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# Guidelines and Recommendations on the Use of Higher Order Finite Elements for Bending Analysis of Plates

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**This paper compares and evaluates various plate finite elements to analyse the static response of thick and thin plates subjected to different loading and boundary conditions. Plate elements are based on different assumptions for the displacement distribution along the thickness direction. Classical (Kirchhoff and Reissner-Mindlin), refined (Reddy and Kant), and other higher-order displacement fields are implemented up to fourth-order expansion. The Unified Formulation UF by the first author is used to derive finite element matrices in terms of fundamental nuclei which consist of  $3 \times 3$  arrays. The MITC4 shear-locking free type formulation is used for the FE approximation. Accuracy of a given plate element is established in terms of the error vs. thickness-to-length parameter. A significant number of finite elements for plates are implemented and compared using displacement and stress variables for various plate problems. Reduced models that are able to detect the 3D solution are built and a Best Plate Diagram (BPD) is introduced to give guidelines for the construction of plate theories based on a given accuracy and number of terms. It is concluded that the UF is a valuable tool to establish, for a given plate problem, the most accurate FE able to furnish results within a certain accuracy range. This allows us to obtain guidelines and recommendations in building refined elements in the bending analysis of plates for various geometries, loadings, and boundary conditions.**

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**Keywords** Finite element, Refined theories, Plate model, Unified formulation

## 1. INTRODUCTION

The development of plate models represents one of the most important issues of structural analysis. The use of two-dimensional models, such as plates and shells, is traditionally

preferred to the more computationally expensive three-dimensional models. Best-known papers are those by Euler [1], Bernoulli [2], Cauchy [3], Poisson [4], Kirchhoff [5], Saint-Venant [6], Love [7], Reissner [8], Mindlin [9], and Vlasov [10]. The three-dimensional deformation state is usually simplified by means of “axiomatic” hypotheses based on conjecture: the cross-section remains plane, the section/thickness deformation can be discarded, shear strains are negligible, etc. . . . Excellent reviews, which give a complete overview of existing Layer-Wise and Equivalent Single Layer theories are those by Ambartsumian [11], Librescu and Reddy [12], Grigolyuk and Kolikov [13], Kapania and Raciti [14, 15], Kapania [16], Noor [17–19], Reddy and Robbins [20], Carrera [21, 22], Qatu [23, 24], and the books by Librescu [25], Reddy [26], and Qatu [27].

A different approach to build a structural model is based on the so-called “asymptotic” method where approximated theories are defined by employing asymptotic-type expansions of unknown variables over the section (beam case), or thickness (plate/shell geometries). Well-known papers on this topic are those by Cicala [28], Fettahlioglu and Steele [29], Berdishevsky [30, 31], Widera et al. [32, 33], and Spencer et al. [34], and the monograph by Cicala [35] and Goldenweizer [36].

The axiomatic and asymptotic techniques have important advantages and drawbacks. The former does not require complex mathematical tools but can lead to quite cumbersome models and the convergence of increasing order terms to the 3D solution cannot be guaranteed. The latter represents a systematic and powerful tool but, especially in the case of multilayered structures, requires the analysis of several parameters to build the model (e.g. the length-to-thickness, stacking sequences, orthotropic ratios). Moreover, the convergence to 3D solutions is guaranteed when one of these parameters vanishes with a consequent lack of information regarding the analysis of thick structures, for instance. This work consists of a companion paper by Carrera and Petrolo [37] where a so-called asymptotic/axiomatic approach has been recently proposed on the basis of closed-form solutions. Here it was shown how asymptotic-like results can

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be obtained starting from a higher-order axiomatic-based model and considering a wide variety of parameters. The present paper extends the technique proposed to the finite element method in order to be able to deal with arbitrary geometries, boundary conditions, and loadings.

The advent of the computational mechanics and the finite element method has enabled a vast number of problems related to complicated geometries and boundary conditions to be solved. Various plate elements are present in open literature, and a brief overview of recent papers is herein conducted. Ganapathi et al. [38] developed a new eight-node  $C^0$  membrane element based on the Reissner-Mindlin plate theory to analyze moderately thick laminates. Polit and Touratier [39] presented a  $C^1$  six-node triangular finite element for geometrical linear and non-linear elastic multilayered composite plates that are able to model both thin and moderately thick plates without the classical pathologies of finite elements. Touratier and Faye [40] proposed elements for moderately thick plates and curved shells. Carrera and Demasi [41, 42] developed a plate element for the accurate description of stress and strain fields in multilayered thick plates subject to static loadings by exploiting two different variational statements: the principle of virtual displacement (PVD) and the Reissner mixed variational theorem (RMVT). The derivation of finite element matrices and numerical assessments were presented. Complete overviews on this topic can be found in the articles by Kant and Swaminathan [43] and Carrera [44].

This work is embedded in the framework of the Carrera Unified Formulation (CUF), which was introduced by the first author during the last decade and deals with refined beam, plate, and shell models. More details and assessments can be found in Carrera et al. [44–47]. CUF allows us to deal with any-order models by considering the order as an input of the analysis since the governing equations, in both strong and weak form, are given stemming from a few “fundamental nuclei” whose form does not depend on either the order of the introduced approximations or on the choices made for the base functions in the thickness direction (for plates/shells) or over the section (for beams). The FE plate formulation proposed is based on MITC4, see Carrera and Brischetto [48].

In this paper, CUF is exploited to implement refined models for isotropic plates and determine the role of each displacement variable in the solution. A term is considered ineffective, i.e. negligible, if its absence does not affect the accuracy of the solution with respect to a reference 3D solution. Reduced kinematic models, based on a set of retained displacement variables, are then obtained for each considered configuration. Moreover, a Best Plate Diagram (BPD) is proposed that gives guidelines to build plate theories with a given accuracy and a reduced number of displacement variables. The combined effect of loading and boundary conditions is evaluated. The paper has been organized as follows: a brief description of the adopted CUF formulation is given in Sections 2, 3, and 4; the method used to evaluate the effectiveness of various plate theories is introduced in Section 5; numerical results are provided in Section 6 and the main conclusions are outlined in Section 7.

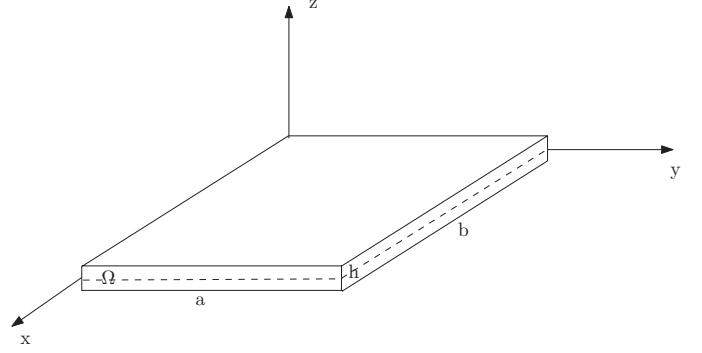


FIG. 1. Plate geometry.

## 2. PRELIMINARIES

The coordinate reference frame is shown in Fig. 1,  $x$  and  $y$  are the in-plane coordinates while  $z$  is the thickness. The displacement vector,  $\mathbf{u}$ , is:

$$\mathbf{u}(x, y, z) = \{u_x \ u_y \ u_z\}^T, \quad (1)$$

the superscript “ $T$ ” represents the transposition operator. Stress and strain components are grouped as follows:

$$\begin{aligned} \boldsymbol{\sigma}_p &= \{\sigma_{xx} \ \sigma_{yy} \ \sigma_{xy}\}^T & \boldsymbol{\epsilon}_p &= \{\epsilon_{xx} \ \epsilon_{yy} \ \epsilon_{xy}\}^T \\ \boldsymbol{\sigma}_n &= \{\sigma_{xz} \ \sigma_{yz} \ \sigma_{zz}\}^T & \boldsymbol{\epsilon}_n &= \{\epsilon_{xz} \ \epsilon_{yz} \ \epsilon_{zz}\}^T, \end{aligned} \quad (2)$$

where  $p$  indicates the in-plane components and  $n$  the out-of-plane components. Linear strain-displacement relations are used:

$$\begin{aligned} \boldsymbol{\epsilon}_p &= \mathbf{D}_p \mathbf{u} \\ \boldsymbol{\epsilon}_n &= \mathbf{D}_n \mathbf{u} = (\mathbf{D}_{n\Omega} + \mathbf{D}_{nz}) \mathbf{u}, \end{aligned} \quad (3)$$

where:

$$\begin{aligned} \mathbf{D}_p &= \begin{bmatrix} \frac{\partial}{\partial x} & 0 & 0 \\ 0 & \frac{\partial}{\partial y} & 0 \\ \frac{\partial}{\partial y} & \frac{\partial}{\partial x} & 0 \end{bmatrix} & \mathbf{D}_{n\Omega} &= \begin{bmatrix} 0 & 0 & \frac{\partial}{\partial x} \\ 0 & 0 & \frac{\partial}{\partial y} \\ 0 & 0 & 0 \end{bmatrix} \\ \mathbf{D}_{nz} &= \begin{bmatrix} \frac{\partial}{\partial z} & 0 & 0 \\ 0 & \frac{\partial}{\partial z} & 0 \\ 0 & 0 & \frac{\partial}{\partial z} \end{bmatrix}. \end{aligned} \quad (4)$$

Hooke’s law is used to compute stress components:

$$\boldsymbol{\sigma} = \mathbf{C}\boldsymbol{\epsilon}. \quad (5)$$

According to Eq. 2, the previous equation becomes:

$$\begin{aligned}\sigma_p &= C_{pp}\epsilon_p + C_{pn}\epsilon_n \\ \sigma_n &= C_{np}\epsilon_p + C_{nn}\epsilon_n,\end{aligned}\quad (6)$$

where the material matrices are grouped as follows:

$$\begin{aligned}C_{pp} &= \begin{bmatrix} C_{11} & C_{12} & 0 \\ C_{21} & C_{22} & 0 \\ 0 & 0 & C_{66} \end{bmatrix} & C_{nn} &= \begin{bmatrix} C_{55} & 0 & 0 \\ 0 & C_{44} & 0 \\ 0 & 0 & C_{33} \end{bmatrix} \\ C_{pn} &= C_{np}^T = \begin{bmatrix} 0 & 0 & C_{13} \\ 0 & 0 & C_{23} \\ 0 & 0 & 0 \end{bmatrix}.\end{aligned}\quad (7)$$

The  $C_{ij}$  components are:

$$\begin{aligned}C_{11} &= C_{22} = C_{33} = \lambda + 2\mu \\ C_{12} &= C_{23} = C_{33} = \lambda \\ C_{44} &= C_{55} = C_{66} = \mu,\end{aligned}\quad (8)$$

where

$$\begin{aligned}\mu &\equiv G = \frac{E}{2(1+\nu)} \\ \lambda &= \frac{\nu E}{(1+\nu)(1-2\nu)}.\end{aligned}\quad (9)$$

Eq. 6 is used in the numerical analysis reported in this paper. Notice that the present plate formulation is not restricted to the isotropic case; it is also applicable to anisotropic laminates.

