int_{\Omega_k} \int_{A_k} \left\{ \delta \boldsymbol{\epsilon}_p^{kT} \boldsymbol{\sigma}_p^k + \delta \boldsymbol{\epsilon}_n^{kT} \boldsymbol{\sigma}_n^k \right\} H_\alpha^k H_\beta^k d\Omega_k dz = \delta L_e \quad (29)$$

where  $\Omega_k$  and  $A_k$  are the integration domains in the plane and in the thickness direction, respectively. The left hand side of the equation represents the variation of the internal work, while the right hand side is the external work.  $\boldsymbol{\sigma}_p^k$  and  $\boldsymbol{\sigma}_n^k$  contain the mechanical (d) and thermal (T) contributions, so:

$$\begin{aligned} \int_{\Omega_k} \int_{A_k} \left\{ \delta \boldsymbol{\epsilon}_p^{kT} (\boldsymbol{\sigma}_{pd}^k - \boldsymbol{\sigma}_{pT}^k) + \delta \boldsymbol{\epsilon}_n^{kT} (\boldsymbol{\sigma}_{nd}^k - \boldsymbol{\sigma}_{nT}^k) \right\} H_\alpha^k H_\beta^k d\Omega_k dz \\ = \delta L_e \end{aligned} \quad (30)$$

In this work no mechanical loads are applied to the shell structure, so the external work is null, except for the thermal stress contribution of the temperature distribution applied, so:

$$\begin{aligned} \int_{\Omega_k} \int_{A_k} \left\{ \delta \boldsymbol{\epsilon}_p^{kT} \boldsymbol{\sigma}_{pd}^k + \delta \boldsymbol{\epsilon}_n^{kT} \boldsymbol{\sigma}_{nd}^k \right\} H_\alpha^k H_\beta^k d\Omega_k dz \\ = \int_{\Omega_k} \int_{A_k} \left\{ \delta \boldsymbol{\epsilon}_p^{kT} \boldsymbol{\sigma}_{pT}^k + \delta \boldsymbol{\epsilon}_n^{kT} \boldsymbol{\sigma}_{nT}^k \right\} H_\alpha^k H_\beta^k d\Omega_k dz \end{aligned} \quad (31)$$

Substituting the constitutive equations (6), the geometrical relations written via the MITC method (26) and applying the Unified Formulation (11) and the FEM approximation (23), one obtains the following governing equations:

$$\delta \mathbf{q}_u^{ktri} : \mathbf{K}_{uu}^{k\tau s i j} \mathbf{q}_u^{k s j} = \boldsymbol{\theta}^{ktri} \quad (32)$$

where  $\mathbf{K}_{uu}^{krsij}$  is a  $3 \times 3$  matrix, called fundamental nucleus of the mechanical stiffness matrix, and its explicit expression is given in [41]. This is the basic element from which the stiffness matrix of the whole structure is computed. The fundamental nucleus is expanded on the indexes  $\tau$  and  $s$  in order to obtain the stiffness matrix of each layer. Then, the matrices of each layer are assembled at multi-layer level depending on the approach considered, ESL or LW.  $\Theta^{kri}$  is a  $3 \times 1$  matrix, called fundamental nucleus of the thermal load, and its explicit expression is given in the following section.  $\mathbf{q}_u^{ksj}$  and  $\delta \mathbf{q}_u^{kri}$  are the nodal displacements and their virtual variation, respectively.

## 6 Heat conduction problem in layered structures

The heat conduction problem is investigated by solving the Fourier heat conduction equation as described in [42] for the plate case. Here the solution is given for the shell case as proposed in [21]. If the values of the temperature are known at the top and bottom surface of the shell, the temperature profile through the thickness can be considered in two different ways. The first method introduces an assumed profile  $\hat{\theta}(z)$  that varies linearly from the top to the bottom as follows:

$$\hat{\theta}(z) = \theta_{bottom} + \frac{\theta_{top} - \theta_{bottom}}{h} * \left( z + \frac{h}{2} \right) \quad z \in \left[ \frac{-h}{2}; \frac{h}{2} \right] \quad (33)$$

Independently by the number of considered layers the linear profile is always the same.

The second one computes  $\hat{\theta}(z)$  by solving the Fourier heat conduction equation. In case of multi-layered structures, in general for the  $k^{th}$  homogeneous orthotropic layer, the differential Fourier equation of heat conduction reads:

$$\left( \frac{K_1^k}{(H_\alpha^k)^2} \right) \frac{\delta^2 \theta}{\delta \alpha^2} + \left( \frac{K_2^k}{(H_\beta^k)^2} \right) \frac{\delta^2 \theta}{\delta \beta^2} + \left( K_3^k \right) \frac{\delta^2 \theta}{\delta z^2} = 0 \quad (34)$$

where  $K_1^k, K_2^k, K_3^k$  are the thermal conductivities coefficients in material coordinates (1, 2, 3) for each orthotropic layer  $k$  and then rotated in the general curvilinear reference system  $(\alpha, \beta, z)$ . In case of multi-layered structures, continuity conditions for the temperature  $\theta$  and the transverse normal heat flux  $q_z$  hold in the thickness direction at each  $k^{th}$  layer interface, reading:

$$\theta_t^k = \theta_b^{k+1} \quad q_{zt}^k = q_{zb}^{k+1} \quad \text{for } k = 1, \dots, N_l - 1 \quad (35)$$

where  $N_l$  is the number of layers in the considered structure. The relationship between the transverse heat flux

and the temperature is given as:

$$q_z^k = K_3^k \frac{\delta \theta}{\delta z} \quad (36)$$

For the  $k^{th}$  layer of the shell structure it is supposed that  $K_1^k, K_2^k, K_3^k$  are constant because in each layer  $H_\alpha^k, H_\beta^k$  are calculated. For each layer both governing equations and boundary conditions are satisfied by assuming the following temperature field:

$$\theta(\alpha, \beta, z) = f(z) \theta_\Omega(\alpha, \beta) \quad (37)$$

where  $\theta_\Omega$  in this paper is considered bi-sinusoidal as follows:

$$\theta_\Omega(\alpha, \beta) = \sin\left(\frac{m\pi\alpha}{a}\right) \sin\left(\frac{n\pi\beta}{b}\right) \quad (38)$$

and  $f(z)$  is assumed as:

$$f(z) = \theta_0 \exp\left(s^k z\right) \quad (39)$$

where  $\theta_0$  is a constant and  $s^k$  a parameter. Substituting 37 in 34 and solving for  $s^k$ :

$$s_{1,2}^k = \pm \sqrt{\frac{\frac{K_1^k}{(H_\alpha^k)^2} \left(\frac{m\pi}{a}\right)^2 + \frac{K_2^k}{(H_\beta^k)^2} \left(\frac{n\pi}{b}\right)^2}{K_3^k}} \quad (40)$$

Therefore:

$$f(z) = \theta_{01}^k \exp\left(s_1^k z\right) + \theta_{02}^k \exp\left(s_2^k z\right)$$

or:

$$f(z) = C_1^k \cosh\left(s_1^k z\right) + C_2^k \sinh\left(s_1^k z\right) \quad (41)$$