### 3. CARRERA UNIFIED FORMULATION

In the framework of the Carrera Unified Formulation, the displacement field along the thickness direction is expressed as an expansion of generic functions,  $F_\tau$ :

$$\mathbf{u} = F_\tau \mathbf{u}_\tau \quad \tau = 0, 1, \dots, N, \quad (10)$$

where  $\mathbf{u}_\tau$  is the displacement vector,  $F_\tau$  are base functions of  $z$ , and  $N$  is the order of the expansion. Different base functions can be used; Taylor-like polynomials are used in this work:

$$F_\tau = z^\tau \quad \tau = 0, 1, \dots, N. \quad (11)$$

Any-order displacement fields can be adopted; a fourth-order model, for example, is defined by:

$$\begin{aligned}u_x &= u_{x0} + z u_{x1} + z^2 u_{x2} + z^3 u_{x3} + z^4 u_{x4} \\ u_y &= u_{y0} + z u_{y1} + z^2 u_{y2} + z^3 u_{y3} + z^4 u_{y4} \\ u_z &= u_{z0} + z u_{z1} + z^2 u_{z2} + z^3 u_{z3} + z^4 u_{z4}.\end{aligned}\quad (12)$$

Classical plate theories can be obtained from the linear expansion. The Reissner-Mindlin plate model approximation (see Reissner and Mindlin [8, 9]), also known as First Order Shear Deformation Theory, FSDT, in the case of laminates, requires two conditions: 1) First-order approximation kinematic fields; 2) the displacement component  $u_z$  has to be constant above the cross-section, i.e.  $u_{z1} = 0$ . The resultant displacement model is:

$$\begin{aligned}u_x &= u_{x0} + z u_{x1} \\ u_y &= u_{y0} + z u_{y1} \\ u_z &= u_{z0}.\end{aligned}\quad (13)$$

The Kirchhoff-type approximation (see [5]), also known as Classical Laminate Theory or CLT, can also be obtained using a penalty technique for the shear correction factor. First-order models require the use of reduced material stiffness coefficients to correct the thickness locking (see Carrera and Brischetto [49, 50]). Higher-order theories from open literature can also be obtained via CUF. Some of these models are considered in this paper for comparison purposes. They are obtained via CUF. According to Pandya (see [51]), the displacement components are given by:

$$\begin{aligned}u_x &= u_{x0} + z u_{x1} + z^2 u_{x2} + z^3 u_{x3} \\ u_y &= u_{y0} + z u_{y1} + z^2 u_{y2} + z^3 u_{y3} \\ u_z &= u_{z0}.\end{aligned}\quad (14)$$

Kant (see [52]) also expanded the displacement component  $u_z(x, y, z)$  in Taylor's series of the thickness coordinate (hereinafter referred to as Kant-1):

$$\begin{aligned}u_x &= u_{x0} + z u_{x1} + z^2 u_{x2} + z^3 u_{x3} \\ u_y &= u_{y0} + z u_{y1} + z^2 u_{y2} + z^3 u_{y3} \\ u_z &= u_{z0} + z u_{z1} + z^2 u_{z2} + z^3 u_{z3}.\end{aligned}\quad (15)$$

Another Kant model (see [53]) is considered in this work (hereinafter referred to as Kant-2):

$$\begin{aligned}u_x &= z u_{x1} + z^3 u_{x3} \\ u_y &= z u_{y1} + z^3 u_{y3} \\ u_z &= u_{z0} + z^2 u_{z2}.\end{aligned}\quad (16)$$

It is important to underline that CUF allows us to choose the higher-order terms to be included with no restrictions. For instance, one can consider a plate theory where incomplete fourth-order expansions are adopted:

$$\begin{aligned}u_x &= \quad + z u_{x1} + \quad + z^3 u_{x3} + z^4 u_{x4} \\ u_y &= \quad + z u_{y1} + z^2 u_{y2} + \quad + z^4 u_{y4} \\ u_z &= u_{z0} + z u_{z1} + z^2 u_{z2} + \quad + z^4 u_{z4}.\end{aligned}\quad (17)$$

#### 4. FE GOVERNING EQUATIONS

The shape functions,  $N_i$ , and the nodal displacement vector,  $\mathbf{q}_{ri}$ , are introduced to rewrite the displacement vector  $\mathbf{u}_\tau$ :

$$\mathbf{q}_{ri} = \{q_{u_{x_{\tau i}}} \ q_{u_{y_{\tau i}}} \ q_{u_{z_{\tau i}}}\}^T \quad (18)$$

$$\mathbf{u} = N_i F_\tau \mathbf{q}_{ri}. \quad (19)$$

For the sake of brevity, the shape functions are not reported here. The four-node plate element with an assumed shear strain field concept of MITC4 type (see [54]) is adopted in this paper. The extension of the MITC4 element to the Unified Formulation has already been discussed in Carrera et al. [55].

Upon substitution of Eq. (19) in Eq. (3) it is possible to obtain:

$$\begin{aligned} \epsilon_p &= F_\tau \mathbf{D}_p (N_i \mathbf{I}) \mathbf{q}_{ri} \\ \epsilon_n &= F_\tau \mathbf{D}_{n\Omega} (N_i \mathbf{I}) \mathbf{q}_{ri} + F_{\tau,z} N_i \mathbf{q}_{ri}, \end{aligned} \quad (20)$$

where  $\mathbf{I}$  stands for the identity matrix. The stiffness matrix of the elements and the external loadings, which are consistent with the model, are obtained via the Principle of Virtual Displacements:

$$\delta L_{int} = \int_V (\delta \epsilon_p^T \sigma_p + \delta \epsilon_n^T \sigma_n) dV = \delta L_{ext}, \quad (21)$$

where  $L_{int}$  stands for the strain energy, and  $L_{ext}$  is the work of the external loadings.  $\delta$  stands for the virtual variation. The virtual variation of the strain energy is rewritten using Eqs. (3), (6), and (19), so in a compact format it becomes:

$$\delta L_{int} = \delta \mathbf{q}_{ri}^T \mathbf{K}^{ij\tau s} \mathbf{q}_{sj}, \quad (22)$$

where  $\mathbf{K}^{ij\tau s}$  is the stiffness matrix in the form of the fundamental nucleus. The following notation is introduced to indicate the line integrals along the thickness direction:

$$\begin{aligned} (\mathbf{Z}_{pp}^{\tau s}, \mathbf{Z}_{pn}^{\tau s}, \mathbf{Z}_{np}^{\tau s}, \mathbf{Z}_{nn}^{\tau s}) &= (\mathbf{C}_{pp}, \mathbf{C}_{pn}, \mathbf{C}_{np}, \mathbf{C}_{nn}) E_{\tau s} \\ (\mathbf{Z}_{pn}^{\tau s,z}, \mathbf{Z}_{nn}^{\tau s,z}, \mathbf{Z}_{np}^{\tau,z s}, \mathbf{Z}_{nn}^{\tau,z s}, \mathbf{Z}_{nn}^{\tau,z,z}) &= (\mathbf{C}_{pn} E_{\tau s,z}, \mathbf{C}_{nn} E_{\tau s,z}, \mathbf{C}_{np} E_{\tau,z s}, \\ &\quad \mathbf{C}_{nn} E_{\tau,z s}, \mathbf{C}_{nn} E_{\tau,z,z}) \\ (E_{\tau s}, E_{\tau s,z}, E_{\tau,z s}, E_{\tau,z,z}) &= \int_A (F_\tau F_s, F_\tau F_{s,z}, \\ &\quad F_{\tau,z} F_s, F_{\tau,z} F_{s,z}) dz. \end{aligned} \quad (23)$$

The compact expression of the stiffness matrix is then outlined:

$$\begin{aligned} \mathbf{K}^{\tau s ij} &= \triangleleft \mathbf{D}_p^T (N_i \mathbf{I}) [\mathbf{Z}_{pp}^{\tau s} \mathbf{D}_p (N_j \mathbf{I}) + \mathbf{Z}_{pn}^{\tau s} \mathbf{D}_{n\Omega} (N_j \mathbf{I}) + \mathbf{Z}_{pn}^{\tau s,z} N_j] \\ &\quad + \mathbf{D}_{n\Omega}^T (N_i \mathbf{I}) [\mathbf{Z}_{np}^{\tau s} \mathbf{D}_p (N_j \mathbf{I}) + \mathbf{Z}_{nn}^{\tau s} \mathbf{D}_{n\Omega} (N_j \mathbf{I}) \\ &\quad + \mathbf{Z}_{nn}^{\tau,z} N_j] + N_i [\mathbf{Z}_{np}^{\tau,z s} \mathbf{D}_p (N_j \mathbf{I}) + \mathbf{Z}_{nn}^{\tau,z s} \mathbf{D}_{n\Omega} (N_j \mathbf{I}) \\ &\quad + \mathbf{Z}_{nn}^{\tau,z,z} N_j] \triangleright_\Omega. \end{aligned} \quad (24)$$

The symbols  $\triangleleft \dots \triangleright_\Omega$  were introduced to denote integrals on  $\Omega$ . The matrix  $\mathbf{K}^{\tau s ij}$  has  $3 \times 3$  components and the 9 terms of

$\mathbf{K}^{\tau s ij}$  are:

$$\begin{aligned} K_{xx}^{\tau s ij} &= Z_{pp11}^{\tau s} \triangleleft N_{i,x} N_{j,x} \triangleright_\Omega + Z_{pp66}^{\tau s} \triangleleft N_{i,y} N_{j,y} \triangleright_\Omega \\ &\quad + Z_{nn55}^{\tau,z} \triangleleft N_i N_j \triangleright_\Omega \\ K_{xy}^{\tau s ij} &= Z_{pp12}^{\tau s} \triangleleft N_{i,x} N_{j,y} \triangleright_\Omega + Z_{pp66}^{\tau s} \triangleleft N_{i,y} N_{j,x} \triangleright_\Omega \\ K_{xz}^{\tau s ij} &= Z_{pn13}^{\tau s} \triangleleft N_{i,x} N_j \triangleright_\Omega + Z_{nn55}^{\tau s} \triangleleft N_i N_{j,x} \triangleright_\Omega \\ K_{yx}^{\tau s ij} &= Z_{pp12}^{\tau s} \triangleleft N_{i,y} N_{j,x} \triangleright_\Omega + Z_{pp66}^{\tau s} \triangleleft N_{i,x} N_{j,y} \triangleright_\Omega \\ K_{yz}^{\tau s ij} &= Z_{pp22}^{\tau s} \triangleleft N_{i,y} N_{j,y} \triangleright_\Omega + Z_{pp66}^{\tau s} \triangleleft N_{i,x} N_{j,x} \triangleright_\Omega \\ &\quad + Z_{nn44}^{\tau,z} \triangleleft N_i N_j \triangleright_\Omega \\ K_{yz}^{\tau s ij} &= Z_{pn23}^{\tau,z} \triangleleft N_{i,y} N_j \triangleright_\Omega + Z_{nn44}^{\tau,z} \triangleleft N_i N_{j,y} \triangleright_\Omega \\ K_{zx}^{\tau s ij} &= Z_{nn55}^{\tau s} \triangleleft N_{i,x} N_j \triangleright_\Omega + Z_{np13}^{\tau,z} \triangleleft N_i N_{j,x} \triangleright_\Omega \\ K_{zy}^{\tau s ij} &= Z_{nn45}^{\tau,z} \triangleleft N_{i,x} N_j \triangleright_\Omega + Z_{nn44}^{\tau,z} \triangleleft N_{i,y} N_j \triangleright_\Omega \\ &\quad + Z_{np23}^{\tau,z} \triangleleft N_i N_{j,y} \triangleright_\Omega \\ K_{zz}^{\tau s ij} &= Z_{nn55}^{\tau s} \triangleleft N_{i,x} N_{j,x} \triangleright_\Omega + Z_{nn44}^{\tau s} \triangleleft N_{i,y} N_{j,y} \triangleright_\Omega \\ &\quad + Z_{nn33}^{\tau,z} \triangleleft N_i N_j \triangleright_\Omega. \end{aligned} \quad (25)$$

It should be noted that no assumptions on the approximation order were made. It is therefore possible to obtain refined plate models without changing the formal expression of the fundamental nucleus.

The loading vector, which is variationally consistent with the model, is derived in the case of a generic concentrate load  $\mathbf{P}$ :

$$\mathbf{P} = \{P_{u_x} \ P_{u_y} \ P_{u_z}\}. \quad (26)$$

Any other loading condition can be similarly treated as usual in FE applications. The virtual work due to  $\mathbf{P}$  is:

$$\delta L_{ext} = \mathbf{P} \delta \mathbf{u}^T. \quad (27)$$

Substituting Eq. (19), the previous equation becomes:

$$\delta L_{ext} = F_\tau N_i \mathbf{P} \delta \mathbf{q}_{ri}^T. \quad (28)$$

This last equation permits the identification of the components of the nucleus which have to be loaded. In the case of a first-order expansion and  $\mathbf{P}$  applied to a node and acting along the  $x$  direction, the virtual external work is:

$$\delta L_{ext} = P_{u_x} \delta u_{x1} + z_p P_{u_x} \delta u_{x2}, \quad (29)$$

where  $z_p$  is the thickness coordinate of the loading application point.

#### 5. AN ASYMPTOTIC/AXIOMATIC APPROACH TO EVALUATE VARIOUS PLATE MODELS

Refined plate theories offer significant advantages in terms of accuracy of the solution and detection of non-classical effects. The drawback of these theories is that a higher computational cost has to be incurred because of the presence of a large number

**TABLE 1**

Locations of the displacement variables within the table layout

$N = 0$	$N = 1$	$N = 2$	$N = 3$	$N = 4$
$u_{x_0}$	$u_{x_1}$	$u_{x_2}$	$u_{x_3}$	$u_{x_4}$
$u_{y_0}$	$u_{y_1}$	$u_{y_2}$	$u_{y_3}$	$u_{y_4}$
$u_{z_0}$	$u_{z_1}$	$u_{z_2}$	$u_{z_3}$	$u_{z_4}$

of displacement variables. The aim of this work is to better understand the influence of each term on the solution in order to evaluate which variables are effective and which are not. The effectiveness of each term is investigated as follows:

1. The problem data is fixed (i.e. geometry, boundary conditions, loadings, materials).
2. A set of output variables is chosen (e.g. maximum displacement, stress/displacement component at a given point).
3. A theory is fixed; that is, the terms that have to be considered in the expansion of  $u_x$ ,  $u_y$  and  $u_z$  are established.
4. A reference solution is used to establish the accuracy (the  $N = 4$  case is assumed as the best-reference result).
5. CUF is used to generate the finite element solution for the theories considered.
6. The effectiveness of each term is numerically established by measuring the error produced with respect to the reference solution.
7. Any term that does not give any contribution to the computation of the mechanical response is not considered as effective in the plate model.
8. The most suitable plate model is then detected for a given structural layout; in other words, the plate model requiring the lowest number of terms to accomplish a give accuracy is built.

A graphic notation is introduced to make the representation of the obtained results more readable. Table 1 shows all the 15 terms of the expansion. The fourth-order model,  $N = 4$ , is related to the following expression:

$$\begin{aligned}
 u_x &= u_{x_0} + zu_{x_1} + z^2u_{x_2} + z^3u_{x_3} + z^4u_{x_4} \\
 u_y &= u_{y_0} + zu_{y_1} + z^2u_{y_2} + z^3u_{y_3} + z^4u_{y_4} \\
 u_z &= u_{z_0} + zu_{z_1} + z^2u_{z_2} + z^3u_{z_3} + z^4u_{z_4}.
 \end{aligned} \quad (30)$$

The first three terms are constant (the first column), the second three terms are linear (the second column), the third three terms are quadratic (the third column), the fourth three terms are cubic

**TABLE 2**Symbolic representation of the kinematic model with  $u_{y_2}$  deactivated

●	●	●	●	●
●	●	○	●	●
●	●	●	●	●

**TABLE 3**

Symbolic representation of the Kant [52] kinematic model

●	●	●	●	○
●	●	●	●	○
●	●	●	●	○

(the fourth column) and the last three terms are fourth-order displacement variables (the fifth column). The total number of expansion terms is 15. A set of symbols is used to denote the active and inactive terms. Each adopted symbol is related to a given structural case. The black color indicates that the term is active; that is, the displacement variable is exploited to compute the results while the white color indicates that the term is inactive; that is, the displacement variable is removed from the displacement model. Table 2 shows the case where the parabolic term of the in-plane displacement of the expansion in the  $y$ -direction is discarded. The explicit displacement model related to Table 2 is:

$$\begin{aligned}
 u_x &= u_{x_0} + zu_{x_1} + z^2u_{x_2} + z^3u_{x_3} + z^4u_{x_4} \\
 u_y &= u_{y_0} + zu_{y_1} + \quad + z^3u_{y_3} + z^4u_{y_4} \\
 u_z &= u_{z_0} + zu_{z_1} + z^2u_{z_2} + z^3u_{z_3} + z^4u_{z_4}.
 \end{aligned} \quad (31)$$

As a further example, Table 3 reports the schematic representation of the Kant-1 (15) plate model. The elimination of a term, as well as the evaluation of its effectiveness in the analysis, can be obtained either by rearranging the rows and columns of the stiffness matrix or by exploiting a penalty technique. The accuracy of a reduced model is evaluated by computing  $\delta_u$  and  $\delta_\sigma$ . These quantities are defined according to the following formulas:

$$\begin{aligned}
 \delta_{u_z} &= \frac{u_z}{u_{z_{ref}}} \times 100\% \\
 \delta_{\sigma_{zz}} &= \frac{\sigma_{zz}}{\sigma_{zz_{ref}}} \times 100\%,
 \end{aligned} \quad (32)$$

where  $u_{z_{ref}}$  and  $\sigma_{zz_{ref}}$  denote the transversal displacement and stress values, which are taken as references. A displacement

**TABLE 4**

Symbols that indicate the status of a displacement variable for different boundary conditions: clamped, 'c', simply-supported, 's', free, 'f'

Active term	Inactive term	
▲	△	ssss
■	□	cfcf
▼	▽	cccc
◆	◇	scsc

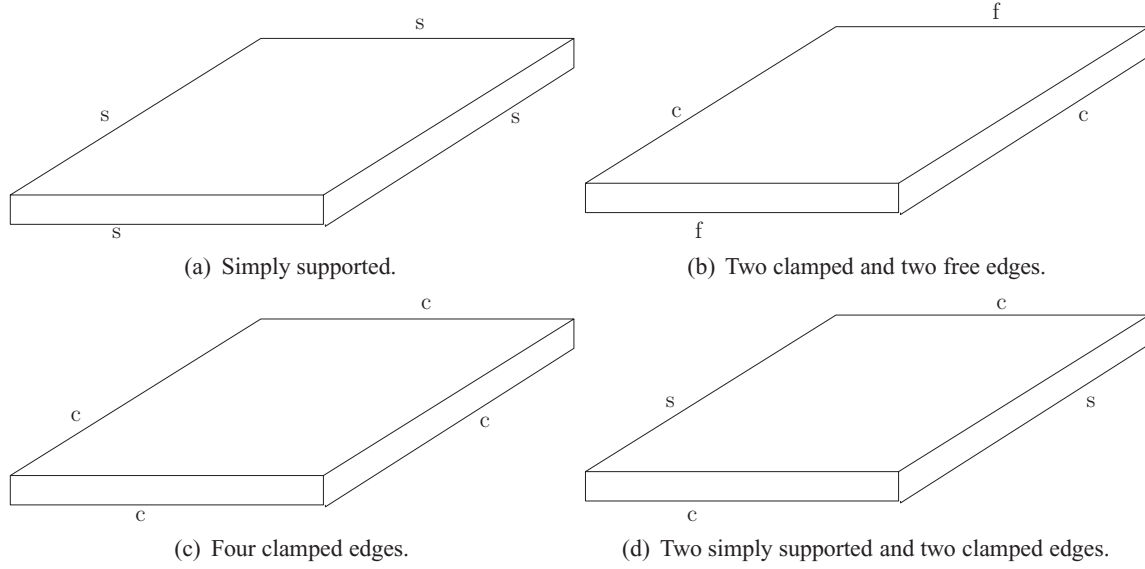


FIG. 2. Adopted boundary conditions.

variable of the expansion is considered to be non-effective with respect to a specific output component when, if neglected (removed from the formulation), it does not introduce any changes in the results according to a fixed accuracy ( $\delta \geq 99.95\%$ ). This value was chosen as it was observed that, if the reduced model that does not include the terms causing a loss in accuracy lower than  $0.05\%$  is considered, the loss in accuracy of the reduced model will be lower than  $0.05\%$ . A companion parameter that will be used is related to the error calculated. If  $u_z$  is considered, the error will be defined as:

$$error = \left\| \frac{u_z - u_{z_{ref}}}{u_{z_{ref}}} \right\| \times 100\%. \quad (33)$$

The approach described allows us to build plate models that are equivalent to a full higher-order model ( $N = 4$  in this paper) having a lower number of displacement variables. A further analysis is then conducted on the reduced models by decreasing the fixed accuracy. This allows us to build a so-called Best Plate Diagram consisting of a “number of term vs. error,” curve which

can be useful to determine the effectiveness of a given theory in terms of accuracy and computational cost.