The solution for a layer  $k$  can be written as:

$$\theta_c(\alpha, \beta, z) = \theta^k = \left[ C_1^k \cosh\left(s_1^k z\right) + C_2^k \sinh\left(s_1^k z\right) \right] \sin\left(\frac{m\pi\alpha}{a}\right) \sin\left(\frac{n\pi\beta}{b}\right) \quad (42)$$

wherein the coefficients  $C_1^k$  and  $C_2^k$  are constant for each layer  $k$ . In 41 for each layer  $k$  two unknowns ( $C_1^k$  and  $C_2^k$ ) remain. Therefore, if the number of layers is  $N_l$ , the number of unknowns is  $(2 N_l)$  and  $(2 N_l)$  equations to determine the unknowns are needed. The first two conditions are given by the temperature at the top and the bottom of the shell structure:

$$\begin{aligned} f(z_{bottom}) &= \hat{\theta}_{bottom} = C_1^1 \cosh\left(s_1^1 z_{bottom}\right) \\ &\quad + C_2^1 \sinh\left(s_1^1 z_{bottom}\right) \\ f(z_{top}) &= \hat{\theta}_{top} = C_1^{N_l} \cosh\left(s_1^{N_l} z_{top}\right) + C_2^{N_l} \sinh\left(s_1^{N_l} z_{top}\right) \end{aligned} \quad (43)$$



Another  $(N_l - 1)$  equations can be obtained from the continuity of temperature at each layer interface as follows:

$$C_1^k \cosh(s_1^k z_t^k) + C_2^k \sinh(s_1^k z_t^k) - C_1^{k+1} \cosh(s_1^{k+1} z_b^{k+1}) - C_2^{k+1} \sinh(s_1^{k+1} z_b^{k+1}) = 0 \quad (44)$$

and another  $(N_l - 1)$  equations can be obtained from the continuity of heat flux through the interfaces as follows:

$$s_1^k K_3^k [C_1^k \sinh(s_1^k z_t^k) + C_2^k \cosh(s_1^k z_t^k)] - s_1^{k+1} K_3^{k+1} [C_1^{k+1} \sinh(s_1^{k+1} z_b^{k+1}) + C_2^{k+1} \cosh(s_1^{k+1} z_b^{k+1})] = 0 \quad (45)$$

In 44 and 45 subscripts  $t$  and  $b$  indicate the top and bottom of each layer. Solving the system given by 43, 44 and 45 the  $(2 N_l)$  coefficients  $C_1^k$  and  $C_2^k$  are obtained. The temperature amplitude in the thickness shell direction is given by:

$$\hat{\theta}_c(z) = \hat{\theta}^k = C_1^k \cosh(s_1^k z) + C_2^k \sinh(s_1^k z) \quad \text{for } k = 1, \dots, N_l \quad (46)$$

So, the explicit expression of the fundamental nucleus  $\Theta^{k\tau i}$ , is:

$$\Theta_\alpha^{k\tau i} = \lambda_6^k J_\alpha^{k\tau} W_{i,\beta}^k + \lambda_1^k J_\beta^{k\tau} W_{i,\alpha}^k \quad (47)$$

$$\Theta_\beta^{k\tau i} = \lambda_2^k J_\alpha^{k\tau} W_{i,\beta}^k + \lambda_6^k J_\beta^{k\tau} W_{i,\alpha}^k \quad (48)$$

$$\Theta_\alpha^{k\tau i} = \lambda_3^k J_{\alpha\beta}^{k\tau,z} W_i^k + \frac{\lambda_2^k}{R_\beta^k} J_\alpha^{k\tau} W_i^k + \frac{\lambda_1^k}{R_\alpha^k} J_\beta^{k\tau} W_i^k \quad (49)$$

Where the following integrals in the domain  $\Omega_k$  are defined:

$$\left( W_i^k; W_{i,\alpha}^k; W_{i,\beta}^k \right) = \int_{\Omega_k} \left( N_i \theta_\Omega; \frac{\partial N_i}{\partial \alpha} \theta_\Omega; \frac{\partial N_i}{\partial \beta} \theta_\Omega \right) d\alpha_k d\beta_k \quad (50)$$

Moreover, the integrals on the domain  $A_k$ , in the thickness direction, are written as:

$$\left( J_\alpha^{k\tau}; J_\beta^{k\tau}; J_{\alpha\beta}^{k\tau,z} \right) = \int_{A_k} \left( F_\tau \hat{\theta}^k H_\alpha^k; F_\tau \hat{\theta}^k H_\beta^k; \frac{\partial F_\tau}{\partial z} \hat{\theta}^k H_\alpha^k H_\beta^k \right) dz \quad (51)$$

## 7 Acronyms

Several refined and advanced two-dimensional models are contained in the Unified Formulation. Depending on the variables description (LW, ESL) and the order of expansion

$N$  of the displacements in  $z$ , a large variety of kinematics shell theories can be obtained. A system of acronyms is given in order to denote these models. The first letter indicates the multi-layer approach which can be Equivalent Single Layer (ESL) or Layer Wise (LW). The number  $N$  indicates the order of expansion used in the thickness direction (from 1 to 4). In the case of LW approach, the same order of expansion is used for each layer. In the case of ESL approach, a letter  $Z$  can be added if the zig-zag effects of displacements is considered by means of Murakami's zig-zag function. Summarizing, ESL1-ESL4 are ESL models. If Murakami zig-zag function is used, these equivalent single layer models are indicated as ESLZ1-ESLZ3. In the case of layer wise approach, the acronym LW is considered in place of ESL, so the acronyms are LW1-LW4. Sometimes the Navier analytical method is employed in place of the FEM method and a subscript ( $a$ ) is used. In the case of the calculated temperature profile the solutions are indicated by  $T_c$ , while  $T_a$  is used for the solutions deriving from an assumed linear temperature profile.

## 8 Numerical results

This section is composed of two parts. The first one is devoted to the assessment of the shell element based on the Unified Formulation by the static analysis of simply supported plates, cylindrical shells and spherical shells. All of them are evaluated applying a thermal load with a sinusoidal in-plane distribution. Before the assessment results of the static analysis a brief discussion about the evaluation of the temperature profile is given. Using the theory that provides the most accurate results, the second part presents some benchmark solutions relative to plates, cylindrical shells and spherical shells with particular lamination and boundary conditions.

### 8.1 Temperature profile evaluation

The temperature profile along the thickness direction is given in Figures 5-6 for the plate structure and the cylindrical shell panel, for both the assumed linear profile and the calculated one.

For the three layered composite plate structure (see Figure 5) the calculated profile is plotted for different thickness ratios  $a/h$ . It is evident that for thin plates the temperature profile can be assumed almost linear, conversely for thick plates the temperature behavior is very far from the

linear one, and large errors can be committed if the temperature profile is assumed as linear.

For the two layered composite cylindrical shell panel, see Figure 6, the calculated profile is plotted for different thickness to radius ratios  $R/h$  and thickness ratio  $a/h = 10$ . One can note that the effect of the curvature on the distribution of the temperature profile is negligible and the difference with the linear profile is only due to the thickness ratio  $a/h = 10$ .

The proposed evaluations of temperature profile clarify the importance of a calculated temperature profile for thick plates and shells in order to avoid large errors in the approximation of thermal load.

## 8.2 Assessment

To assess the efficiency of this shell element, three reference problems are considered: the first one is a crossply square multilayered plate with lamination  $(0^\circ/90^\circ/0^\circ)$  and simply-supported boundary conditions and is compared with the 3D elasticity solution given by Bhaskar et al. in [43]. The second problem is a square cylindrical panel, analytically analyzed, with lamination  $(0^\circ/90^\circ)$  and simply-supported boundary conditions. The last one is a square spherical panel, analytically analyzed, with lamination  $(0^\circ/90^\circ)$ . The boundary conditions are simply-supported. All of them are analyzed by applying a thermal load with a bi-sinusoidal in-plane distribution:

$$\theta(\alpha, \beta, z) = \hat{\theta}(z) \sin\left(\frac{m\pi\alpha}{a}\right) \sin\left(\frac{n\pi\beta}{b}\right) \quad (52)$$

where  $m = n = 1$ . These three problems are briefly described in the following sections.

### 8.2.1 Multilayered plate

The structure analyzed by Bhaskar et al. [43] (see Figure 3) is a composite multilayered square plate with lamination  $(0^\circ/90^\circ/0^\circ)$ . The physical properties of the material of the plate, composite, are given in Table 1. The geometrical dimensions are:  $a = b = 1\text{ m}$ . The temperature boundary conditions are:  $\hat{\theta}_{top} = 1\text{ K}$ ,  $\hat{\theta}_{bottom} = -1\text{ K}$ . The results are presented for different thickness ratios  $a/h = 2, 10, 50, 100$ . A mesh grid of  $10 \times 10$  elements is taken to ensure the convergence of the solution (see Table 2). The values of the transversal displacement  $w$  and the transverse shear stress  $\sigma_{\alpha z}$  are listed in Table 3 for the temperature profile calculated solving the Fourier heat conduction equation and compared with the assumed linear temperature profile. Other results in terms of transverse shear

stress and transversal displacement are shown in Figures 7-10. All the FEM solutions, with an assumed linear temperature profile, lead to accurate results with respect to the 3D solution [43], that makes use of the same linear profile assumption. The results agree also with the analytical solutions provided for all the thickness ratios, except in the case of FSDT model. Indeed, plate elements that present a constant transverse normal strain such as FSDT lead to inaccurate results for both thick and thin plates. It is confirmed what found in [12]: at least a parabolic expansion for the displacements  $(u, v, w)$  is required to capture the linear thermal strains that are related to a linear through-the-thickness temperature distribution. The results obtained with the calculated temperature profile are close to them of the assumed linear profile for plates with thickness ratios  $a/h = 50, 100$ , while for plates with thickness ratios  $a/h = 2, 10$  the thermal profile is clearly non linear and results are different from the linear cases even if the displacement field is approximated by refined models.

In general, LW theories perform better than ESL ones and generally a lower-order expansion of the displacements is sufficient. Equivalent single layer analyses are quite satisfactory only for the transverse displacement if applied to thin plates  $a/h = 100$ , but not for the solution of the transverse shear stresses, as shown in Figures 7-10. On the other hand, higher-order LW theories lead to better results but computationally more expensive.

**Table 1:** Physical data for multilayered plate, cylindrical and spherical shell.

Material	Composite	Carbon
$\frac{E_L}{E_T}$	25.0	25.0
$\frac{G_{LT}}{G_{TT}}$	$\frac{0.5}{0.2}$	$\frac{0.5}{0.2}$
$\nu$	0.25	0.25
$\frac{\alpha_T}{\alpha_L}$	1125.0	3.0
$\frac{K_L}{K_T}$	$\frac{36.42}{0.96}$	$\frac{36.42}{0.96}$

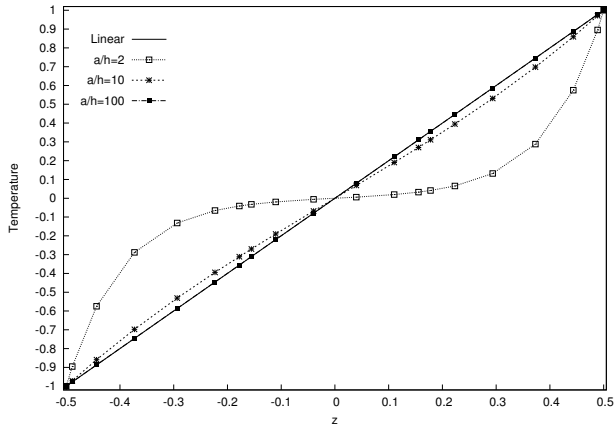


Figure 5: Temperature Profiles for different thickness ratios ( $a/h$ ). Composite plate.

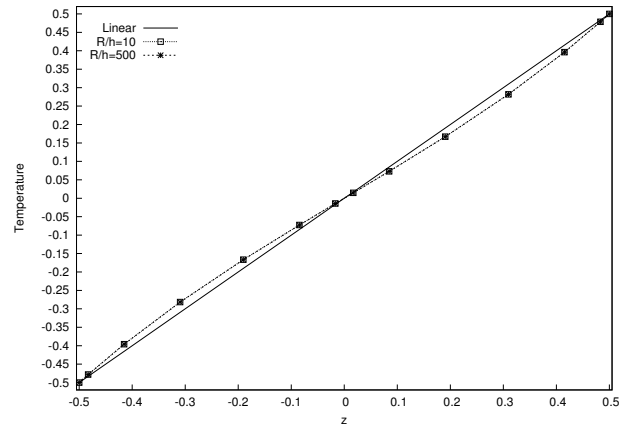


Figure 6: Temperature Profiles for different radius to thickness ratios ( $R/h$ ). Composite cylinder.

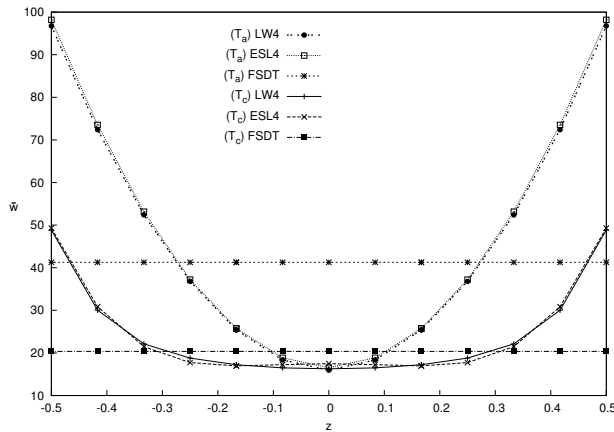


Figure 7: Transverse displacement  $w$  along the thickness, with thickness ratio ( $a/h$ ) = 2. Composite plate.

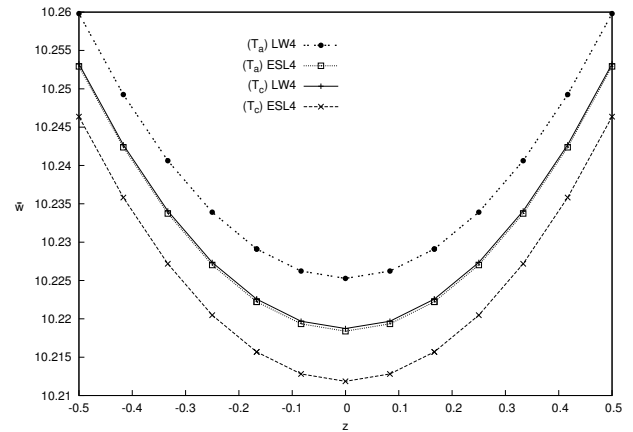


Figure 8: Transverse displacement  $w$  along the thickness, with thickness ratio ( $a/h$ ) = 100. Composite plate.