The asymptotic/axiomatic approach is herein applied by making different parameters vary. The influence of the following plate characteristics in determining a plate model is evaluated:

1. Boundary conditions.
2. Loadings.
3. The length-to-thickness ratio ( $a/h$ ).

## 6. RESULTS AND DISCUSSION

An isotropic plate is considered. The Young modulus,  $E$ , is equal to 73 [GPa] and the Poisson ratio,  $\nu$ , is equal to 0.34. The geometry of the plate is shown in Fig. 1, where  $a$  is 0.1 [m], and  $b$  is equal to  $a$ .  $u_z$ ,  $\sigma_{xx}$ ,  $\sigma_{yy}$ , and  $\sigma_{zz}$  are computed at  $[a/2, b/2, 0]$ , while  $\sigma_{xz}$  is computed at  $[0, b/2, h/2]$  and  $\sigma_{yz}$  is computed at  $[a/2, 0, h/2]$ . Stresses are computed through Hooke's laws. Four-node plate elements have been used, and a uniform mesh of  $15 \times 15$  elements has been adopted after a convergence study.

TABLE 5

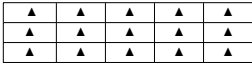
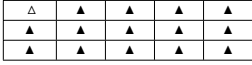
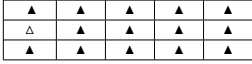
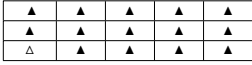
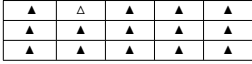
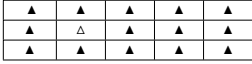
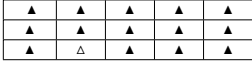
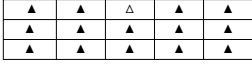
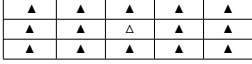
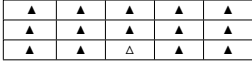
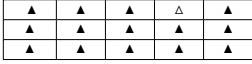
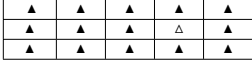
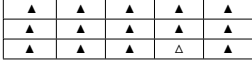
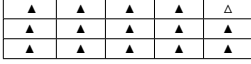
Comparison of the the 3D model and the fourth-order FEM model solutions in the case of simply supported plate with a distributed load:  $\bar{u}_z = u_z \frac{100ETh^3}{p_za^4}$ ,  $\bar{\sigma}_{xx} = \frac{\sigma_{xx}}{p_z(a/h)}$ ,  $\bar{\sigma}_{xz} = \frac{\sigma_{xz}}{p_z(a/h)^2}$ ,  $\bar{\sigma}_{zz} = \frac{\sigma_{zz}}{p_z(a/h)}$

	$\bar{u}_{zN=4FEM}$	$\bar{u}_{z3D}$	$\bar{\sigma}_{xxN=4FEM}$	$\bar{\sigma}_{xx3D}$	$\bar{\sigma}_{xzN=4FEM}$	$\bar{\sigma}_{xz3D}$	$\bar{\sigma}_{zzN=4FEM}$	$\bar{\sigma}_{zz3D}$
$a/h = 100$	2.7134	2.7248	0.2040	0.2037	0.2390	0.2387	0.0866	0.0100
$a/h = 10$	2.8255	2.8345	0.2070	0.2068	0.2386	0.2383	0.1041	0.1000
$a/h = 5$	3.1978	3.2056	0.2172	0.2168	0.2378	0.2371	0.2024	0.2002
$a/h = 2$	7.3840	7.3826	0.3171	0.3145	0.2308	0.2277	0.5099	0.5000

## GUIDELINES FOR BENDING ANALYSIS OF PLATES

**TABLE 6**

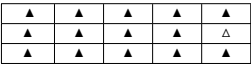
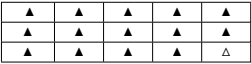
Influence of each displacement variable of a fourth-order model on the solution, simply supported plate with a distributed load,  
 $a/h = 10$

	$\delta_{u_z}$ [%]	$\delta_{\sigma_{xx}}$ [%]	$\delta_{\sigma_{xz}}$ [%]	$\delta_{\sigma_{zz}}$ [%]
	100.0	100.0	100.0	100.0
	100.0	100.5	100.0	100.1
	100.0	99.9	100.0	100.1
	0.13	1.18	1.27	75.59
	19.5	14.7	269.3	93.9
	19.5	18.8	16.11	93.90
	100.0	101.4	100.0	136.3
	100.0	100.0	100.0	100.1
	100.0	100.0	100.0	100.1
	95.4	76.6	101.6	-598.1
	99.7	99.7	72.2	100.0
	99.7	99.7	99.9	100.0
	100.0	100.0	100.0	100.5
	100.0	100.0	100.0	100.0

(Continued on next page)

TABLE 6

Influence of each displacement variable of a fourth-order model on the solution, simply supported plate with a distributed load,  $a/h = 10$  (Continued)

	$\delta_{u_z}$ [%]	$\delta_{\sigma_{xx}}$ [%]	$\delta_{\sigma_{xz}}$ [%]	$\delta_{\sigma_{zz}}$ [%]
	100.0	100.0	100.0	100.0
	100.0	100.7	100.0	125.7

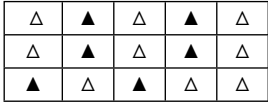
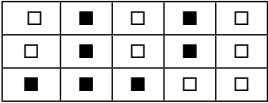
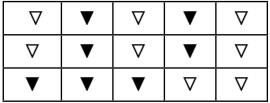
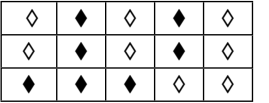
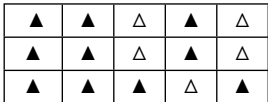
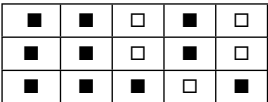
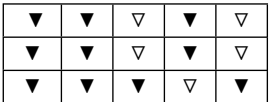
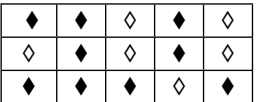
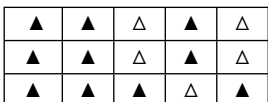
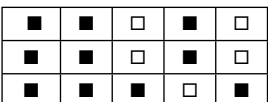
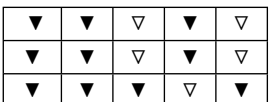
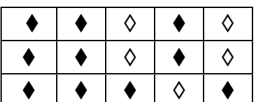
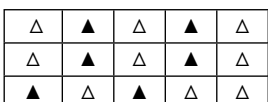
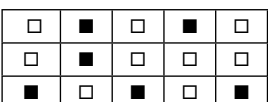
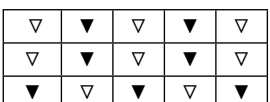
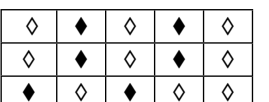
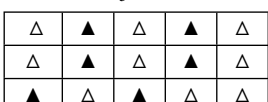
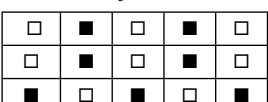
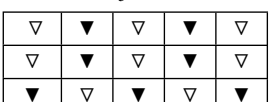
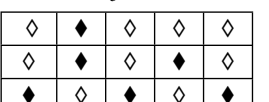
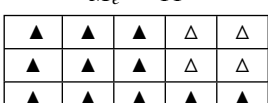
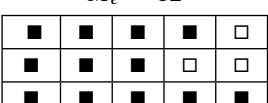
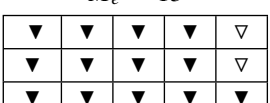

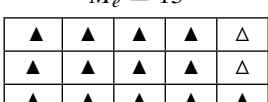
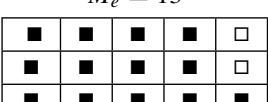
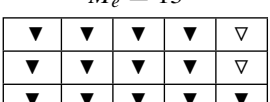
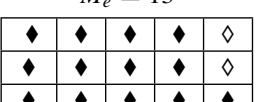
### 6.1. Boundary Condition Effects

The investigation of the role of each displacement variable under different boundary conditions is conducted as a first assessment. Four different boundary condition sets are considered:

1. *ssss*, four simply-supported edges;
2. *cfcf*, two clamped and two free edges;
3. *cccc*, four clamped edges;
4. *scsc*, two simply-supported and two clamped edges.

TABLE 7

Comparison of the sets of effective terms for a plate with a distributed load with different boundary conditions,  $a/h = 10$

	<i>ssss</i>	<i>cfcf</i>	<i>cccc</i>	<i>scsc</i>
$u_z$	$M_e = 6$ 	$M_e = 7$ 	$M_e = 7$ 	$M_e = 7$ 
$\sigma_{xx}$	$M_e = 10$ 	$M_e = 10$ 	$M_e = 10$ 	$M_e = 9$ 
$\sigma_{yy}$	$M_e = 10$ 	$M_e = 10$ 	$M_e = 10$ 	$M_e = 10$ 
$\sigma_{xz}$	$M_e = 6$ 	$M_e = 6$ 	$M_e = 7$ 	$M_e = 6$ 
$\sigma_{yz}$	$M_e = 6$ 	$M_e = 7$ 	$M_e = 7$ 	$M_e = 6$ 
$\sigma_{zz}$	$M_e = 11$ 	$M_e = 12$ 	$M_e = 13$ 	$M_e = 12$ 
COMBINED	$M_e = 13$ 	$M_e = 13$ 	$M_e = 13$ 	$M_e = 13$ 

**TABLE 8**  
Accuracy of different models for a plate with a distributed load,  $a/h = 10$

					$\delta_{u_z}$ [%]	$\delta_{\sigma_{xx}}$ [%]	$\delta_{\sigma_{yy}}$ [%]	$\delta_{\sigma_{xz}}$ [%]	$\delta_{\sigma_{yz}}$ [%]	$\delta_{\sigma_{zz}}$ [%]
SSSS										
▲	▲	▲	▲	△	100.0	100.0	100.0	100.0	100.0	100.0
▲	▲	▲	▲	△						
▲	▲	▲	▲	▲						
CPT					96.1	98.4	98.4	66.8	66.8	1974.2
FSDT					100.9	98.4	98.4	66.8	66.8	1974.2
Kant-2					99.9	100.2	100.2	100.0	100.0	79.4
cfcf										
■	■	■	■	□	100.0	100.0	100.0	100.0	100.0	100.0
■	■	■	■	□						
■	■	■	■	■						
CPT					91.2	98.3	94.9	239.5	-67.86	1171.3
FSDT					103.1	98.7	97.2	83.2	65.4	1183.7
Kant-2					99.9	100.0	101.0	98.8	99.7	80.4
scsc										
▼	▼	▼	▼	▽	100.0	100.0	100.0	100.0	100.0	100.0
▼	▼	▼	▼	▽						
▼	▼	▼	▼	▼						
CPT					88.4	96.2	96.2	87.5	87.5	1074.0
FSDT					102.8	97.1	97.1	81.4	81.4	1084.8
Kant-2					99.8	99.9	99.9	98.9	98.9	80.0
scsc										
◆	◆	◆	◆	◇	100.0	100.0	100.0	100.0	100.0	100.0
◆	◆	◆	◆	◇						
◆	◆	◆	◆	◆						
CPT					90.2	94.6	97.7	66.1	84.7	1281.1
FSDT					102.4	98.3	97.9	67.3	81.8	1304.3
Kant-2					99.8	100.6	100.0	100.0	98.9	79.8