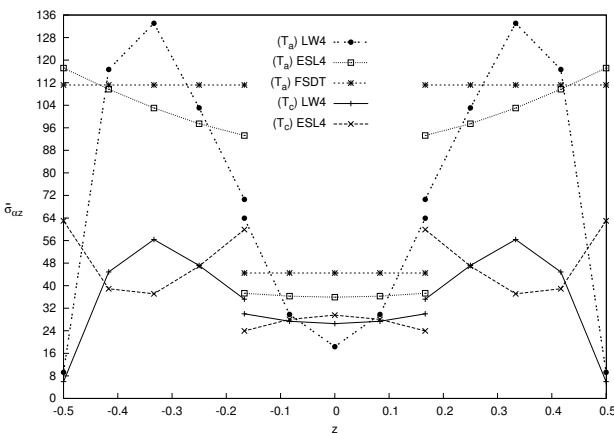


Figure 9: Transverse shear stress  $\sigma_{az}$  along the thickness, with thickness ratio ( $a/h$ ) = 2. Composite plate.

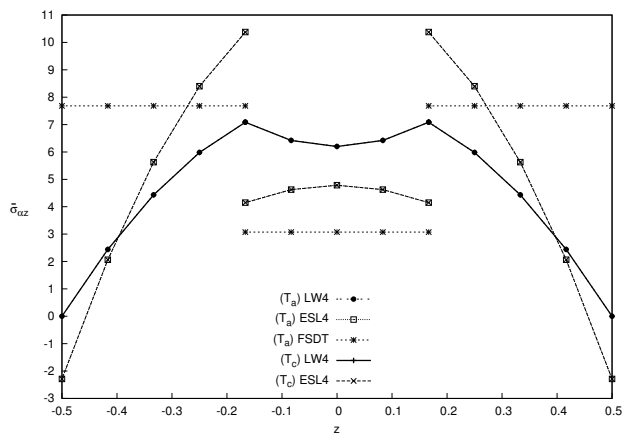


Figure 10: Transverse shear stress  $\sigma_{az}$  along the thickness, with thickness ratio ( $a/h$ ) = 100. Composite plate.

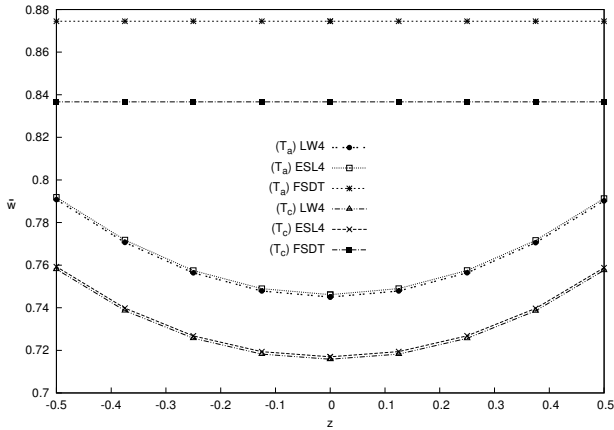


Figure 11: Transverse displacement  $w$  along the thickness, with radius to thickness ratio  $(R/h) = 10$ . Composite cylindrical panel.

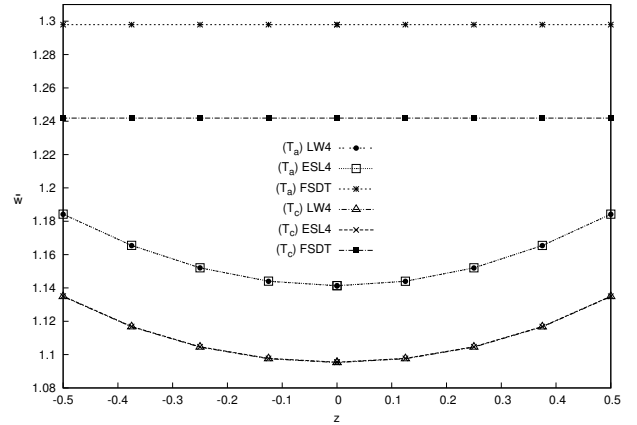


Figure 12: Transverse displacement  $w$  along the thickness, with radius to thickness ratio  $(R/h) = 500$ . Composite cylindrical panel.

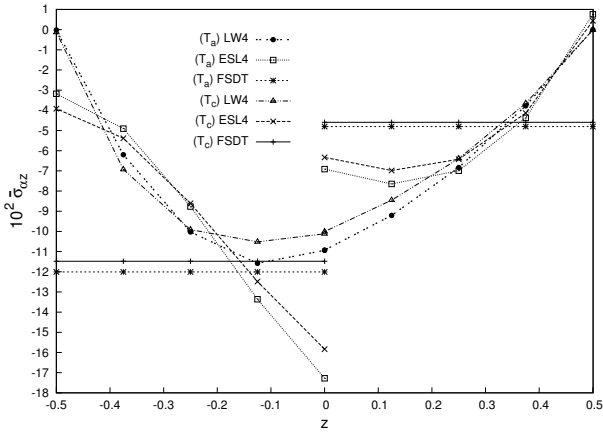


Figure 13: Transverse shear stress  $\sigma_{az}$  along the thickness, with radius to thickness ratio  $(R/h) = 10$ . Composite cylindrical panel.

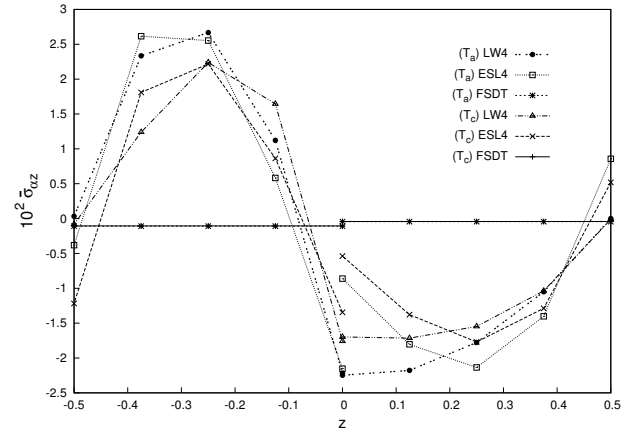
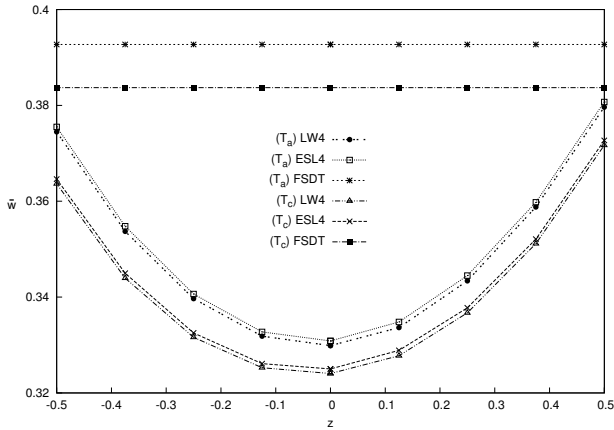


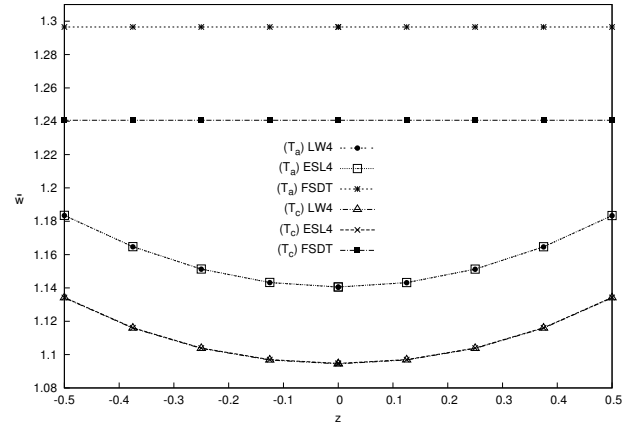
Figure 14: Transverse shear stress  $\sigma_{az}$  along the thickness, with radius to thickness ratio  $(R/h) = 500$ . Composite cylindrical panel.