Fig. 2 shows each set and Table 4 presents the relative symbolic representation, which is exploited to report the results in a compact manner. A bi-sinusoidal transverse distributed load is applied at the top surface. Its expression is:

$$P_z = p_z \sin\left(\frac{mx}{a}\right) \cos\left(\frac{ny}{b}\right), \quad (34)$$

where  $p_z$  is the amplitude and is equal to 1 [kPa];  $m$  and  $n$  are the wave numbers in the two in-plane plate directions, both of them

are equal to 1. A 3D solution is used to identify which order of the expansion has to be used to obtain 3D-like results; that is, which higher-order plate model is needed to detect a 3D exact solution. Table 5 shows the comparison of the 3D solution and the fourth-order FEM one. The fourth-order model solution,  $N = 4$ , offers an excellent match with the exact solution for

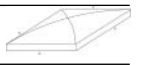
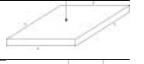
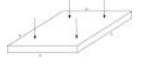
**TABLE 9**

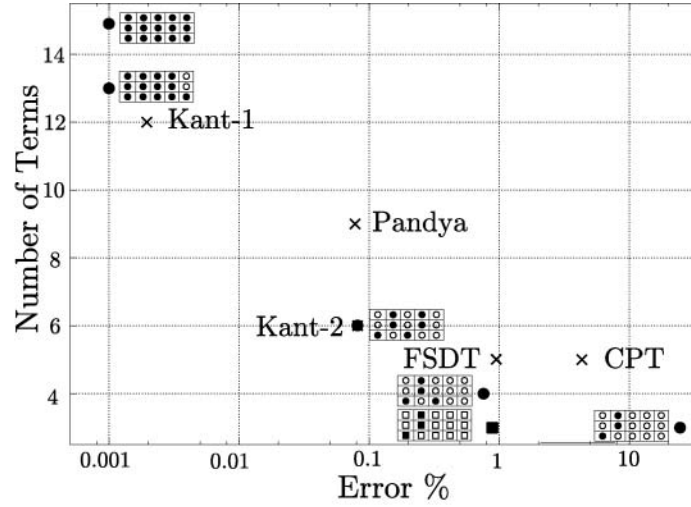
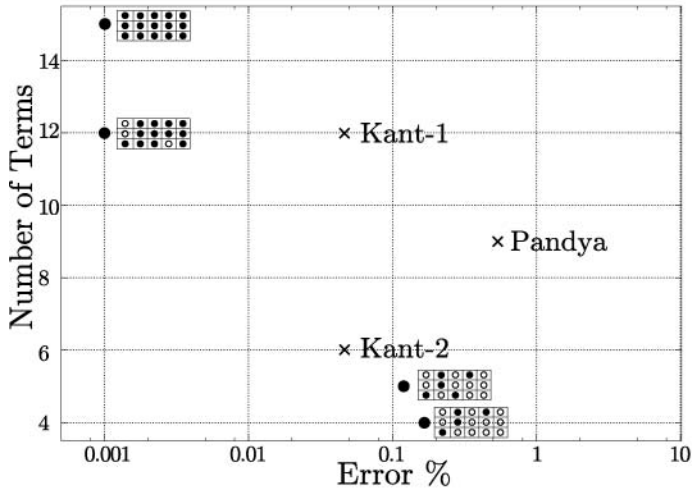
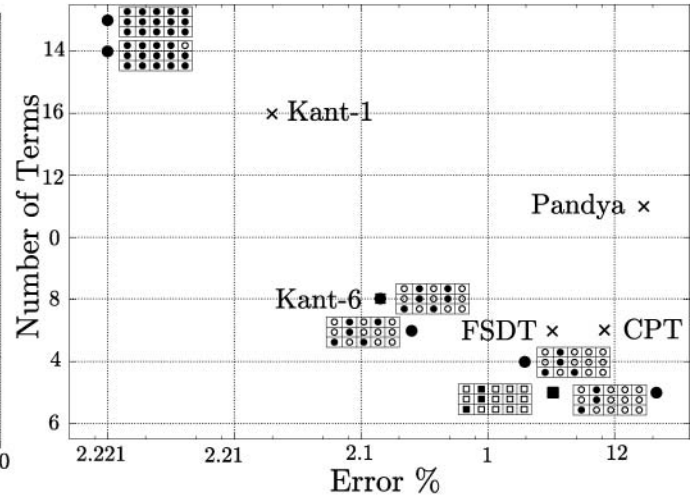
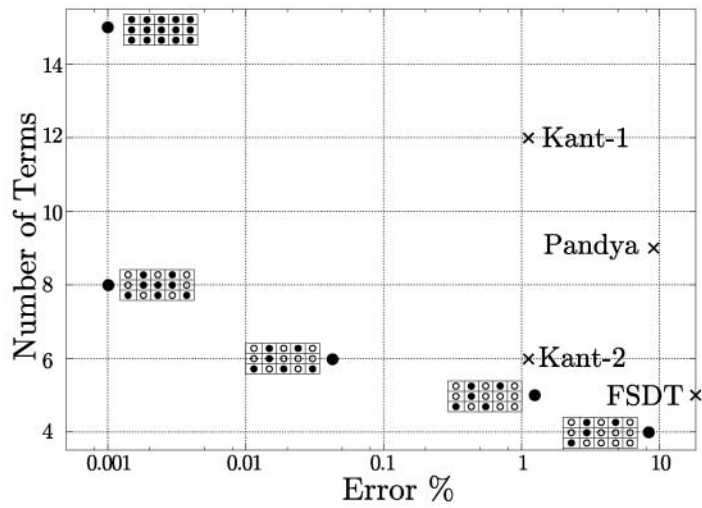
Sets of effective terms for all the considered boundary conditions

▲▼◆	▲▼◆	▲▼◆	▲▼◆	△ □ ▽ ◇
▲▼◆	▲▼◆	▲▼◆	▲▼◆	△ □ ▽ ◇
▲▼◆	▲▼◆	▲▼◆	▲▼◆	▲▼◆

**TABLE 10**

Symbols that indicate the status of a displacement variable for different loading conditions

Active term	Inactive term	
▲	△	<i>Distributed Load</i> 
■	□	<i>Point Load</i> 
▼	▽	<i>Four Point Loads</i> 

(a) Simply supported plate. Output  $u_z$ .(b) Simply supported plate. Output  $\sigma_{xz}$ .(c) Two clamped e two free edges plate. Output  $u_z$ .(d) Two clamped e two free edges plate. Output  $\sigma_{xz}$ .

**FIG. 3.** Number of terms vs. error for different output variables by various models for a simply supported plate and two edges free and two edges clamped plate with a distributed load,  $a/h = 10$ . ■ model with the thickness locking corrected.

TABLE 11

Fourth-order FEM model solutions in the case of simply supported plate subject to point loading conditions,  $a/h = 10$ 

	$u_{z,N=4}$ [m]	$\sigma_{xx,N=4}$ [Pa]	$\sigma_{yz,N=4}$ [Pa]	$\sigma_{zz,N=4}$ [Pa]
Point Load	$1.7662 \times 10^{-7}$	$1.9336 \times 10^5$	$7.0444 \times 10^3$	$3.1373 \times 10^4$
Four Point Loads	$1.7662 \times 10^{-7}$	$1.9336 \times 10^5$	$6.2392 \times 10^3$	$3.1373 \times 10^4$

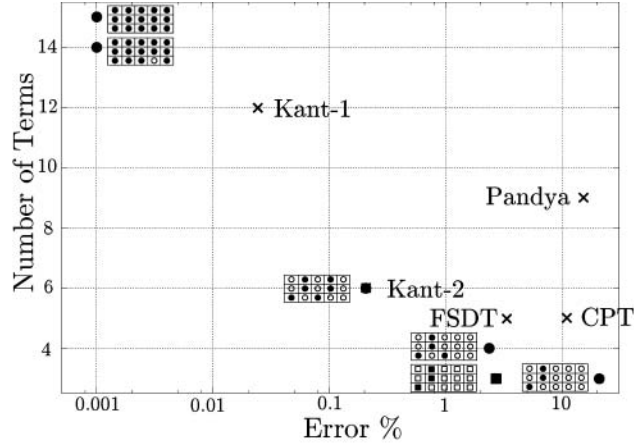
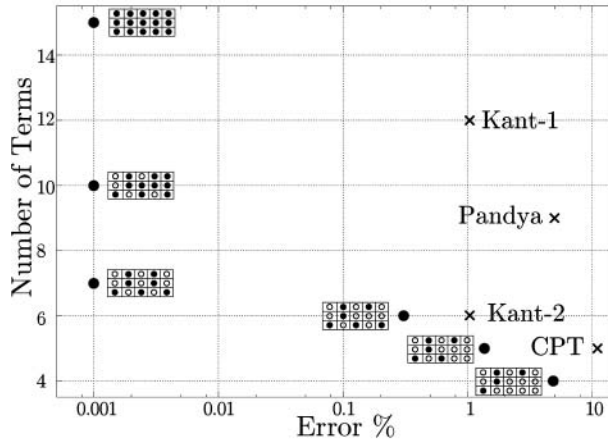
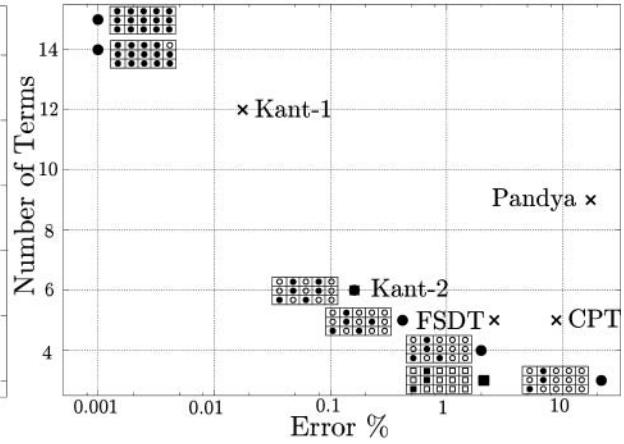
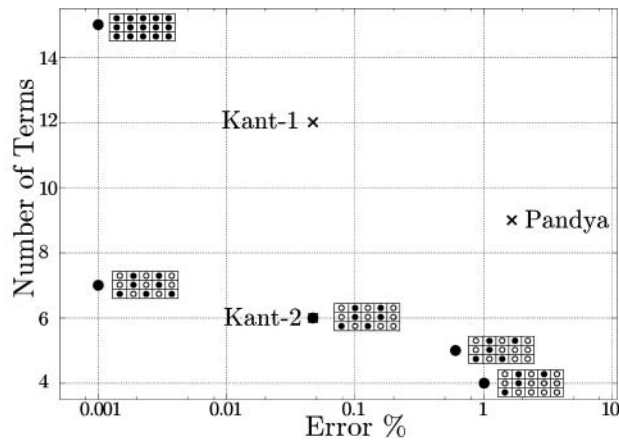
(a) Clamped Plate. Output  $u_z$ .(b) Clamped Plate. Output  $\sigma_{xz}$ .(c) Two simply supported and two clamped edges plate. Output  $u_z$ .(d) Two simply supported and two clamped edges plate. Output  $\sigma_{xz}$ .

FIG. 4. Number of terms vs. error for different output variables by various models for a clamped plate and two simply-supported edges and two clamped edges plate with a distributed load,  $a/h = 10$ . ■ model with the thickness locking corrected.

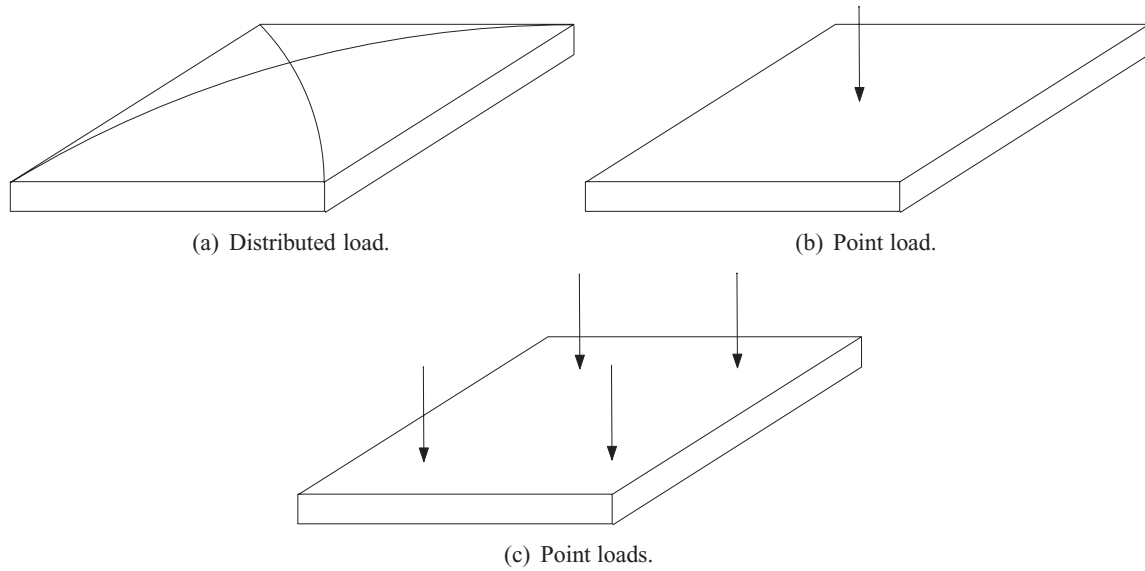


FIG. 5. Adopted loading conditions.

about all the length-to-thickness ratios and considered outputs, therefore it has been chosen as the reference for the following analyses. The effectiveness of each term is investigated for each set of boundary conditions. Table 6 shows the analysis of a moderately thick simply-supported plate. The first column shows the plate models considered. The remaining columns report model accuracies in determining a given output variable compared to a full fourth-order model. Table 7 has been obtained by exploiting Table 6: the sets of displacement variables which are needed to exactly detect various output variables are given. Each row refers to an output variable. Each column considers a different set of boundary conditions.  $M_e$  indicates the number of terms (i.e. the computational cost) of the models that are equivalent to the fourth-order one. The last row shows the expansion terms needed to detect all the considered outputs. The latter combined models are used to build Table 8, which shows a comparison with the accuracies given by CPT, FSDT, and Kant-2 plate models. Table 9 summarizes the combined displacement fields that are needed to detect the exact solution for all the considered boundary conditions. The results that have been shown so far give us the reduced plate models, which detect a 3D-like solution. Starting from these reduced models, it is possible to determine further theories that provide solutions with a certain degree of error compared to the full fourth-order one. Fig. 3 and Fig. 4, show the number of terms needed to detect an output variable with a given error. The corresponding plate models are also indicated. For the linear models, the results are reported with and without the correction of the thickness locking. Models from the open literature have also been included. The accepted error range varies from 0% to 10%. The analyses undertaken suggest what follows:

1. The reduced plate models, which are equivalent to a fourth-order theory, vary significantly if different output variables are considered.
2. The influence of boundary conditions is not high.
3. Classical models are inadequate to deal with shear stresses.
4. A significant computational cost reduction is obtained only if a limited number of output variables has to be detected.
5. The number of terms vs. error diagram shows that all the theories derived from the present approach are able to satisfy a given error requirement with a lower computational cost than the open literature models considered. This shows the strength of the present technique for detecting the possible best theories for a given structural problem.

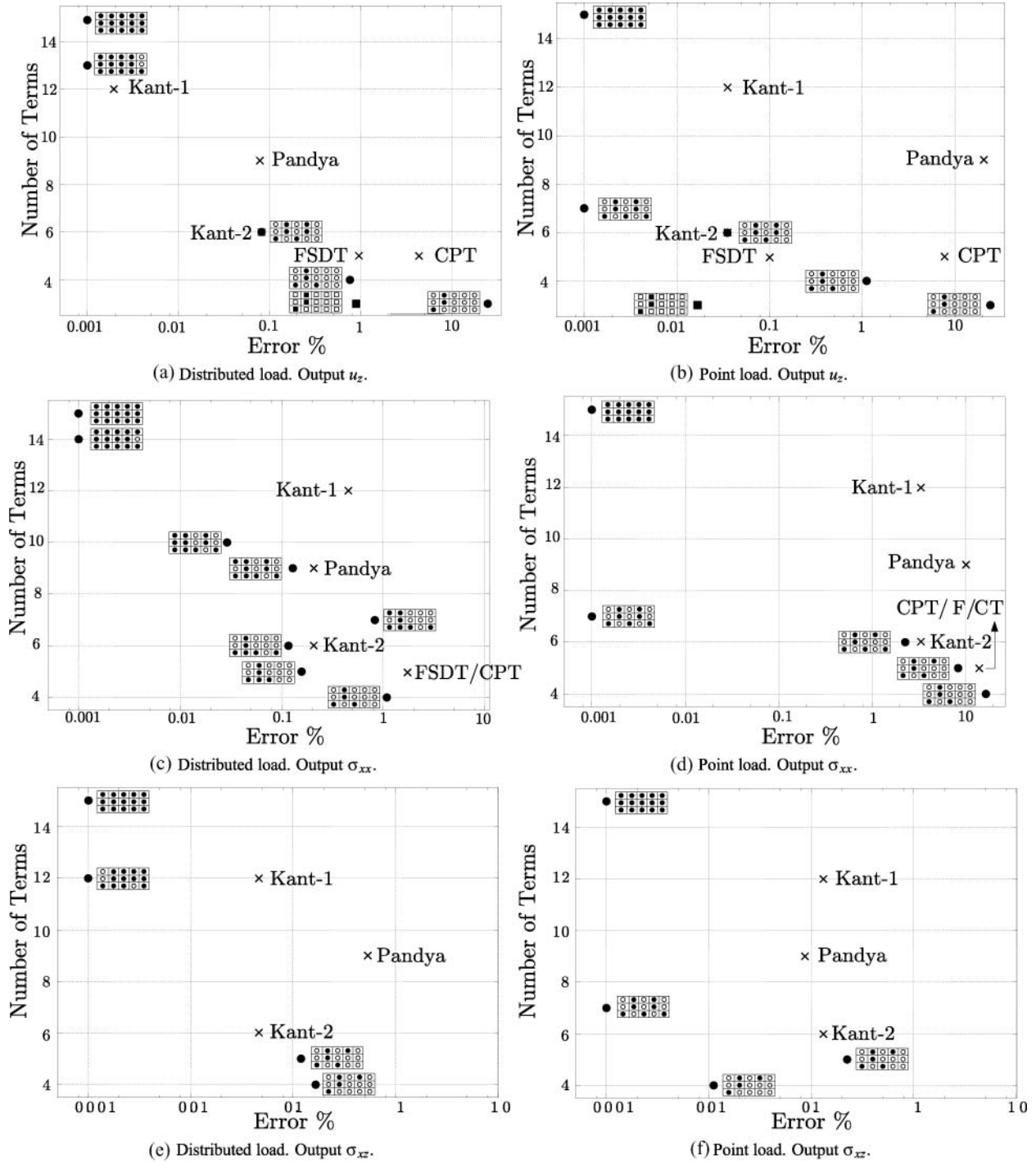
## 6.2. Loading Effects

The effect of the loading conditions is herein investigated. A simply supported plate is considered. Three different loading conditions are taken into account:

1. A bi-sinusoidal distributed load (Fig. 5 (a));
2. A point load (Fig. 5 (b));
3. Four point loads (Fig. 5 (c)).

The point load, which is applied at the top surface of the plate, has an intensity of 10 N while the four point loads have an intensity of 2.5 N each. Table 10 reports the symbolic representation of each loading set. Table 11 shows the fourth-order FEM solution for the point load condition. Table 5 values are used for

## GUIDELINES FOR BENDING ANALYSIS OF PLATES






**FIG. 6.** Number of terms vs. error for different output variables and loading conditions by various models for a simply-supported plate,  $a/h = 10$ . ■ model with the thickness locking corrected.

the distributed loading case. The sets of displacement variables, which are needed to detect various outputs precisely, are given in Table 12. Each row refers to a different output variable. Each column considers a different loading condition.  $M_e$  indicates the number of terms of the models that are equivalent to the fourth-order one. The last row shows the plate models that are

needed to detect all the considered outputs precisely. The latter combined models are used to build Table 13, which shows a comparison of the accuracies given by CPT, FSDT, and Kant-2 plate models. Table 14 summarizes the combined displacement fields, able to detect the exact solution for all the considered loading conditions. Fig. 6 shows the number of terms needed

**TABLE 12**  
Comparison of the sets of effective terms for a simply supported plate subject to different loading conditions

																																																
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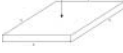
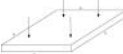
to compute an output variable with a given error. The corresponding plate models are also indicated together with models retrieved from open literature. For the linear models, the results are reported with and without the thickness locking correction. The accepted error range varies from 0% to 10%. No differences between the one point load case and the four point load one have been observed; therefore the graphs related to the four point load are not reported. The following remarks arise from the analyses carried out:

1. The sets of effective displacement variables vary if distributed or concentrated loads are considered, whereas

there are no differences if one or multiple point loads are used.

2. The effect of the considered output variable is significant in the case of distributed load, it is almost negligible if point loads are considered.
3. The validity of the number of terms vs. error diagram for providing guidelines for the construction of plate theories accomplishing a given accuracy is confirmed. As the error is fixed, the theories derived generally lie beneath classical and other refined models; that is, the proposed models are able to fulfill a certain accuracy demand with a lower computational cost.