Table 2: Convergence study. Plate with thickness ratio  $a/h = 100$ , cylindrical panel and spherical panel with radius to thickness ratio  $R/h = 500$ . All the cases are computed for the calculated temperature profile  $T_c$  and with a LW4 theory.

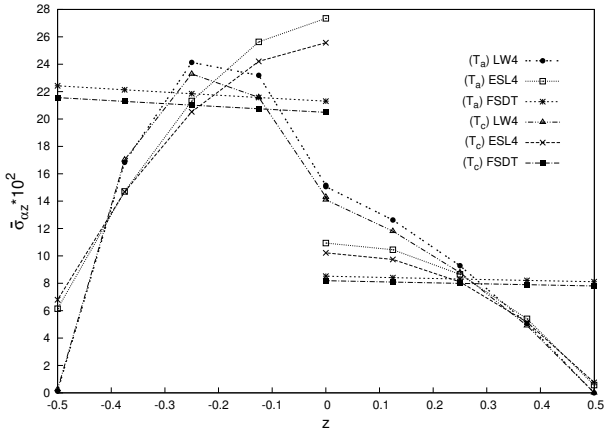
	Mesh	$4 \times 4$	$6 \times 6$	$8 \times 8$	$10 \times 10$	<i>Analytical</i>
<i>Plate</i>	$w$	10.27	10.26	10.25	10.25	10.25
	$\sigma_{xz}$	7.466	7.213	7.102	7.084	7.069
<i>Cylindrical</i>	$w$	1.0966	1.0955	1.0953	1.0952	1.0953
	$\sigma_{az}$	-1.7090	-1.7131	-1.7093	-1.6983	-1.6957
<i>Spherical</i>	$w$	1.0958	1.0948	1.0946	1.0945	1.0945
	$\sigma_{az}$	-2.2848	-2.2065	-2.1562	-2.1461	-2.1403



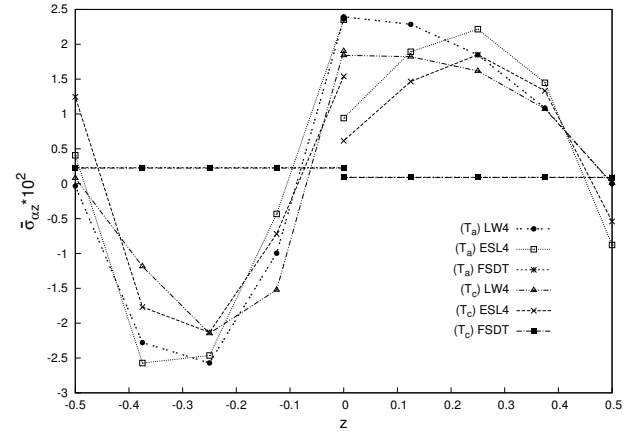
**Figure 15:** Transverse displacement  $w$  along the thickness, with radius to thickness ratio  $(R/h) = 10$ . Composite spherical panel.



**Figure 16:** Transverse displacement  $w$  along the thickness, with radius to thickness ratio  $(R/h) = 500$ . Composite spherical panel.



**Figure 17:** Transverse shear stress  $\sigma_{\alpha z}$  along the thickness, with radius to thickness ratio  $(R/h) = 10$ . Composite spherical panel.



**Figure 18:** Transverse shear stress  $\sigma_{\alpha z}$  along the thickness, with radius to thickness ratio  $(R/h) = 500$ . Composite spherical panel.

### 8.2.2 Multilayered cylindrical panel

In this section, a cylindrical composite panel with lamination  $(0^\circ/90^\circ)$  is analyzed (see Figure 2). The lamination angle is  $0^\circ$  for the bottom layer and  $90^\circ$  for the top layer. The geometrical dimensions are:  $a = 1$  m and  $b = 1$  m, global thickness  $h_{tot} = 0.1$  m, curvature radius  $R_\alpha = \infty$ . The physical properties of the carbon are given in Table 1. The temperature boundary conditions are:  $\hat{\theta}_{top} = 0.5$  K,  $\hat{\theta}_{bottom} = -0.5$  K for all the cases. The results are compared with the corresponding closed form solutions obtained with the Navier method and are presented for different radius to thickness ratios  $R_\beta/h_{tot} = 10, 50, 100, 500$  with the corresponding curvature radii  $R_\beta = 1, 5, 10, 50$ . A mesh grid of  $10 \times 10$  elements is taken to ensure the convergence of the solution (see Table 2). The values of the transversal displacement  $w$  and the transverse shear

stress  $\sigma_{\alpha z}$  are listed in Table 4 for the temperature profile calculated solving the Fourier’s heat conduction equation and compared with the assumed linear temperature profile. Other results in terms of transverse shear stress and transversal displacement are shown in Figures 11-14.

All the FEs, in both calculated and assumed linear cases, lead to accurate results with respect to the analytical solutions for all the thickness ratios, except for FSDT elements. The difference between the calculated temperature profile and the assumed linear one is a constant and it is not affected by the curvature of the cylinder  $R_\beta$ ; this difference is due to the thickness ratio which is  $a/h = 10$ . In general, LW theories perform better than ESL ones and often also with a lower-order expansion of the unknowns. Equivalent single layer analyses are quite satisfactory for the transverse displacement, even when lower radii to thick-

**Table 3:** Plate with lamination (0°/90°/0°). Transverse displacement  $w = w(a/2, b/2) * h_{tot}$ , evaluated at  $z = \pm h/2$ . Transverse shear stress  $\sigma_{xz} = \sigma_{xz}(a, 0)$ , evaluated at  $z = +h/6$ .