**TABLE 13**  
Accuracy of different plate models for a simply supported plate subject to various loading conditions,  $a/h = 10$

					$\delta_{u_z}$ [%]	$\delta_{\sigma_{xx}}$ [%]	$\delta_{\sigma_{yy}}$ [%]	$\delta_{\sigma_{xz}}$ [%]	$\delta_{\sigma_{xz}}$ [%]	$\delta_{\sigma_{zz}}$ [%]
										
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CPT					96.1	98.4	98.4	66.8	66.8	1974.2
FSDT					100.9	98.4	98.4	66.8	66.8	1974.2
Kant-2					99.9	100.3	100.3	100.0	100.0	79.4
										
□	■	□	■	□	100.0	100.0	100.0	100.0	100.0	100.0
□	■	□	■	□						
■	□	■	□	■						
CPT					91.4	83.7	83.7	67.0	67.0	540.1
FSDT					99.9	83.7	83.7	67.0	67.0	540.1
Kant-2					100.0	96.5	96.5	99.9	99.9	39.5
										
▽	▼	▽	▼	▽	100.0	100.0	100.0	100.0	100.0	100.0
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▼	▽	▼	▽	▼						
CPT					91.4	83.7	83.7	67.0	67.0	540.1
FSDT					99.9	83.7	83.7	67.0	67.0	540.1
Kant-2					100.0	96.5	96.5	99.9	99.9	39.5

### 6.3. Loading, Boundary Condition, and Thickness Effects

The combined effect of loading and boundary conditions is considered. The influence of the length-to-thickness ratio,  $a/h$ , is also investigated. The loading sets considered are the same as those seen in the previous analyses. Each output variable is associated to a symbol as shown in Table 15. A thin plate is first considered by assuming  $a/h$  equals 100. Table 16 shows the sets of displacement variables that are needed to detect various output variables previously. Each row refers to a different loading condition, each column to different boundary conditions. Table 17 reports the plate models that precisely detect all the considered outputs for all the considered boundary and loading conditions. Tables 18 to 21 report the same type of results for

$a/h$  equal to 10 and 5, respectively. The obtained plate models are summarized in Table 22 where all the considered  $a/h$  cases are shown. Table 23 presents a comparison of the accuracies given by the plate models obtained and the classical theories. A plate with two clamped and two free edges is considered. For the sake of brevity, the accuracy data which are related to the other boundary conditions are not reported here. Fig. 7 shows the total number of terms as function of the error and  $a/h$ . This plot has been obtained following the approach exploited for

**TABLE 15**

Symbols that indicate the status of a displacement variable for different output variables

Active term	Inactive term	
▲	△	$u_z$
■	□	$\sigma_{xx}$
▼	▽	$\sigma_{xz}$
◆	◇	$\sigma_{zz}$

**TABLE 14**

Sets of effective terms for the considered loading conditions

▲□▽	▲■▼	▲□▽	▲■▼	△□▽
▲□▽	▲■▼	▲□▽	▲■▼	△□▽
▲■▼	▲□▽	▲■▼	▲□▽	▲■▼



**TABLE 17**

Sets of effective terms for the considered boundary and loading conditions,  $a/h = 100$

$\triangle \square \nabla \diamond$	$\blacktriangle \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$	$\triangle \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$
$\triangle \square \nabla \diamond$	$\blacktriangle \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$	$\triangle \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$
$\blacktriangle \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$	$\blacktriangle \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$	$\triangle \square \nabla \diamond$

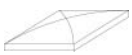
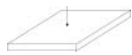
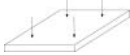
**TABLE 19**

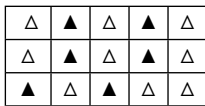
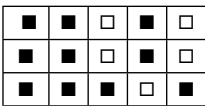
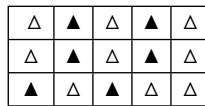
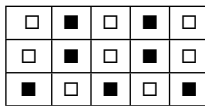
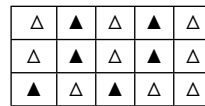
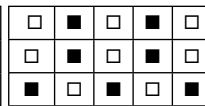
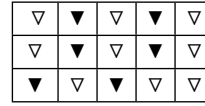
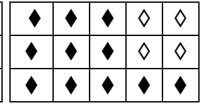
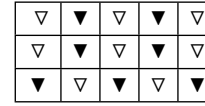
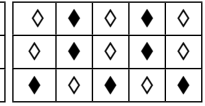
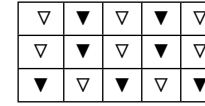
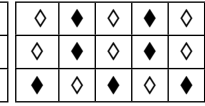
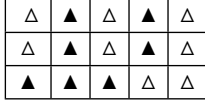
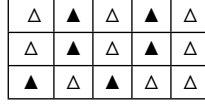
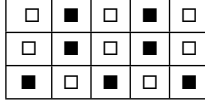
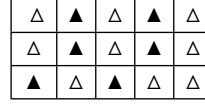
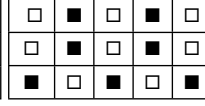
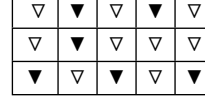
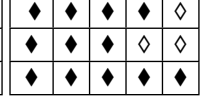
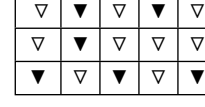
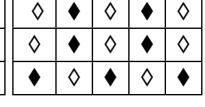
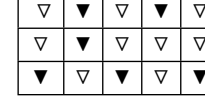
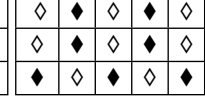
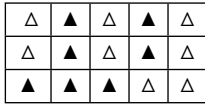
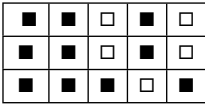
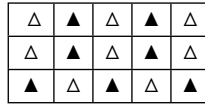
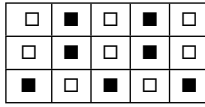
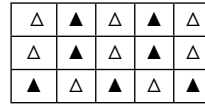
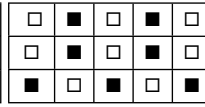
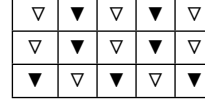
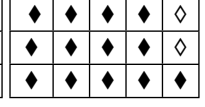
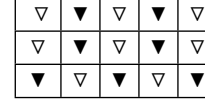
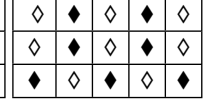
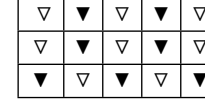
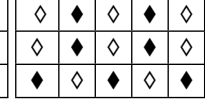
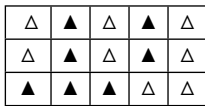
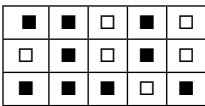
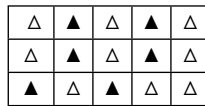
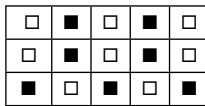
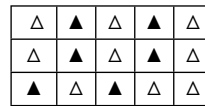
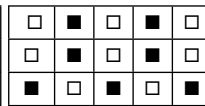
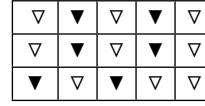
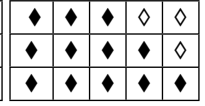
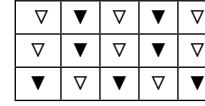
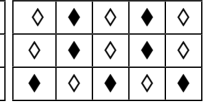
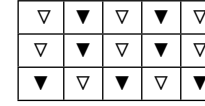
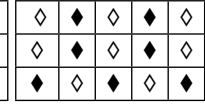
Sets of effective terms for all the considered boundary and loading conditions,  $a/h = 10$

$\triangle \blacktriangledown \blacklozenge$	$\blacktriangle \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$	$\blacktriangle \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$
$\triangle \blacktriangledown \blacklozenge$	$\blacktriangle \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$	$\blacktriangle \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$
$\blacktriangle \blacktriangledown \blacklozenge$	$\blacktriangle \blacktriangledown \blacklozenge$	$\blacktriangle \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$	$\blacktriangle \blacktriangledown \blacklozenge$

**TABLE 18**

Comparison of the sets of effective terms for plates with different boundary and loading conditions,  $a/h = 10$ .  $\blacktriangle = u_z$ ,  $\blacksquare = \sigma_{xx}$ ,  $\blacktriangledown = \sigma_{xz}$  and  $\blacklozenge = \sigma_{zz}$

						
<i>SSSS</i>						
						
<i>cfcf</i>						
						
<i>cccc</i>						
						
<i>SSC</i>						

**TABLE 20**

Comparison of the sets of effective terms for plates with different boundary and loading conditions,  $a/h = 5$ ,  $\blacktriangle = u_z$ ,  $\blacksquare = \sigma_{xx}$ ,  $\blacktriangledown = \sigma_{xz}$  and  $\blacklozenge = \sigma_{zz}$

<i>SSSS</i>					
<i>cfcf</i>					
<i>cccc</i>					
<i>scsc</i>					

**TABLE 21**

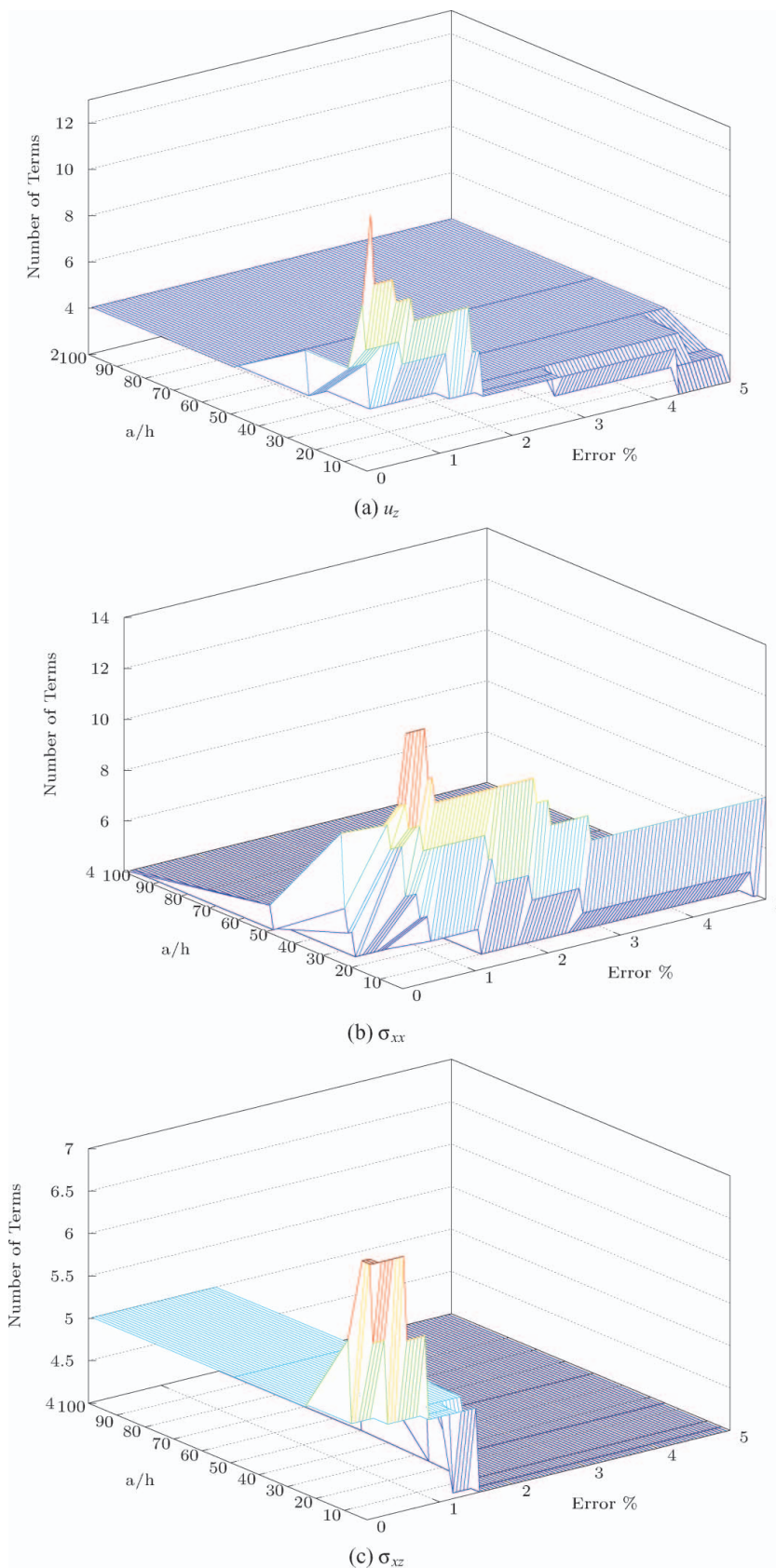
Sets of effective terms for the considered boundary and loading conditions,  $a/h = 5$