		$a/h$	2	10	50	100	
$w$	3D[43]		96.79	17.39	10.50	10.26	
$\sigma_{xz}$	3D[43]		63.92	60.54	14.07	7.073	
$w$	LW4 <sub>a</sub>	$T_a$	96.78	17.39	10.50	10.26	
		$T_c$	49.09	16.39	10.47	10.25	
	LW4	$T_a$	96.77	17.39	10.50	10.26	
		$T_c$	48.85	16.39	10.47	10.25	
	LW1	$T_a$	89.23	17.62	11.14	10.91	
		$T_c$	44.17	16.69	11.11	10.91	
	ESLZ3	$T_a$	94.85	17.37	10.50	10.26	
		$T_c$	50.08	16.41	10.47	10.25	
	ESL4 <sub>a</sub>	$T_a$	98.21	16.90	10.47	10.25	
		$T_c$	49.55	15.93	10.44	10.25	
	ESL4	$T_a$	98.20	16.90	10.47	10.25	
		$T_c$	49.29	15.93	10.44	10.25	
	ESL2	$T_a$	83.45	14.96	10.38	10.23	
		$T_c$	40.87	14.09	10.35	10.22	
	FSDT	$T_a$	41.27	18.33	15.17	15.06	
		$T_c$	20.35	17.26	15.13	15.05	
	$\sigma_{xz}$	LW4 <sub>a</sub>	$T_a$	63.82	60.54	14.07	7.073
			$T_c$	30.11	57.07	14.04	7.069
		LW4	$T_a$	63.93	60.66	14.10	7.088
			$T_c$	30.00	57.18	14.07	7.084
		LW1	$T_a$	42.54	58.78	13.69	6.883
			$T_c$	31.69	56.35	13.21	6.879
		ESLZ3	$T_a$	27.42	52.61	12.45	6.263
			$T_c$	23.42	50.43	12.43	6.260
ESL4 <sub>a</sub>		$T_a$	37.25	36.33	8.251	4.143	
		$T_c$	24.04	34.47	8.232	4.140	
ESL4		$T_a$	37.30	36.41	8.268	4.152	
		$T_c$	23.96	34.55	8.250	4.149	
ESL2		$T_a$	11.58	16.21	3.624	1.819	
		$T_c$	6.065	15.31	3.616	1.818	
FSDT		$T_a$	44.48	28.00	6.127	3.073	
		$T_c$	22.09	26.41	6.112	3.071	

ness ratios are considered ( $R/h = 10$ ), but not for the transverse shear stress, as shown in Figures 11-14.



**Table 4:** Cylindrical panel with lamination ( $0^\circ/90^\circ$ ). Transverse displacement  $w = w(a/2, b/2)$ , transverse shear stress  $\sigma_{az} = \sigma_{az}(a, 0) \cdot 10^2$ , evaluated at  $z = 0$ .

		$R_\beta/h$	10	50	100	500	
<i>w</i>	<i>LW4<sub>a</sub></i>	<i>T<sub>a</sub></i>	0.7450	1.1192	1.1359	1.1412	
		<i>T<sub>c</sub></i>	0.7188	1.0748	1.0904	1.0953	
	<i>LW4</i>	<i>T<sub>a</sub></i>	0.7450	1.1192	1.1359	1.1412	
		<i>T<sub>c</sub></i>	0.7158	1.0743	1.0902	1.0952	
	<i>LW1</i>	<i>T<sub>a</sub></i>	0.7712	1.1538	1.1706	1.1759	
		<i>T<sub>c</sub></i>	0.7412	1.1082	1.1243	1.1293	
	<i>ESLZ3</i>	<i>T<sub>a</sub></i>	0.7454	1.1177	1.1342	1.1396	
		<i>T<sub>c</sub></i>	0.7147	1.0717	1.0875	1.0926	
	<i>ESL4<sub>a</sub></i>	<i>T<sub>a</sub></i>	0.7461	1.1194	1.1360	1.1413	
		<i>T<sub>c</sub></i>	0.7199	1.0751	1.0907	1.0955	
	<i>ESL4</i>	<i>T<sub>a</sub></i>	0.7461	1.1194	1.1360	1.1413	
		<i>T<sub>c</sub></i>	0.7170	1.0746	1.0904	1.0955	
	<i>ESL2</i>	<i>T<sub>a</sub></i>	0.7455	1.1152	1.1316	1.1369	
		<i>T<sub>c</sub></i>	0.7150	1.0695	1.0852	1.0902	
	<i>FSDT</i>	<i>T<sub>a</sub></i>	0.8745	1.2781	1.2941	1.2979	
		<i>T<sub>c</sub></i>	0.8367	1.2229	1.2382	1.2419	
	$\sigma_{az}$	<i>LW4<sub>a</sub></i>	<i>T<sub>a</sub></i>	-10.901	-3.7541	-2.8789	-2.2428
			<i>T<sub>c</sub></i>	-10.051	-3.1516	-2.3086	-1.6957
<i>LW4</i>		<i>T<sub>a</sub></i>	-10.923	-3.7615	-2.8845	-2.2471	
		<i>T<sub>c</sub></i>	-10.011	-3.1485	-2.3086	-1.6983	
<i>LW1</i>		<i>T<sub>a</sub></i>	-8.3115	-4.0011	-3.5188	-3.1781	
		<i>T<sub>c</sub></i>	-7.7544	-3.6140	-3.1507	-2.8235	
<i>ESLZ3</i>		<i>T<sub>a</sub></i>	-10.522	-3.5686	-2.7832	-2.2277	
		<i>T<sub>c</sub></i>	-9.7816	-3.1176	-2.3651	-1.8329	
<i>ESL4<sub>a</sub></i>		<i>T<sub>a</sub></i>	-6.8978	-1.7276	-1.2097	-0.8599	
		<i>T<sub>c</sub></i>	-6.3568	-1.3735	-0.8747	-0.5374	
<i>ESL4</i>		<i>T<sub>a</sub></i>	-6.9120	-1.7309	-1.2120	-0.8614	
		<i>T<sub>c</sub></i>	-6.3345	-1.3701	-0.8732	-0.5377	
<i>ESL2</i>		<i>T<sub>a</sub></i>	-5.6195	-1.7814	-1.4294	-1.2006	
		<i>T<sub>c</sub></i>	-5.3090	-1.6296	-1.2921	-1.0727	
<i>FSDT</i>		<i>T<sub>a</sub></i>	-4.8037	-0.6032	-0.2571	-0.0436	
		<i>T<sub>c</sub></i>	-4.5916	-0.5767	-0.2457	-0.0418	

**Table 5:** Spherical panel with lamination ( $0^\circ/90^\circ$ ). Transverse displacement  $w = w(a/2, b/2)$ , evaluated at  $z = 0$ , transverse shear stress  $\sigma_{az} = \sigma_{az}(a, 0) * 10^2$  evaluated at  $z = -h/4$ .

$R/h$		10	50	100	500		
$w$	$LW_{4a}$	$T_a$	0.3299	1.0507	1.1174	1.1404	
		$T_c$	0.3203	1.0087	1.0725	1.0945	
	$LW_4$	$T_a$	0.3299	1.0507	1.1174	1.1405	
		$T_c$	0.3240	1.0091	1.0726	1.0945	
	$LW_1$	$T_a$	0.3386	1.0836	1.1516	1.1751	
		$T_c$	0.3325	1.0414	1.1062	1.1285	
	$ESLZ_3$	$T_a$	0.3306	1.0496	1.1159	1.1388	
		$T_c$	0.3235	1.0071	1.0701	1.0918	
	$ESL_{4a}$	$T_a$	0.3309	1.0511	1.1176	1.1406	
		$T_c$	0.3213	1.0093	1.0728	1.0947	
	$ESL_4$	$T_a$	0.3309	1.0511	1.1176	1.1406	
		$T_c$	0.3250	1.0096	1.0729	1.0947	
	$ESL_2$	$T_a$	0.3315	1.0477	1.1134	1.1361	
		$T_c$	0.3248	1.0054	1.0679	1.0895	
	$FSDT$	$T_a$	0.3927	1.1967	1.2709	1.2965	
		$T_c$	0.3837	1.1459	1.2163	1.2406	
	$\sigma_{az}$	$LW_{4a}$	$T_a$	24.096	1.1199	-1.3854	-2.5674
			$T_c$	23.379	1.3972	-1.0041	-2.1403
$LW_4$		$T_a$	24.131	1.1212	-1.3877	-2.5714	
		$T_c$	23.289	1.3831	-1.0122	-2.1461	
$LW_1$		$T_a$	19.500	1.5598	-0.5309	-1.5931	
		$T_c$	18.818	1.6718	-0.3315	-1.3536	
$ESLZ_3$		$T_a$	21.061	2.4154	0.1275	-1.0771	
		$T_c$	20.328	2.5096	0.3181	-0.8407	
$ESL_{4a}$		$T_a$	21.281	0.8662	-1.3842	-2.4603	
		$T_c$	20.593	1.0635	-1.0932	-2.1270	
$ESL_4$		$T_a$	21.312	0.8673	-1.3865	-2.4641	
		$T_c$	20.521	1.0521	-1.1000	-2.1325	
$ESL_2$		$T_a$	20.284	3.0723	0.9925	-0.0983	
		$T_c$	19.598	3.1406	1.1473	0.0970	
$FSDT$		$T_a$	21.845	3.9515	1.5581	0.2307	
		$T_c$	21.011	3.8024	1.5047	0.2240	

**Table 6:** Benchmark problems. Case 1 with anti-symmetric lamination  $\pm 45^\circ$ . Plate, cylindrical and spherical panel, transverse displacement  $w = 10 w(a/2, b/2, +h/2)$  and transverse shear stress  $\sigma_{az} = 10^2 \sigma_{az}(a, b/2, 0)$ .

		Plate		Cylindrical		Spherical			
		$a/h$	10	100	$R/h$	10	100	10	100
$w$	LW4	$T_a$	1375.4	4607.8	4.8796	7.4058	2.6295	7.4971	
		$T_c$	1293.5	4604.7	4.5741	7.0165	2.5533	7.1029	
	LW1	$T_a$	1274.3	4895.2	4.7358	7.5077	2.4930	7.6005	
		$T_c$	1203.2	4892.2	4.4477	7.1343	2.4305	7.2202	
	ESLZ3	$T_a$	1307.7	4586.4	4.7120	7.3351	2.5257	7.4200	
		$T_c$	1232.7	4583.4	4.4154	6.9528	2.4542	7.0325	
	ESL4	$T_a$	1250.8	4550.1	4.7274	7.3978	2.5388	7.4776	
		$T_c$	1177.8	4547.1	4.4239	7.0045	2.4615	7.0793	
	ESL2	$T_a$	1164.1	4501.7	4.7187	7.2171	2.5756	7.3229	
		$T_c$	1096.3	4498.8	4.4192	6.8330	2.5029	6.9316	
	FSDT	$T_a$	1294.9	7222.5	5.0729	7.5539	2.6941	7.6548	
		$T_c$	1219.4	7217.9	4.8079	7.2151	2.6483	7.3099	
$\sigma_{az}$	LW4	$T_a$	-85.932	-56.181	15.624	2.4665	24.592	3.7828	
		$T_c$	-81.708	-56.148	14.483	1.9577	22.964	3.2169	
	LW1	$T_a$	-98.335	-56.066	17.223	5.5156	23.450	6.7916	
		$T_c$	-93.436	-56.033	16.299	5.1621	22.161	6.3780	
	ESLZ3	$T_a$	-97.289	-55.800	17.350	2.6837	26.823	4.0857	
		$T_c$	-92.420	-55.767	16.512	2.5503	25.369	3.8853	
	ESL4	$T_a$	-80.357	-52.295	16.651	0.3544	27.067	1.8573	
		$T_c$	-76.298	-52.264	15.711	0.2248	25.347	1.6474	
	ESL2	$T_a$	-61.532	-37.094	15.936	2.8891	23.900	4.2028	
		$T_c$	-58.089	-37.072	15.343	2.9668	22.689	4.2054	
	FSDT	$T_a$	-92.857	-45.626	12.861	2.2035	20.014	3.2885	
		$T_c$	-87.595	-45.598	12.219	2.0645	18.994	3.0974	

### 8.2.3 Multilayered spherical panel

In this section, a square, spherical panel is analysed (see Figure 1). The temperature boundary conditions are:  $\hat{\theta}_{top} = 0.5$  K,  $\hat{\theta}_{bottom} = -0.5$  K for all the cases. The results are compared with the analytical solutions obtained with the Navier method. A mesh grid of  $10 \times 10$  elements is taken to ensure the convergence of the solution (see Table 2). The geometrical dimensions are:  $a = 1$  m and  $b = 1$  m, global thickness  $h_{tot} = 0.1$  m and curvature radii  $R_\alpha = R_\beta = R$ . The physical properties of the carbon are given in Table 1. The results are presented for different radius to thickness ratios  $R/h_{tot} = 10, 50, 100, 500$  with the corresponding

curvature radius  $R = 1.0, 5.0, 10.0, 50.0$ ). The lamination angle is  $0^\circ$  for the bottom layer and  $90^\circ$  for the top layer. The values of the transversal displacement  $w$  and the transverse shear stress  $\sigma_{az}$  are listed in Table 5 for the temperature profile calculated solving the Fourier heat conduction equation and compared with the assumed linear temperature profile. Other results in terms of transverse shear stress and transversal displacement are shown in Figures 15-18.

Considerations similar to the previous problem can be made. All the FEs, in both calculated and assumed linear cases, lead to accurate results with respect to the analytical solutions for all the thickness ratios, except for FSDT

**Table 7:** Benchmark problems. Case 2 with clamped-free boundary condition. Plate, cylindrical and spherical panel, transverse displacement  $w = 10 w(a/2, b/2, +h/2)$  and transverse shear stress  $\sigma_{\alpha z} = 10^2 \sigma_{\alpha z}(a, b/2, 0)$ .