$\blacktriangle\blacktriangledown\blacklozenge$	$\blacktriangle\blacksquare\blacktriangledown$	$\triangle\circ\blacktriangledown\blacklozenge$	$\blacktriangle\blacksquare\blacktriangledown\blacklozenge$	$\triangle\circ\blacktriangledown\blacklozenge$
$\blacktriangle\blacktriangledown\blacklozenge$	$\blacktriangle\blacksquare\blacktriangledown\blacklozenge$	$\triangle\circ\blacktriangledown\blacklozenge$	$\blacktriangle\blacksquare\blacktriangledown\blacklozenge$	$\triangle\circ\blacktriangledown\blacklozenge$
$\blacktriangle\blacksquare\blacktriangledown\blacklozenge$	$\blacktriangle\blacksquare\blacktriangledown\blacklozenge$	$\blacktriangle\blacksquare\blacktriangledown\blacklozenge$	$\triangle\blacksquare\blacktriangledown\blacklozenge$	$\blacktriangle\blacksquare\blacktriangledown\blacklozenge$

have been considered. A fourth-order solution has been adopted as reference. The accuracy analysis has been conducted via a so-called mixed asymptotic/axiomatic approach, which determines the role of each displacement variable in computing a given displacement/stress variable. The present approach has proved its validity in constructing:

1. Reduced plate models equivalent to a full higher-order theory;
2. Reduced plate models able to fulfil a given accuracy input.

## GUIDELINES FOR BENDING ANALYSIS OF PLATES



**FIG. 7.** Influence of  $a/h$ , the output variable, and the tolerance on the total number of model terms for a simply supported plate with a distributed load. (Color figure available online)

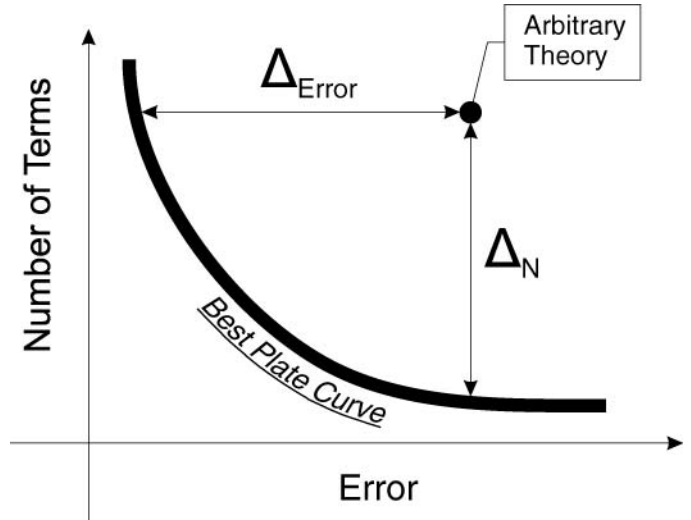
**TABLE 22**

Influence of  $a/h$  on the displacement field terms for combined boundary and loading conditions

$a/h = 100$	$\triangle \square \nabla \diamond$	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$	$\triangle \blacksquare \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$
	$\triangle \square \nabla \diamond$	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$	$\triangle \blacksquare \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$
	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$	$\triangle \square \nabla \diamond$
$a/h = 10$	$\triangle \blacksquare \blacktriangledown \blacklozenge$	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$
	$\triangle \blacksquare \blacktriangledown \blacklozenge$	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$
	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$
$a/h = 5$	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$
	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$
	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\triangle \blacksquare \blacktriangledown \blacklozenge$	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$

As far as the first type of models is concerned, the following conclusions can be drawn:

- All the parameters considered (loadings, boundary conditions, and thickness) are important to determine the sets of effective terms; that is, as one of these parameters changes, a different plate model is required.



**FIG. 8.** An example of Best Plate Diagram (BPD).

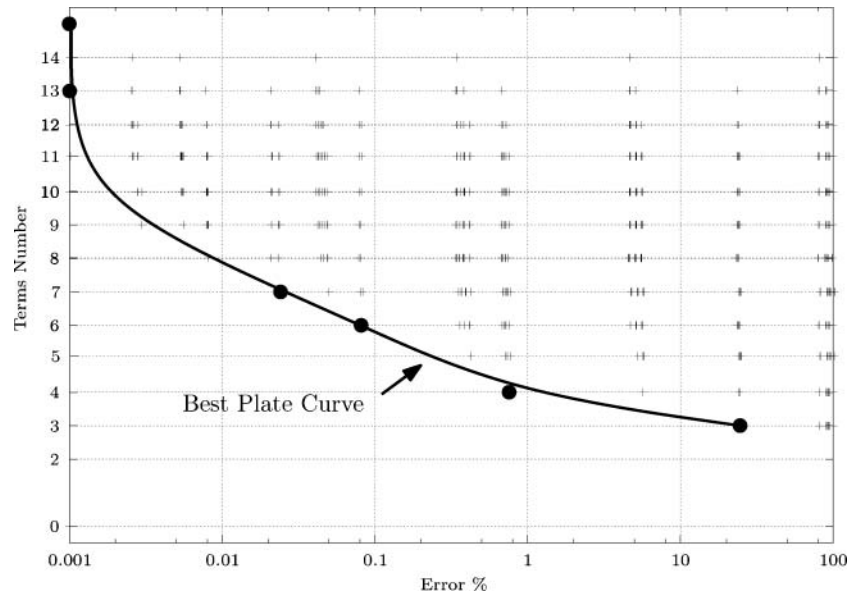
- The influence of the length-to-thickness ratio is particularly strong.
- The use of full models is mandatory when a complete set of results is needed, since only few can be discarded.

**TABLE 23**

Accuracy of different displacement models in the case of a plate with two clamped and two free edges subject to a point load

	$\delta_{u_z}$ [%]	$\delta_{\sigma_{xx}}$ [%]	$\delta_{\sigma_{yy}}$ [%]	$\delta_{\sigma_{xz}}$ [%]	$\delta_{\sigma_{zz}}$ [%]	$\delta_{\sigma_{zz}}$ [%]
$a/h = 100$						
$\triangle \square \nabla \diamond$	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$	$\triangle \blacksquare \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$		
$\triangle \square \nabla \diamond$	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$	$\triangle \blacksquare \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$		
$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$	$\triangle \square \nabla \diamond$		
CPT	102.7	98.5	98.0	158.6	710.2	1337.4
FSDT	102.9	98.5	98.0	80.0	114.3	1337.6
Kant-2	100.0	100.0	100.0	99.9	117.9	99.0
$a/h = 10$						
$\triangle \blacksquare \blacktriangledown \blacklozenge$	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$		
$\triangle \blacksquare \blacktriangledown \blacklozenge$	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$		
$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$		
CPT	86.2	82.5	79.2	202.9	-106.014	449.824
FSDT	101.6	82.7	80.24	83.6	72.7	453.2
Kant-2	100.0	96.2	95.7	98.7	97.7	39.8
$a/h = 5$						
$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$		
$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\triangle \square \nabla \diamond$		
$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$	$\triangle \blacksquare \blacktriangledown \blacklozenge$	$\blacktriangle \blacksquare \blacktriangledown \blacklozenge$		
CPT	60.7	70.6	65.7	228.9	-90.6	189.9
FSDT	102.8	70.9	68.1	88.7	71.6	193.5
Kant-2	100.4	89.7	88.7	96.0	99.1	-23.1

## GUIDELINES FOR BENDING ANALYSIS OF PLATES



**FIG. 9.** Accuracy of all the possible combinations of plate models in computing  $u_z$  for the simply supported plate loaded by a distributed load (each “+” indicates a different plate model).

The construction of the second type of plate modes has highlighted that Unified Formulation allows us, for a given problem, to obtain a diagram that in terms of accuracy (input) gives an answer to the following fundamental questions (see Figs. 3, 4, and 6):

- What is the “minimum” number of the terms,  $N_{min}$ , to be used in a Finite Element Plate model?
- Which are the terms to be retained, that is, which are the generalized displacement variables to be used as FE dof’s?

To the best of the authors’ knowledge, there are no other available methods that can provide this kind of results. The present method of analysis is able to create plots like the one in Fig. 8 that gives the number of terms as function of the permitted error. This plot can be defined as Best finite element Plate Diagram, BPD, since it allows to edit an arbitrary given theory in order to have a lower amount of terms for a given error (vertical shift,  $\Delta_N$ ) or to increase the accuracy keeping the computational cost constant (horizontal shift,  $\Delta_{error}$ ). Most times, the plot presented appears as a hyperbole. CUF makes the computation of such a plot possible. Note that the diagram has the following properties:

- It changes by changing problems ( $a/h$ , loadings, boundary conditions, etc.);
- It changes by changing output variable (displacement/stress components, or a combination of these).

The validity of the BPD is tested by computing the accuracy of all the plate models obtainable as a combination of the 15 terms of the fourth-order theory. The results are reported in Fig. 9 in the case of a simply supported plate loaded by a

distributed load,  $u_z$  is considered as output variable. The BPD perfectly matches the lower boundaries of the region where all the models lie. This confirms that the BPD represents the best theory (i.e. the least cumbersome) for a given error. The BPD permits the evaluation of any existing plate FE elements, as in the previous sections. The distance from the BPD of a given known FE model represents a guideline to recommend any other plate theory. More complex analyses will be conducted in future work to investigate the effects of the orthotropic ratio  $E_L/E_T$ , stacking sequence layout and variable kinematic description such as layer-wise or equivalent single layer models. Some of these analyses have been already studied in [37], where the attention was restricted to closed-form form solution.

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