		Plate		Cylindrical		Spherical				
		$a/h$	10	100	$R/h$	10	100	10	100	
$w$	LW4	$T_a$	1226.8	3096.9	4.5476	5.3307	-0.3479	3.7373		
		$T_c$	1157.7	3094.9	4.3447	5.1018	-0.1750	3.6265		
	LW1	$T_a$	1210.9	3274.4	4.6164	5.3874	-0.3642	3.8575		
		$T_c$	1145.2	3272.3	4.4100	5.1571	-0.1923	3.7404		
	ESLZ3	$T_a$	1222.0	3096.8	4.5239	5.2945	-0.3407	3.7334		
		$T_c$	1155.2	3094.9	4.3237	5.0705	-0.1681	3.6250		
	ESL4	$T_a$	1183.3	3083.9	4.5479	5.3215	-0.3423	3.7477		
		$T_c$	1116.2	3081.9	4.3460	5.0941	-0.1696	3.6368		
	ESL2	$T_a$	986.40	3048.6	4.5152	5.2473	-0.3369	3.7263		
		$T_c$	929.51	3046.7	4.3093	5.0195	-0.1652	3.6119		
	FSDT	$T_a$	1039.7	4356.4	4.7747	5.5564	-0.5836	3.9916		
		$T_c$	979.72	4353.6	4.5553	5.3129	-0.4107	3.8624		
$\sigma_{\alpha z}$	LW4	$T_a$	-2619.8	-522.26	-3.9875	-5.1179	15.086	0.2373		
		$T_c$	-2519.9	-522.01	-3.9154	-5.0207	13.754	-0.0535		
	LW1	$T_a$	-2655.5	-579.97	-1.5318	-2.3709	14.765	1.8361		
		$T_c$	-2520.6	-579.67	-1.5150	-2.3397	13.594	1.5630		
	ESLZ3	$T_a$	-2436.3	-537.18	-2.9825	-3.9052	14.614	0.8821		
		$T_c$	-2303.2	-536.90	-2.8354	-3.7432	13.481	0.6988		
	ESL4	$T_a$	-2695.3	-465.54	-3.3266	-4.2840	14.134	0.4614		
		$T_c$	-2582.7	-465.30	-3.2708	-4.2082	12.905	0.1929		
	ESL2	$T_a$	-2287.1	-240.08	-2.8190	-3.6334	13.936	0.7940		
		$T_c$	-2159.2	-239.94	-2.7079	-3.5120	12.809	0.5937		
	FSDT	$T_a$	-4195.5	-446.19	-6.0953	-6.9657	13.570	-2.0926		
		$T_c$	-3955.6	-445.91	-5.7896	-6.6468	12.506	-2.1219		

elements. The difference between the calculated temperature profile solutions and the assumed linear one is constant and it is not affected by the curvature  $R_\beta$ . In general, LW theories perform better than ESL ones and generally a lower-order expansion of the unknowns is sufficient. Equivalent single layer analyses are quite satisfactory only for the transverse displacement, even for lower radii to thickness ratios ( $R/h = 10$ ). Figures 15-18 show a different behavior for the transverse shear stress, where higher-order LW models are required to get accurate results.

### 8.3 FEM benchmark solutions

Similar plates, cylindrical shells and spherical shells are analyzed, considering two new problems that have not reference analytical solutions:

1. Structures with lamination  $\pm 45^\circ$  under bisinusoidal load and simply-supported boundary conditions.
2. Structures with clamped-free boundary conditions: edges parallel to  $\beta$ -direction clamped and those parallel to  $\alpha$ -direction free. The lamination is equal to the assessment cases.

### 8.3.1 Anti-symmetric lamination $\pm 45^\circ$

The first structure analyzed is a composite multilayered square plate with lamination  $(-45^\circ/45^\circ/-45^\circ)$ . The physical properties of the material, the geometrical data and the temperature boundary conditions are the same of the assessment cases. The structure is simply supported. The results are presented for different thickness ratios  $a/h = 10, 100$ . The same mesh grid of  $10 \times 10$  elements of the assessment cases is taken to ensure the convergence of the solution. The values of the transversal displacement  $w$  and the transverse shear stress  $\sigma_{\alpha z}$  are listed in Table 6.

The second structure analyzed is a composite square cylindrical panel with lamination  $(-45^\circ/45^\circ)$ . The lamination angle is  $-45^\circ$  for the bottom layer and  $45^\circ$  for the top layer. The physical properties of the material, the geometrical data and the temperature boundary conditions are the same of the assessment cases. The structure is simply supported. The results are presented for different radius to thickness ratios  $R/h = 10, 100$ . The same mesh grid of  $10 \times 10$  elements of the assessment cases is taken to ensure the convergence of the solution. The values of the transversal displacement  $w$  and the transverse shear stress  $\sigma_{\alpha z}$  are listed in Table 6.

The last structure analyzed is a composite square spherical panel with lamination  $(-45^\circ/45^\circ)$ . The lamination angle is  $-45^\circ$  for the bottom layer and  $45^\circ$  for the top layer. The physical properties of the material, the geometrical data and the temperature boundary conditions are the same of the assessment cases. The structure is simply supported. The results are presented for different radius to thickness ratios  $R/h = 10, 100$ . The same mesh grid of  $10 \times 10$  elements of the assessment cases is taken to ensure the convergence of the solution. The values of the transversal displacement  $w$  and the transverse shear stress  $\sigma_{\alpha z}$  are listed in Table 6.

### 8.3.2 Clamped-free boundary conditions

In this part, the structures are considered with clamped-free boundary conditions: edges parallel to  $\beta$ -direction clamped and those parallel to  $\alpha$ -direction free.

The first structure analyzed is a composite multilayered square plate. The physical properties of the material, the lamination angle, the geometrical data and the temperature boundary conditions are the same of the assessment cases. The results are presented for different thickness ratios  $a/h = 10, 100$ . The same mesh grid of  $10 \times 10$  elements of the assessment cases is taken to ensure the convergence

of the solution. The values of the transversal displacement  $w$  and the transverse shear stress  $\sigma_{\alpha z}$  are listed in Table 7.

The second structure analyzed is a composite square cylindrical panel. The physical properties of the material, the lamination angle, the geometrical data and the temperature boundary conditions are the same of the assessment cases. The results are presented for different radius to thickness ratios  $R/h = 10, 100$ . The same mesh grid of  $10 \times 10$  elements of the assessment cases is taken to ensure the convergence of the solution. The values of the transversal displacement  $w$  and the transverse shear stress  $\sigma_{\alpha z}$  are listed in Table 7.

The last structure analyzed is a composite square spherical panel. The physical properties of the material, the lamination angle, the geometrical data and the temperature boundary conditions are the same of the assessment cases. The results are presented for different radius to thickness ratios  $R/h = 10, 100$ . The same mesh grid of  $10 \times 10$  elements of the assessment cases is taken to ensure the convergence of the solution. The values of the transversal displacement  $w$  and the transverse shear stress  $\sigma_{\alpha z}$  are listed in Table 7.

## 9 Conclusions

This paper has dealt with the static analysis of composite shells by means of a finite element based on the Unified Formulation by Carrera. An assessment of the element has been performed by analyzing simply-supported cross-ply plates, cylindrical and spherical shells under bi-sinusoidal thermal load with both calculated thermal profile (solving the Fourier heat conduction equation) and assumed linear temperature profile. The results have been presented in terms of both transversal displacements and transverse shear stresses, for various thickness ratios and curvature ratios. The performances of the shell element have been tested, and the different theories (classical and refined) contained in the CUF have been compared. The conclusions that can be drawn are the following:

1. The shell element is locking free, for all the LW and ESL models considered. The results converge to the reference solution by increasing both the mesh and the order of expansion of the displacements in the thickness direction.
2. LW models work better than ESLZ theories, and these last perform better than ESL models in thick shell cases.

3. The classical models, such as FSDT, lead to erroneous results in the analysis of thermal problem, even in the case of thin shells.
4. The use of LW models produces better results for both thick and thin shells. Their use becomes mandatory if an accurate description of transverse stresses along the thickness and fulfillment of interlaminar continuity conditions are required.

Finally, some benchmark solutions have been calculated for plates, cylindrical and spherical shells that have not analytical reference solutions: lamination ( $-45^\circ/45^\circ$ ) and clamped-free boundary conditions have been considered. The results have been presented in terms of transversal displacement and transverse shear stresses in the form of tables.

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